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JUNCTURE STRESS FIELDS IN MULTICELLULAR SHELL STRUCTURES

VOL. II STRESSES AND DEFORMATIONS OF FIXED-EDGE SEGMENTAL CYLINDRICAL SHELLS

by

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Technical Report

JUNCTURE STRESS FIELDS IN MULTICELLULAR
SHELL STRUCTURES

Final Report

Nine Volumes

Vol. II Stresses and Deformations of Fixed-Edge
Segmental Cylindrical Shells

by

E. Y. W. Tsui
F. A. Brogan
P. Stern

Contract NAS 8-11480 to National Aeronautics and Space Administration
George C. Marshall Space Flight Center, Huntsville, Alabama

FOREWORD

This report is the result of a study on the numerical analysis of stresses and deformations of fixed-edge orthotropic and isotropic segmental cylindrical shells under uniform and hydrostatic pressures as well as linear thermal gradient across the thickness of the shell. Work on this study was performed by staff members of Lockheed Missiles & Space Company in cooperation with the George C. Marshall Space Flight Center of the National Aeronautics and Space Administration under Contract NAS 8-11480. Contract technical representative was H. Coldwater.

This volume is the second of a nine-volume final report of studies conducted by the department of Solid Mechanics, Aerospace Sciences Laboratory, Lockheed Missiles & Space Company. Project Manager was K. J. Forsberg; E. Y. W. Tsui was Technical Director for the work.

The nine volumes of the final report have the following titles:

- Vol. I Numerical Methods of Solving Large Matrices
- Vol. II Stresses and Deformations of Fixed-Edge Segmental Cylindrical Shells
- Vol. III Stresses and Deformations of Fixed-Edge Segmental Conical Shells
- Vol. IV Stresses and Deformations of Fixed-Edge Segmental Spherical Shells
- Vol. V Influence Coefficients of Segmental Shells
- Vol. VI Analysis of Multicellular Propellant Pressure Vessels by the Stiffness Method
- Vol. VII Buckling Analysis of Segmental Orthotropic Cylinders under Uniform Stress Distribution
- Vol. VIII Buckling Analysis of Segmental Orthotropic Cylinders under Non-uniform Stress Distribution
- Vol. IX Summary of Results and Recommendations

SUMMARY

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This volume presents a set of basic equations for orthotropic thin elastic cylindrical shells and a digital program for the analysis of static response of segmental cylinders with fixed edges under the following loading conditions:

- Linearly varying pressure
- Uniform pressure
- Linear thermal gradient through the thickness of shell

The problem is solved numerically by means of finite-difference technique, using a direct method of solving a large system of simultaneous equations. Also, a numerical example showing the stresses and deformations of a segmental orthotropic cylinder under uniform pressure is presented.

Author

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NOTATION

a_i , b_i , c_i , A , B , C ,	}	nondimensional parameters defined in text
$D()$, $E()$, $M()$, $N()$,		
$\bar{Q}()$, $S()$, u , v , w , x ,		
$Z()$, θ , ρ , κ		
b		spacing of stiffeners (stringers and ribs)
c		eccentricity of stiffener with respect to middle-surface of shell
d		depth of stiffener
D		flexural rigidity of shell = $Eh^3/12(1 - \nu^2)$
E		modulus of elasticity
F_i		boundary force at Station i
F^f		boundary forces of fixed-edge shell due to applied forces or thermal gradients
G		shear modulus
h		thickness of shell
\bar{h} , \bar{k}		mesh spacings in x - and θ - coordinate directions
m , n		number of columns and rows of the mesh
i , j		dummy subscripts
k , k_{ij}		stiffness influence coefficients
L		length of cylindrical segment
$\hat{M}()$, $\hat{N}()$		moments and stress resultants
$P()$		surface or body forces
$Q()$		transverse shears
R		radius

R_i	concentrated forces at corners of shell boundaries in the z-direction
t	width of stiffener
T	change of temperature
$\hat{u}, \hat{v}, \hat{w}$	displacement components in directions $\hat{x}, \hat{\theta}, \hat{z}$
$\hat{x}, \hat{\theta}, \hat{z}$	cylindrical coordinates
$\hat{\theta}_c$	angle subtending one-half width of cylindrical segment
ξ, η	orthogonal coordinates along boundaries of shell
δ_i	boundary deformations (displacements or rotations) at Station i
$\epsilon(), \gamma()$	direct and shear strains
$\hat{\chi}()$	changes of curvature or torsion of middle-surface
ν	Poisson's ratio
$\omega()$	rotations of the normal at the middle-surface
$(), \hat{x}$	$\frac{\partial()}{\partial \hat{x}}$
$()_i^j$	functions at a discrete point i, j where i, j implies the \hat{x} - and $\hat{\theta}$ -directions respectively
ϵ^T	thermal strain = coefficients of linear expansion times the change of temperature, T
Φ	rotation in the middle-surface around the normal

Additional notations and symbols are defined in the text.

Section 1
INTRODUCTION

As a result of an investigation of juncture stress fields peculiar to the multicellular pressure vessels (Fig. 1), a theory for the prediction of the membrane and bending stresses and the corresponding deformations for such shell structures was formulated.*

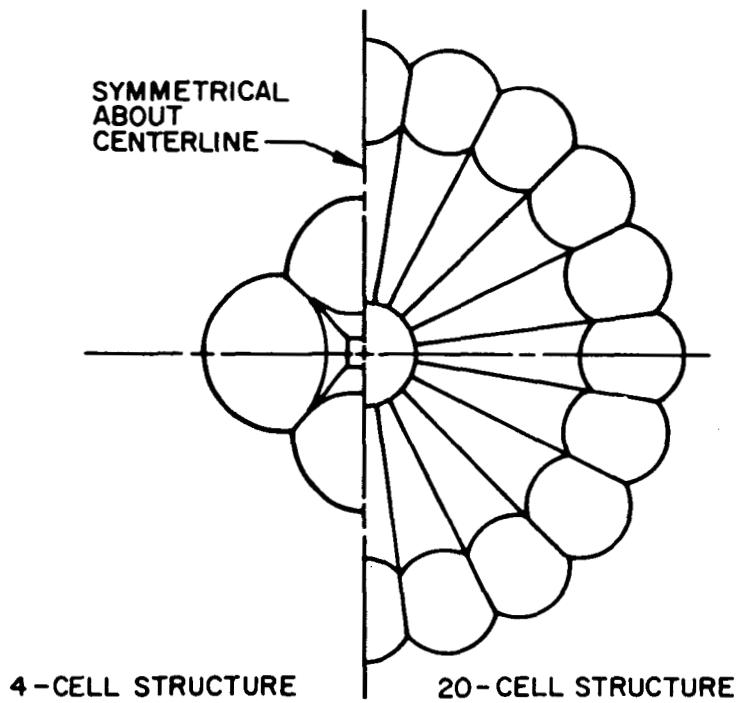


Fig. 1 Multicellular Shell Structure

*"Investigation of Juncture Stress Fields in Multicellular Shell Structures," by E. Y. W. Tsui, F. A. Brogan, J. M. Massard, P. Stern, and C. E. Stuhlman, Technical Report M-03-63-1, Lockheed Missiles & Space Company, Sunnyvale, Calif., Feb 1964 - NASA CR-61050.

Due to the fact that analytic solutions are still lacking, it was decided to solve the problem numerically by means of finite-difference technique. To ensure the feasibility of such a numerical solution, a direct method of solving large matrices with a high-speed digital computer was also developed.

According to the previous work, if the stiffness or displacement method is used, the total forces and hence the corresponding stresses along the juncture of the shell segments (Fig. 2) may be expressed concisely in the following matrix form

$$F = k\delta + F^f \quad (1.1)$$

where k is the stiffness matrix, δ are the deformations, and F^f are the fixed-end forces due to applied loads or thermal gradients. In view of this situation, it is logical to solve the problem systematically by the established general procedure of analysis already described.* This procedure may be stated briefly as follows:

1. Determination of the fixed-end forces, F^f , along the boundary as well as stresses and deformations in the interior of shell segments due to loads
2. Determination of the influence coefficients, k_{ij} , along the boundaries of shell segments, i.e., the induced forces at points i due to unit deformations ($\delta = 1$) at points j
3. Determination of the actual deformations, δ , along the shell boundaries; this requires the satisfaction of both compatibility and equilibrium conditions at the junctures of the structure

Once all the work involved in these three steps is completed, the total stresses and deformations in the specific discrete interior locations may be obtained.

*"Investigation of Juncture Stress Fields in Multicellular Shell Structures," by E. Y. W. Tsui, F. A. Brogan, J. M. Massard, P. Stern, and C. E. Stuhlman, Technical Report M-03-63-1, Lockheed Missiles & Space Company, Sunnyvale, Calif., Feb 1964 - NASA CR-61050.

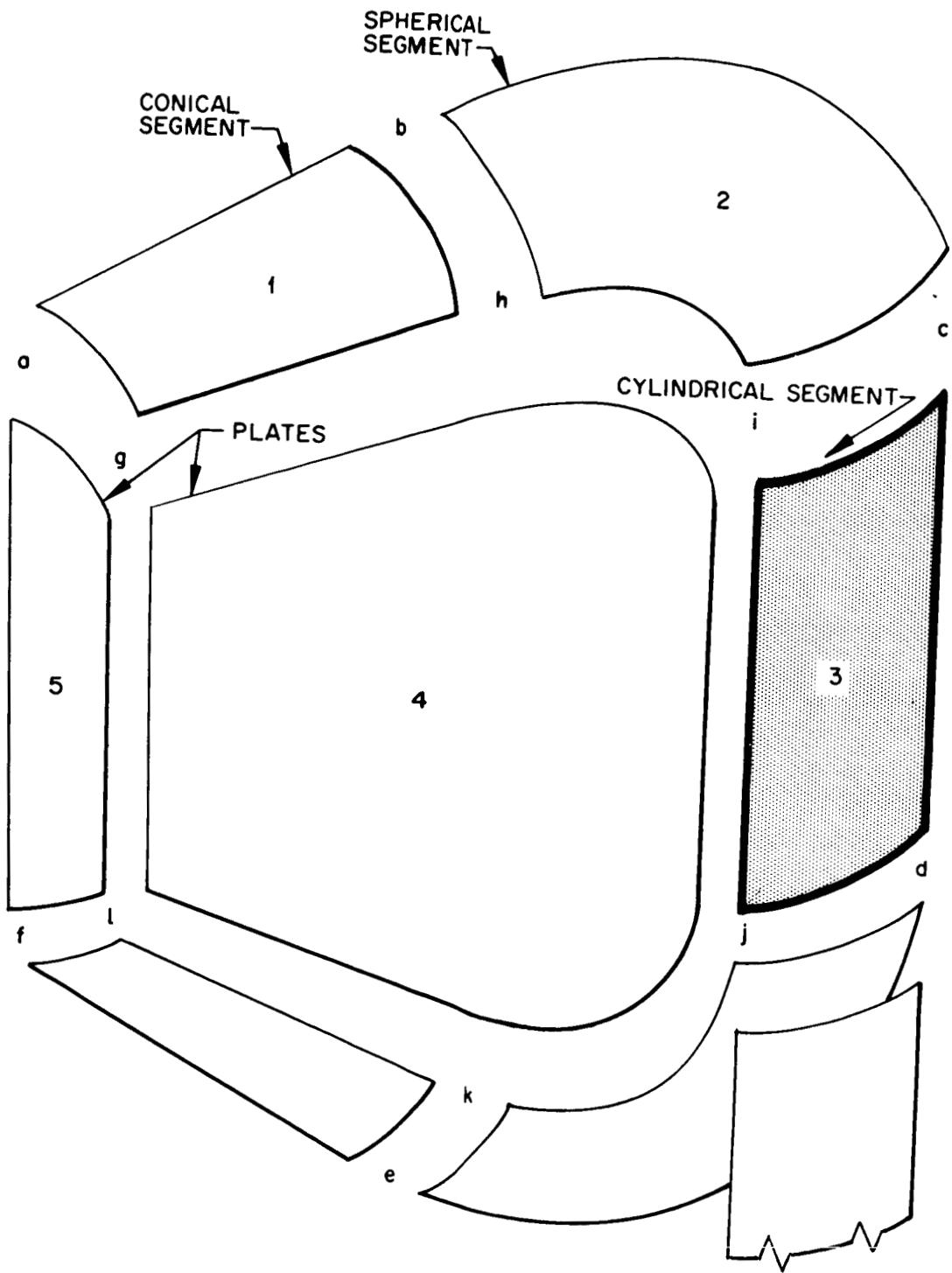


Fig. 2 Basic Shell Elements of Multicellular Structure

This volume presents results of the work involved in Step 1 only and covers the following items:

- Nondimensional formulation of the problem
- Detailed description of a workable digital program for the generation of solutions
- Tabulation of stresses and deformations of an orthotropic segmental cylinder with fixed edge under uniform internal pressure

Section 2

FORMULATION OF PROBLEM

The necessary analytical expressions for an anisotropic cylinder have already been given.* In order that this report be complete in the sense of equations, all the required equations specialized to that of an anisotropic cylinder are presented. Further, boundary conditions for a boundary which coincides with coordinate axes are given. Finally, the governing equations are written in difference form and the ordering of the equations are given so that they can be solved by the "direct method" described in Vol. I.

2.1 ANALYTICAL FORMULATION

The geometry of the cylindrical panel under consideration is shown in Fig. 3. Symmetry is noted in the θ -direction. This condition does not restrict the equation system which follows; it only affects the method of solution.

Anisotropy of the cylinder is due to ribs and stringers of the same size which are equally spaced in both directions as shown in Fig. 4. If the spacing b between ribs and stringers is large, it is necessary to analyze the cylinder as if it were composed of shell panels and stiffeners. However, when the spacing becomes small the structure assumes an anisotropic shell. Figure 5 shows further details of an element of rib and stringer geometry.

To obtain a formulation which will yield solutions covering a wide range of shell parameters, all geometrical and dependent variables have been nondimensionalized and normalized in the following manner:

*"Investigation of Juncture Stress Fields in Multicellular Shell Structures," by E. Y. W. Tsui, F. A. Brogan, J. M. Massard, P. Stern, and C. E. Stuhlmeyer, Technical Report M-03-63-1, Lockheed Missiles & Space Company, Sunnyvale, Calif., Feb 1964 - NASA CR-61050.

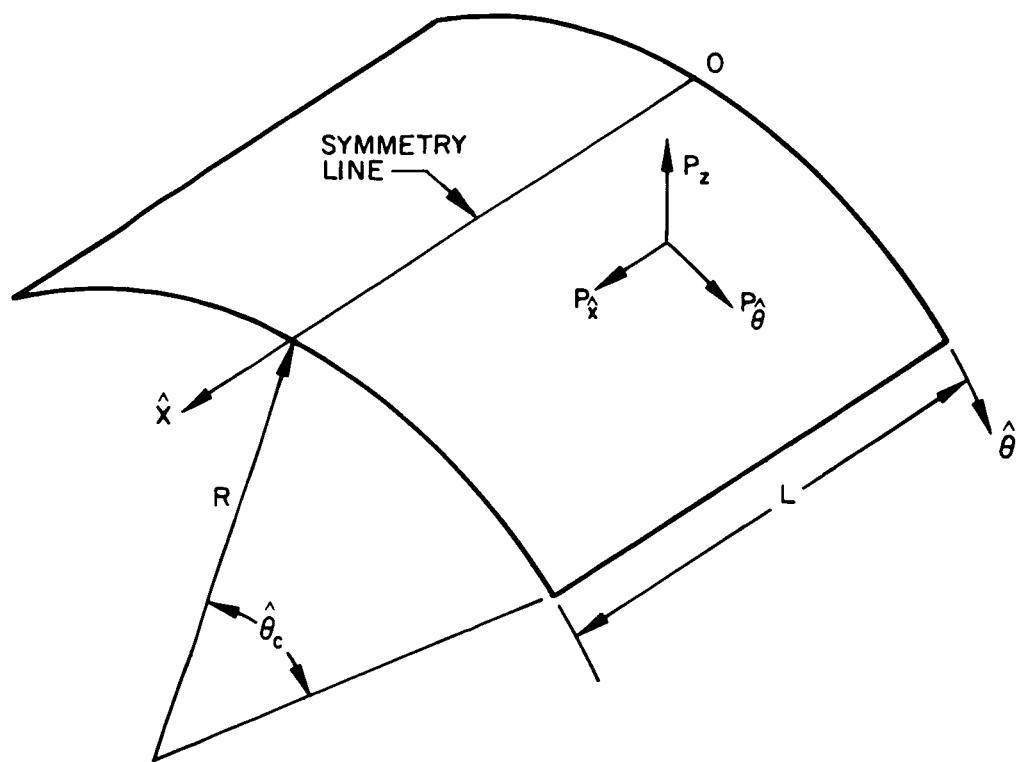


Fig. 3 Geometry of Cylindrical Panel

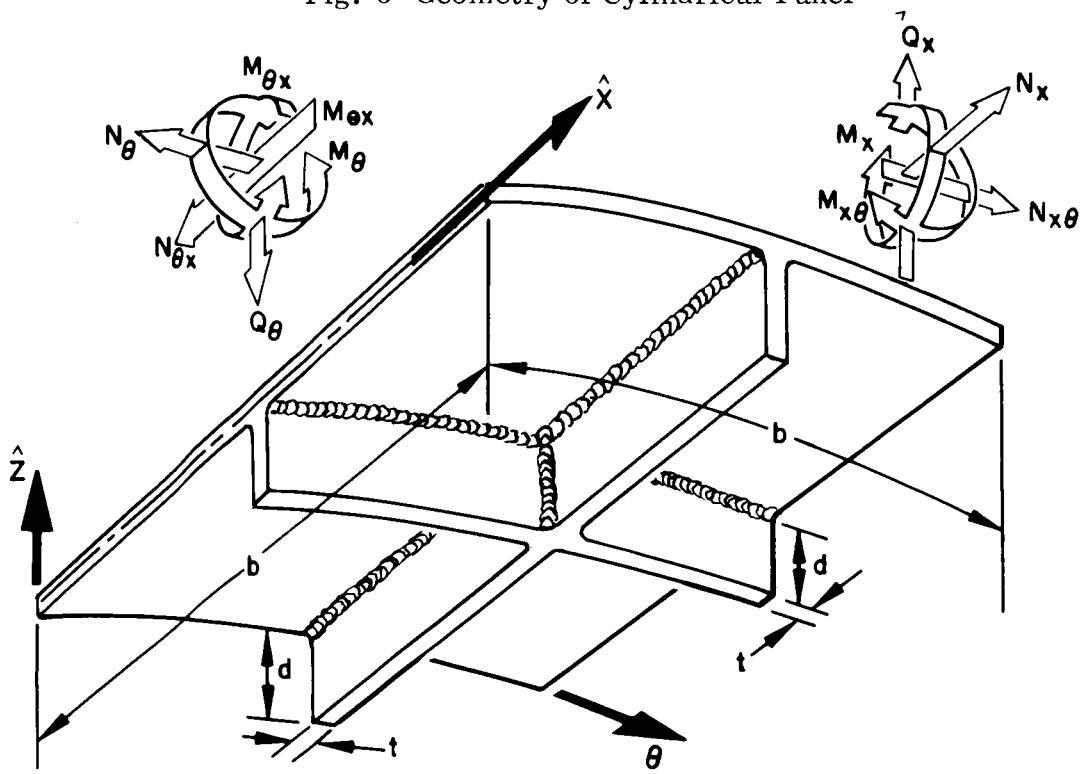


Fig. 4 Element of Stiffened Shell

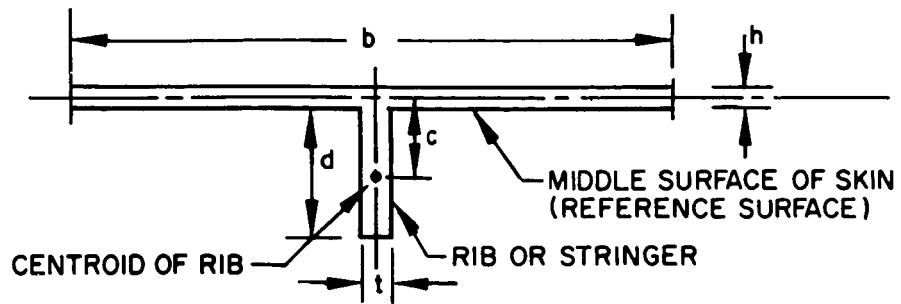


Fig. 5 Rib and Stringer Geometry

For the coordinate system

$$x = \hat{x}/L \quad (2.1a)$$

$$\theta = \hat{\theta}/\theta_c \quad (2.1b)$$

$$z = \hat{z}/R \quad (2.1c)$$

$$\rho = R\theta_c/L \quad (2.1d)$$

$$\bar{h} = x/n \quad (2.1e)$$

$$\bar{k} = \theta/m \quad (2.1f)$$

For the dependent variables

$$u = \hat{u}/R \quad (2.2a)$$

$$v = \hat{v}/R \quad (2.2b)$$

$$w = \hat{w}/R \quad (2.2c)$$

$$N = \hat{N}/Eh \quad (2.2d)$$

$$M = \hat{M}/EhR \quad (2.2e)$$

$$\chi = R\hat{\chi} \quad (2.2f)$$

Consequently, the basic shell equations can be written as shown in Secs. 2.1.1 through 2.1.4.

2.1.1 Rotation-Displacement Relations

Positive displacements and rotations of the reference surface (Fig. 6) are shown in Fig. 7 and are related by the equations:

$$\omega_x = -(\rho/\theta_c) w_{,x} \quad (2.3a)$$

$$\omega_\theta = v - (1/\theta_c) w_{,x} \quad (2.3b)$$

$$\Phi = [(\rho/\theta_c) v_{,x} - (1/\theta_c) u_{,\theta}] / 2 \quad (2.3c)$$

2.1.2 Strain-Displacement Relations

The strains of the reference surface are related to displacements by

$$\bar{\epsilon}_x = (\rho/\theta_c) u_{,x} \quad (2.4a)$$

$$\bar{\epsilon}_\theta = 1/\theta_c v_{,\theta} + w \quad (2.4b)$$

$$\bar{\gamma} = (1/\theta_c) u_{,x} + (\rho/\theta_c) v_{,x} \quad (2.4c)$$

and the changes of curvature and torsion are

$$\chi_x = -(\rho/\theta_c)^2 w_{,xx} \quad (2.5a)$$

$$\chi_\theta = (1/\theta_c) v_{,\theta} - (1/\theta_c)^2 w_{,\theta\theta} \quad (2.5b)$$

$$\chi_{x\theta} = (\rho/\theta_c) v_{,x} - (\rho/\theta_c)^2 w_{,x\theta} \quad (2.5c)$$

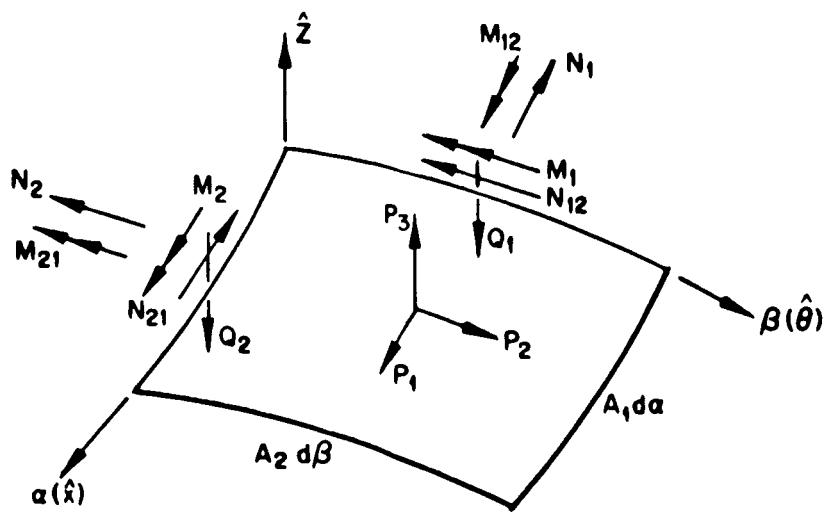


Fig. 6 Stress Resultants, Moments and Loads

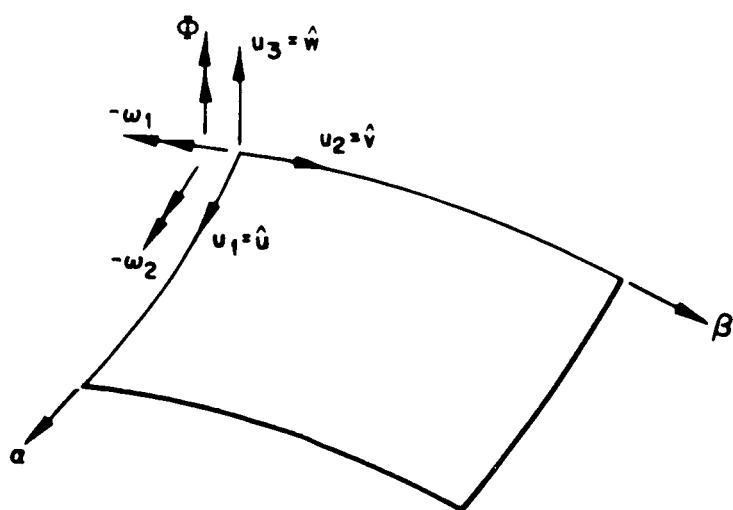


Fig. 7 Displacements and Rotations

The strains at a distance z from the reference surface are given by

$$\epsilon_x = \bar{\epsilon}_x + z\chi_x \quad (2.6a)$$

$$\epsilon_\theta = \bar{\epsilon}_\theta + z\chi_\theta \quad (2.6b)$$

$$\gamma_{x\theta} = \bar{\gamma}_{x\theta} + 2z\chi_{x\theta} \quad (2.6c)$$

2.1.3 Constitutive Relations

Positive stress resultants are shown in Fig. 6. These stress resultants are related to strains by the equations

$$N_x = E_x \bar{\epsilon}_x + E_\nu \bar{\epsilon}_\theta - S_c \chi_x + N^T \quad (2.7a)$$

$$N_\theta = E_\nu \bar{\epsilon}_x + E_\theta \bar{\epsilon}_\theta - S_\theta \chi_\theta + N^T \quad (2.7b)$$

$$N_{\theta x} = E_{\theta x} \bar{\gamma} \quad (2.7c)$$

$$N_{x\theta} = E_{\theta x} \bar{\gamma} + D_{\theta x}(\chi_{x\theta}) \quad (2.7d)$$

$$M_x = D_x(\chi_x) + D_\nu(\chi_\theta) - S_c \bar{\epsilon}_x + M^T \quad (2.7e)$$

$$M_\theta = D_x(\chi_x) + D_\theta(\chi_\theta) - S_c \bar{\epsilon}_\theta + M^T \quad (2.7f)$$

$$M_{x\theta} = M_{\theta x} = D_{\theta x}(\chi_{x\theta}) \quad (2.7g)$$

$$Q_x = (\rho/\theta_c) \left[D_x(\chi_x),_x + D_\nu(\chi_\theta),_x - S_c \bar{\epsilon}_{x,x} + M_{,x}^T \right] + (D_{\theta x}/\theta_c)(R\chi_{x\theta}),_\theta \quad (2.7h)$$

$$Q_\theta = (\rho/\theta_c) D_{\theta x}(\chi_{x\theta}),_x + (1/\theta_c) \left[D_x(\chi_x),_\theta + D_\theta(\chi_\theta),_\theta - S_c \bar{\epsilon}_{\theta,\theta} + M_{,\theta}^T \right] \quad (2.7i)$$

where

$$E_x = E_\theta = [1 + (t/h)(d/b)(1 - \nu^2)] / (1 - \nu^2)$$

$$E_\nu = \nu / (1 - \nu^2)$$

$$E_{\theta x} = 1/2(1 + \nu)$$

$$S_c = (c/R)(t/h)(d/b)$$

$$D_x = D_\theta = \kappa \{1 + (t/h)(d/b) [(d/h)^2 + 12(c/h)^2] (1 - \nu^2)\} / (1 - \nu^2)$$

$$D_\nu = \nu \kappa / (1 - \nu^2)$$

$$D_{\theta x} = \kappa \{1 + 2(t/h)^3 (d/b)\} / (1 + \nu)$$

$$N^T = -[1/(1 - \nu) h] \left[\int_{-h/2}^{h/2} \alpha T dz + (t/b) \int_{-(d+h/2)}^{-h/2} \alpha T dz \right]$$

$$M^T = -[1/(1 - \nu) hR] \left[\int_{-h/2}^{h/2} \alpha T dz + (t/b) \int_{-(d+h/2)}^{-h/2} \alpha T dz \right]$$

$$\kappa = \frac{h^2}{12 R^2}$$

2.1.4 Governing Differential Equations

The governing differential equations for an anisotropic cylinder in terms of the displacement components u , v , and w are given by

$$a_1 u_{xx} + a_2 u_{\theta\theta} + a_3 v_{x\theta} + a_4 w_{xxx} + a_5 w_x = A \quad (2.8a)$$

$$b_1 u_{,x\theta} + b_2 v_{,xx} + b_3 v_{,\theta\theta} + b_4 w_{,xx\theta} + b_5 w_{,\theta\theta\theta} + b_6 w_{,\theta} = B \quad (2.8b)$$

$$c_1 u_{,xxx} + c_2 u_{,x} + c_3 v_{,xx\theta} + c_4 v_{,\theta\theta\theta} + c_5 v_{,\theta} + c_6 w_{,xxxx} + c_7 w_{,xx\theta\theta} + c_8 w_{,\theta\theta\theta\theta} \\ + c_9 w_{,\theta\theta} + c_{10} w = C \quad (2.8c)$$

where

$$a_1 = (\rho/\theta_c)^2 E_x$$

$$b_1 = a_3$$

$$a_2 = E_{\theta x}/\theta_c^2$$

$$b_2 = (\rho/\theta_c)^2 (E_{\theta x} + 2D_{\theta x})$$

$$a_3 = (\rho/\theta_c^2) (E_\nu + E_{\theta x})$$

$$b_3 = (1/\theta_c^2) (E_\theta + D_\theta - 2S_c)$$

$$a_4 = S_c (\rho/\theta_c)^3$$

$$b_4 = -(\rho^2/\theta_c^3) (D_\nu + 2D_{\theta x})$$

$$a_5 = (\rho/\theta_c) E_\nu$$

$$b_5 = -(1/\theta_c^3) (D_\theta - S_c)$$

$$b_6 = (1/\theta_c) (E_\theta - S_c)$$

$$c_1 = (\rho/\theta_c)^3 S_c$$

$$c_6 = (\rho/\theta_c)^4 D_x$$

$$c_2 = a_5$$

$$c_7 = 2(\rho^2/\theta_c^4) (D_\nu + D_{\theta x})$$

$$c_3 = b_4$$

$$c_8 = (1/\theta_c^4) D_\theta$$

$$c_4 = b_5$$

$$c_9 = (2/\theta_c^2) S_c$$

$$c_5 = b_6$$

$$c_{10} = E_\theta$$

$$A = -(p_x R/Eh) - (p/\theta_c) N_x^T$$

$$B = -(p_\theta R/Eh) - (1/\theta_c) N_\theta^T - (1/\theta_c) M_\theta^T$$

$$C = (p_z R/Eh) - N^T + (p/\theta_c)^2 M_{xx}^T + (1/\theta_c)^2 M_{\theta\theta}^T$$

The right-hand vectors of Eq. (2.8) for three types of loading and the corresponding dimensional displacement components are given below:

- Uniform normal pressure

$$A = 0$$

$$B = 0$$

$$C = p_z R/Eh = 1.0$$

$$\hat{u} = u p_z R^2/Eh$$

$$\hat{v} = v p_z R^2/Eh$$

$$\hat{w} = w p_z R^2/Eh$$

- Linearly varying pressure

$$A = 0$$

$$B = 0$$

$$C = (R/Eh)(p_\theta + p_x)$$

$$\hat{u} = u R^2/Eh$$

$$\hat{v} = v R^2/Eh$$

$$\hat{w} = w R^2/Eh$$

- Linear temperature gradient

$$A = 0$$

$$B = 0$$

$$C = \alpha \left[\frac{T_1}{1 - \nu} + \left(\frac{t}{h} \right) \left(\frac{d}{b} \right) (T_i - T_o) \right]$$

where

$$\begin{aligned}\alpha &= \text{coefficient of thermal expansion} \\ T_o &= \text{ambient temperature at which the structure is stress free} \\ T_1 &= \frac{T_e + T_i}{2} - T_o \\ T_e, T_i &= \text{external and internal temperatures of shell skin} \\ \hat{u} &= uR \\ \hat{v} &= vR \\ \hat{w} &= wR\end{aligned}$$

It should be noted that Eqs. (2.7) and (2.8) reduce to the case of isotropic shells if $d = t = 0$.

2.2 BOUNDARY CONDITIONS

The boundary conditions have already been given* for an arbitrary shell with a smooth boundary curve. These boundary conditions are in terms of displacements and stress resultants. If the boundary curve is not smooth it is possible that a concentrated load must be applied at the slope discontinuity. This concentrated load can be established by use of the work done by the boundary forces,

$$\begin{aligned}w^b &= \int_c \left[\left(N_\eta + \frac{M_{\xi\eta}}{R} \right) u_\eta + \left(N_\xi - \frac{M_{\xi\eta}}{R_\eta} \right) u_\xi \right. \\ &\quad \left. + Q_3 w + M_\xi \omega_\eta - \frac{M_{\xi\eta}}{A_\eta} \frac{\partial w}{\partial \eta} \right] d\xi \quad (2.9)\end{aligned}$$

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The last term in equation can be integrated by parts to yield

$$W^b = \int_c \left[\left(N_\eta + \frac{M_{\zeta\eta}}{R} \right) u_\eta + \left(N_\zeta - \frac{M_{\zeta\eta}}{R_\eta} \right) u_\zeta + \left(Q_3 + \frac{1}{A_\eta} \frac{\partial M_{\zeta\eta}}{\partial \eta} \right) w \right. \\ \left. + M_\zeta \omega_\eta \right] d\zeta - \sum_{i=1}^K R_i w(i) \quad (2.10)$$

where

$$R_i = [M_{\zeta\eta}(\eta_2) - M_{\zeta\eta}(\eta_1)]_i$$

Thus R_i is a concentrated load in the direction of z at which the boundary curve has a discontinuous slope.

At this point a specific boundary curve is considered. This is shown in Fig. 8 as \overline{abcd} . The boundary forces for various sections of this boundary curve are given

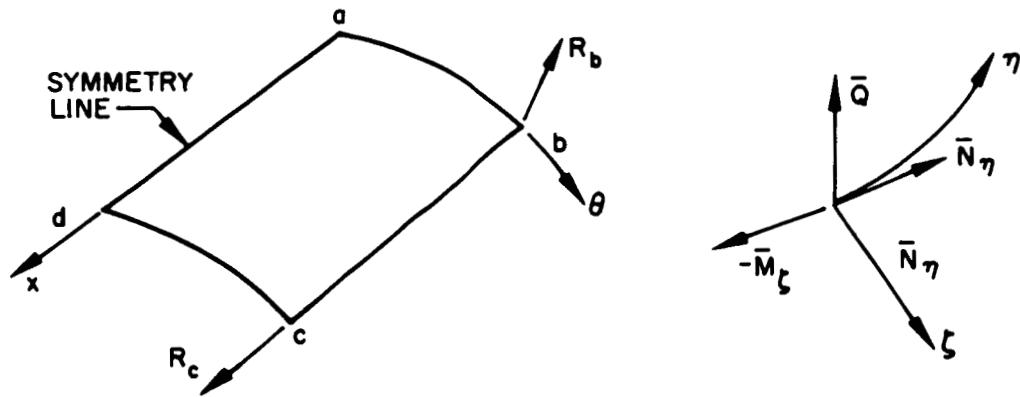


Fig. 8 Boundary Curve and Forces

by the equations:

\overline{ab} :

$$\overline{N}_\eta = N_{x\theta} + \frac{1}{R} M_{x\theta}$$

$$\overline{N}_\zeta = N_x$$

$$\overline{Q} = - \left[Q_x + \frac{1}{R} M_{x\theta, \theta} \right]$$

$$\overline{M}_\zeta = M_x$$

$$R_b = 2 M_{x\theta}$$

\overline{bc} :

$$\overline{N}_\eta = - N_{\theta x}$$

$$\overline{N}_\zeta = N_\theta$$

$$\overline{Q} = [Q_\theta + M_{\theta x, x}]$$

$$\overline{M}_\zeta = M_\theta$$

\overline{cd} :

$$\overline{N}_\eta = N_{x\theta} + \frac{1}{R} M_{x\theta}$$

$$\overline{N}_\zeta = N_x$$

$$\overline{Q} = Q_x + \frac{1}{R} M_{x\theta, \theta}$$

$$\overline{M}_\zeta = M_x$$

At the corners b and c there is a possible reaction load given by

$$R_i = 2 M_{x\theta}(b; c)$$

For a fixed-edge cylinder, the displacement components are all zero. Thus, for the boundary curve \overline{abcd} , the required boundary conditions are

$$\overline{ab}: u \equiv v \equiv w \equiv \frac{\partial w}{\partial x} \equiv 0$$

$$\overline{bc}: u \equiv v \equiv w \equiv \frac{\partial w}{\partial \theta} \equiv 0$$

$$\overline{cd}: u \equiv v \equiv w \equiv \frac{\partial w}{\partial x} \equiv 0$$

Along \overline{ad} the symmetry conditions are

$$v = 0$$

$$\omega_\theta = 0$$

$$N_{\theta x} = 0 ; M_{\theta x} = 0$$

$$Q_\theta = 0$$

2.3 STRESSES IN SKIN AND STIFFENERS

Once the stress resultants and couples are known, the corresponding maximum and minimum stresses of an isotropic shell can be computed by the relations

$$\sigma_x = \frac{1}{h} \hat{N}_x \pm \frac{6}{h^2} \hat{M}_x \quad (2.11)$$

$$\sigma_\theta = \frac{1}{h} \hat{N}_\theta \pm \frac{6}{h^2} \hat{M}_\theta \quad (2.12)$$

This is based on the assumption of a linear stress variation through the thickness given as

$$\sigma_i = \bar{\sigma}_i + z\sigma_i^b \quad (2.13)$$

where $\bar{\sigma}_i$ is a membrane stress and $z\sigma_i^b$ is the stress due to bending.

For a shell composed of ribs and stringers the expression for stresses cannot be obtained in the above manner. When stresses are required it is not necessary to first compute stress resultants and couples based on strains and curvature changes. However, the computation of stresses in the skin and stiffeners of an orthotropic shell in which the stress variation between stiffeners is small must be obtained by use of the basic equations for stress. These equations are given by

Stress in skin:

$$\sigma_x = \frac{E}{1 - \nu^2} [(\bar{\epsilon}_1 + \nu\bar{\epsilon}_2) + z(\chi_1 + \nu\chi_2)] \quad (2.14)$$

$$\sigma_\theta = \frac{E}{1 - \nu^2} [(\bar{\epsilon}_2 + \nu\bar{\epsilon}_1) + z(\chi_2 + \nu\chi_1)] \quad (2.15)$$

Stress in stringer:

$$\sigma_x = E(\bar{\epsilon}_1 + z\chi_1) \quad \text{where } |z| > \frac{h}{2} \quad (2.16)$$

Stress in rib:

$$\sigma_\theta = E(\bar{\epsilon}_2 + z\chi_2) \quad \text{where } |z| > \frac{h}{2} \quad (2.17)$$

In the digital program there is an option to printout the above strains.

Section 3
NUMERICAL ANALYSIS

The finite-difference method is used to solve the governing equations of a cylindrical shell segment with fixed edges. The scheme in this numerical method of solution is to replace the problem of a continuous coordinate system by a problem defined at a finite number of coordinate points. To accomplish this discretization, the continuous two dimensional (x, θ) domain of the cylinder is covered by a uniform rectangular net as shown in Fig. 9. Lattice points of this net which are within the domain \tilde{D}

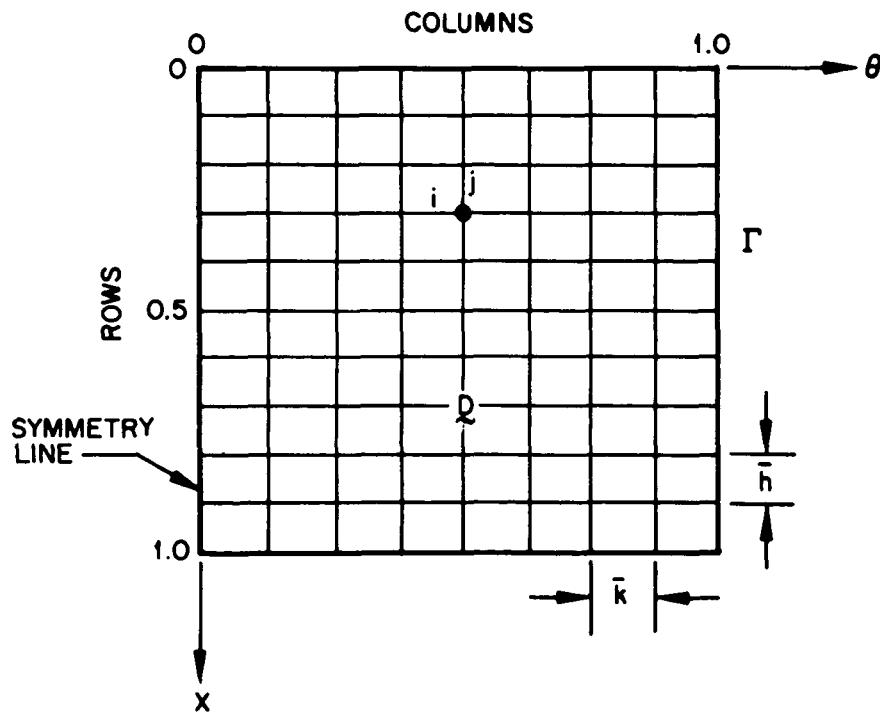


Fig. 9 Domain and Boundary of Cylindrical Shell Segment

are called mesh points and lattice points on the boundary curve Γ are called boundary points. At these lattice points the dependent variables (u, v, w) of the governing differential equations are replaced by the discrete values of u_i^j, v_i^j, w_i^j . The subscript i of u_i^j denotes the row number and corresponds to the x -coordinate while the superscript j denotes the column number and corresponds to the θ -coordinate. In general, the boundary curve does not coincide with the net. However, in the problem under consideration, this complication does not exist.

The difference equations which are a set of algebraic equations representing the governing equations and boundary conditions are formed by first approximating the derivatives at a given point by a function of the variable at neighboring points. These functions replace the derivatives of the governing equations. Thus, at each mesh point three algebraic equations can be written in terms of neighboring points. When the boundary conditions are accounted for in these equations the resulting set of simultaneous algebraic equations

$$\tilde{A}X = \tilde{B}$$

replaces the continuous problem. The solution of this set of algebraic equations can be accomplished by methods described in Vol. I.

3.1 APPROXIMATION OF DERIVATIVES

To transform the governing equations to difference form, the derivatives of u, v, w are expressed in terms of their values at neighboring mesh points. These derivatives are determined by a Taylor series approximation* for a rectangular net and are given by the following equations:

*"Investigation of Juncture Stress Fields in Multicellular Shell Structures," by E. Y. W. Tsui, F. A. Brogan, J. M. Massard, P. Stern, and C. E. Stuhlman, Technical Report M-03-63-1, Lockheed Missiles & Space Company, Sunnyvale, Calif., Feb 1964 - NASA CR-61050.

$$f_{,x} = 1/2\bar{h} \left(f_1^o - f_{-1}^o \right) \quad (3.1a)$$

$$f_{,xx} = 1/\bar{h}^2 \left(f_1^o - 2f_0^o + f_{-1}^o \right) \quad (3.1b)$$

$$f_{,xxx} = 1/2\bar{h}^3 \left(f_2^o - 2f_1^o + 2f_{-1}^o - f_{-2}^o \right) \quad (3.1c)$$

$$f_{,xxxx} = 1/\bar{h}^4 \left(f_2^o - 4f_1^o + 6f_0^o - 4f_{-1}^o + f_{-2}^o \right) \quad (3.1d)$$

$$f_{,\theta} = 1/2\bar{k} \left(f_o^1 - f_o^{-1} \right) \quad (3.1e)$$

$$f_{,\theta\theta} = 1/\bar{k}^2 \left(f_o^1 - 2f_o^o + f_o^{-1} \right) \quad (3.1f)$$

$$f_{,\theta\theta\theta} = 1/\bar{k}^3 \left(f_o^2 - 2f_o^1 + 2f_o^{-1} - f_o^{-2} \right) \quad (3.1g)$$

$$f_{,\theta\theta\theta\theta} = 1/\bar{k}^4 \left(f_o^2 - 4f_o^1 + 6f_o^o - 4f_o^{-1} + f_o^{-2} \right) \quad (3.1h)$$

$$f_{,x\theta} = 1/4\bar{h}\bar{k} \left(f_1^1 - f_{-1}^1 - f_1^{-1} + f_{-1}^{-1} \right) \quad (3.1i)$$

$$f_{,xx\theta} = 1/2\bar{h}^2\bar{k} \left(-2f_o^1 + 2f_o^{-1} + f_1^1 + f_{-1}^1 - f_1^{-1} - f_{-1}^{-1} \right) \quad (3.1j)$$

$$f_{,x\theta\theta} = 1/2\bar{h}\bar{k}^2 \left(-2f_1^o + 2f_{-1}^o + f_1^1 + f_{-1}^1 - f_{-1}^{-1} - f_{-1}^{-1} \right) \quad (3.1k)$$

$$f_{,xx\theta\theta} = 1/\bar{h}^2\bar{k}^2 \left(-2f_1^o - 2f_{-1}^o - 2f_o^1 - 2f_o^{-1} + f_1^1 + f_{-1}^1 + f_{-1}^{-1} + 4f_o^o \right) \quad (3.1l)$$

Lower order approximations to be used as noted

$$u_{,xxx} = 1/\bar{h}^3 \left[u_2^o - 3u_1^o + 3u_0^o - u_{-1}^o \right] \quad (3.1m)$$

$$u_{,xxx} = 1/\bar{h}^3 \left[u_1^o - 3u_0^o + 3u_{-1}^o - u_{-2}^o \right] \quad (3.1n)$$

$$u_{,\theta\theta\theta} = 1/\bar{k}^3 \left[u_o^1 - 3u_o^o + 3u_o^{-1} - u_o^{-2} \right] \quad (3.1o)$$

3.2 DIFFERENCE EQUATIONS

The formation of the difference equations is effected in a straightforward manner by substituting the appropriate expressions of Eqs. (3.1) into the governing equations [Eqs. (2.8)]. A complication arises when the equations are written at each mesh point and the boundary conditions are included in the appropriate equations; i.e., it is not possible to obtain a sufficient number of equations as unknowns. The application of central difference approximations requires that low-order approximations be used near the boundary so that this problem will not occur. It is noted that Eq. (2.8c) contains a third derivative of u with respect to x and a third derivative of v with respect to θ . The third derivatives as given by Eqs. (3.1c) and (3.1g) are in terms of five points hence is a higher order approximation (only four points are necessary for a third derivative). It was decided to incorporate the four point derivative [Eq. (3.1o)] for $v_{\theta\theta\theta}$ throughout while the third derivative in x would only be changed to Eqs. (3.1m) and (3.1n) when the equations are written one row from the boundary. After these substitutions, the governing equations in difference form at a point o, o are as follows:

$$A_1(u_1^o + u_{-1}^o) + A_2 u_o^o + A_3(v_1^1 + v_o^{-1}) + A_4(v_1^1 - v_{-1}^1 - v_1^{-1} + v_{-1}^{-1}) + A_5(w_1^o - w_{-1}^o) \\ + A_6(w_2^o - w_{-2}^o) = A_o^o \quad (3.2a)$$

$$B_1(u_1^1 - u_{-1}^1 - u_1^{-1} + u_{-1}^{-1}) + B_2(v_1^o + v_{-1}^o) + B_3 v_o^o + B_4(v_o^1 + v_o^{-1}) + B_5(w_o^1 - w_o^{-1}) \\ + B_6(w_1^1 + w_{-1}^1 - w_1^{-1} - w_{-1}^{-1}) + B_7(w_o^2 - w_o^{-2}) = B_o^o \quad (3.2b)$$

$$C_1(u_2^o - u_{-2}^o) + C_2(u_1^o - u_{-1}^o) + C_3 v_o^{-2} + 3C_3 v_o^o + C_4 v_o^1 + C_5 v_o^{-1} + C_6(v_1^1 + v_{-1}^1 - v_1^{-1} - v_{-1}^{-1}) \\ + C_7(w_2^o + w_{-2}^o) + C_8(w_1^o + w_{-1}^o) + C_9(w_o^1 + w_o^{-1}) + C_{10}(w_1^1 + w_{-1}^1 + w_1^{-1} + w_{-1}^{-1}) \\ + C_{11}(w_o^2 + w_o^{-2}) + C_{12} w_o^o = C_o^o \quad (3.2c)$$

where

$$A_1 = a_1/\bar{h}$$

$$A_2 = -2(A_1 + A_3)$$

$$A_3 = a_2/\bar{k}^2$$

$$A_4 = a_3/4\bar{h}\bar{k}$$

$$A_5 = a_4/2\bar{h} - 2A_6$$

$$A_6 = a_5/2\bar{h}^3$$

$$A_o^o = A$$

$$B_1 = A_4$$

$$B_2 = b_2/\bar{h}^2$$

$$B_3 = -2(B_2 + B_4)$$

$$B_4 = b_3/\bar{k}^2$$

$$B_5 = -2(B_6 + B_7) + b_6/2\bar{k}$$

$$B_6 = b_4/2\bar{h}^2\bar{k}$$

$$B_7 = b_5/2\bar{k}^3$$

$$B_o^o = B$$

$$C_1 = c_1/2\bar{h}^3$$

$$C_2 = c_2/2\bar{h} - 2C_1$$

$$C_3 = -c_4/\bar{k}^3$$

$$C_4 = -2C_6 - C_3 + c_5/2\bar{k}$$

$$C_5 = -C_4 - 4C_3$$

$$C_6 = c_3/2\bar{h}^2\bar{k}$$

$$C_7 = c_6/\bar{h}^4$$

$$C_8 = -(4C_7 + 2C_{10})$$

$$C_9 = -2(C_{10} + 2C_{11}) + c_9/\bar{k}^2$$

$$C_{10} = c_7/\bar{h}^2\bar{k}^2$$

$$C_{11} = c_8/\bar{k}^4$$

$$C_{12} = 6C_7 + 4C_{10} + 6C_{11} - c_{10} - 2C_9/\bar{k}^2$$

$$C_0^0 = C$$

The complete set of difference equations are obtained by writing these equations at each mesh point. Along lines of symmetry only two equations are necessary since one of the variables will be zero. After the incorporation of fixed edge boundary conditions, a sufficient number of equations for unknowns yields a set of simultaneous algebraic equations which are written in matrix form as

$$\tilde{A}\tilde{X} = \tilde{B}$$

Unless care is exercised in ordering the equations and unknowns, the square matrix \tilde{A} can be full. From the aspect of solving a large number of equations (Vol. I), the ordering is important. To establish an insight into the idea of the ordering employed, it is noticed from the difference expressions [Eqs. (3.1)] that the highest derivatives are in terms of at most two rows "above;" two rows "below;" two columns to the "left," and two columns to the "right" of a given meshpoint. If all the equations for a given column were written and stored in submatrix form, the unknowns would involve two columns to the "right" and "left." Thus, any column would involve, at most, five submatrices. The matrix \tilde{A} is accordingly partitioned in the manner shown below, where m is the number of columns in the finite difference net.

$$\tilde{A} = \begin{bmatrix} E_1 & F_1 & G_1 & \dots & \dots & 0 \\ D_2 & E_2 & F_2 & G_2 & & \\ C_3 & D_3 & F_3 & F_3 & G_3 & 0 \\ \vdots & \vdots & \vdots & & & \\ & & & & & G_{n-2} \\ & & & & & F_{m-1} \\ C_m & D_m & E_m & & & \end{bmatrix}$$

This matrix \tilde{A} is obtained by writing Eqs. (3.1) in \tilde{D} and not on the boundary Γ . The boundary and symmetry conditions have been used to eliminate certain equations. Fixed-edge boundary conditions are well-suited for this formulation, since they do not require complex algebraic expressions. Specifically, if Eqs. (3.1) are written one column from the boundary, then the submatrix F_m is zero ($u = v = w = 0$) and the submatrix G_m contains only w terms which are reflected into E_m due to the boundary condition that does not permit rotation. Along a symmetry line all terms are either reflected with the same or opposite sign. This fact accounts for the missing C_1 and D_1 matrices. Similar alterations are made in each matrix to account for boundary and symmetry conditions.

3.3 GRADING OF MESH

For a cylinder the solution in Ω is not uniform; at the boundary ($x = 0, 1$) the variation of the dependent variables occurs in a small interval. Such boundary behavior is characteristic of shell-type problems. For a cylinder, this behavior occurs only in the x -direction. To reveal the solution in sufficient detail in this region requires a small mesh spacing. To obtain this spacing without increasing the number of unknowns, some type of variable spacing must be incorporated. A rather simple method which accomplishes the small spacing and which does not destroy the form of the matrix A is called grading. In grading, the mesh spacing is changed by a factor of two from the parent mesh as shown in Fig. 10.

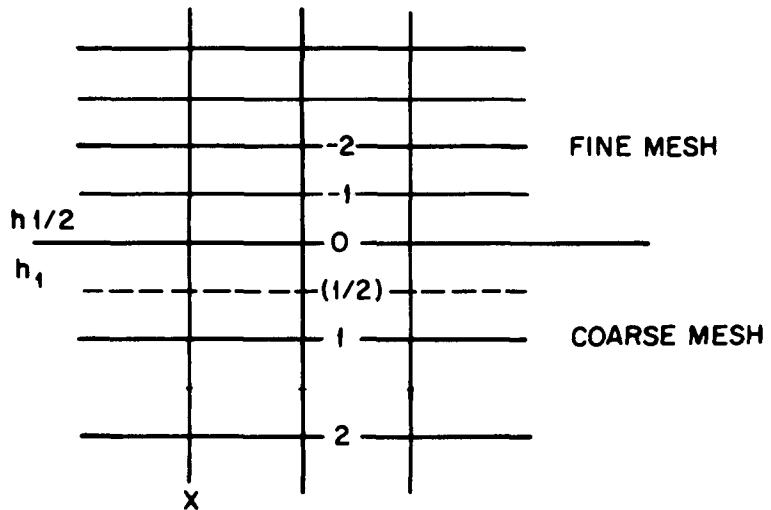


Fig. 10 Grading by Factor of Two

The major problem associated with grading mesh in the x-direction is obtaining the equation at the interface between different meshes. The following method is employed:

- Point o is considered as an ordinary point of the fine mesh and derivatives are expressed in terms of points $-2, -1, 0, 1/2, 1$ where point $1/2$ is obtained by interpolation of 1 and 0 .
- Points, such as 1 which is interior to the coarse net, are considered to be in the mesh space \bar{h} and use points $-2, 0, 1, 2, 3$.
- Points, such as -1 which is interior to the fine mesh, are considered to have mesh spacing $\bar{h}/2$ and use points $-3, -2, -1, 0, 1/2$.

Section 4

DIGITAL PROGRAM

4.1 GENERAL DESCRIPTION

The present program provides solutions for fixed-edge stiffened and unstiffened cylindrical panels under loads and changes of temperature. The method of solution consists basically in obtaining the displacement components u , v , and w at various discrete stations of the structure by finite-difference approximation (see Secs. 2 and 3). The corresponding strains and stresses may then be computed.

The program is designed to compute the fixed-edge forces due to intermediate loads or thermal gradient. However, displacements, strains, and stresses in the loaded region are also evaluated simultaneously. The following program options are available:

- Construction
 - (a) Isotropic
 - (b) Orthotropic
- Finite-Difference Mesh
 - (a) Uniform spacing
 - (b) Graded spacing in the x-direction
 - (c) Symmetry in the x-direction
- Loading conditions
 - (a) Uniform normal pressure
 - (b) Linear normal pressure
 - (c) Linear temperature gradient through the skin thickness

There are no restrictions on the geometrical dimensions of panels. However, the accuracy with which the basic differential equations are approximated may vary for different configurations of the cylinder.

The finite-difference mesh network is specified completely by prescribing the number of rows and columns exclusive of the boundaries, together with the grading options which have been chosen. Rows in the finite-difference mesh are parallel to the θ -axis, and columns are parallel to the x-axis. The number of rows may vary from 4 to 24 and the number of columns from 4 to 80. Thus, a maximum of 5760 unknowns can be solved. Greater accuracy near the boundaries can often be obtained by selecting grading. By this means, it is possible to use a mesh spacing at the boundary as little as 1/32 of that at the middle portion of the panel.

There are certain restrictions on the use of the grading option. When such an option is used, a separate input card is required to specify a mesh spacing exponent $MM(J)$ for each row J . The finite-difference equations written along row J , then use the mesh spacing $XH/2^{**MM(J)}$. This distance must be the least of the two distances from row J to the row above and the row below. XH is the basic input mesh spacing along the x-direction. For any row J , $MM(J)$ and $MM(J+1)$ must not differ by more than 1. Also, three consecutive rows cannot have three distinct exponents. $MM(J)$ may vary from 0 to 5.

Figure 11 shows the flow diagram of this program. The description of symbols, and input data are shown in Tables 1 and 2.

4.2 NUMERICAL EXAMPLE

The analysis of the cylindrical panel with ribs and stringers (orthotropic shell) shown in Fig. 12 will serve as an example to illustrate input data format and the type of information that can be obtained through use of the program described in this volume.

The example is for the loading option of uniform normal pressure ($P_z = \text{constant}$) in which two symmetry lines are present. The grading option is used to obtain a reasonable solution. The actual mesh spacing which yields solutions of desired accuracy must be obtained by exploring runs with different mesh spacing. The 14 row 20 column mesh used in this example proved satisfactory for the particular geometry.

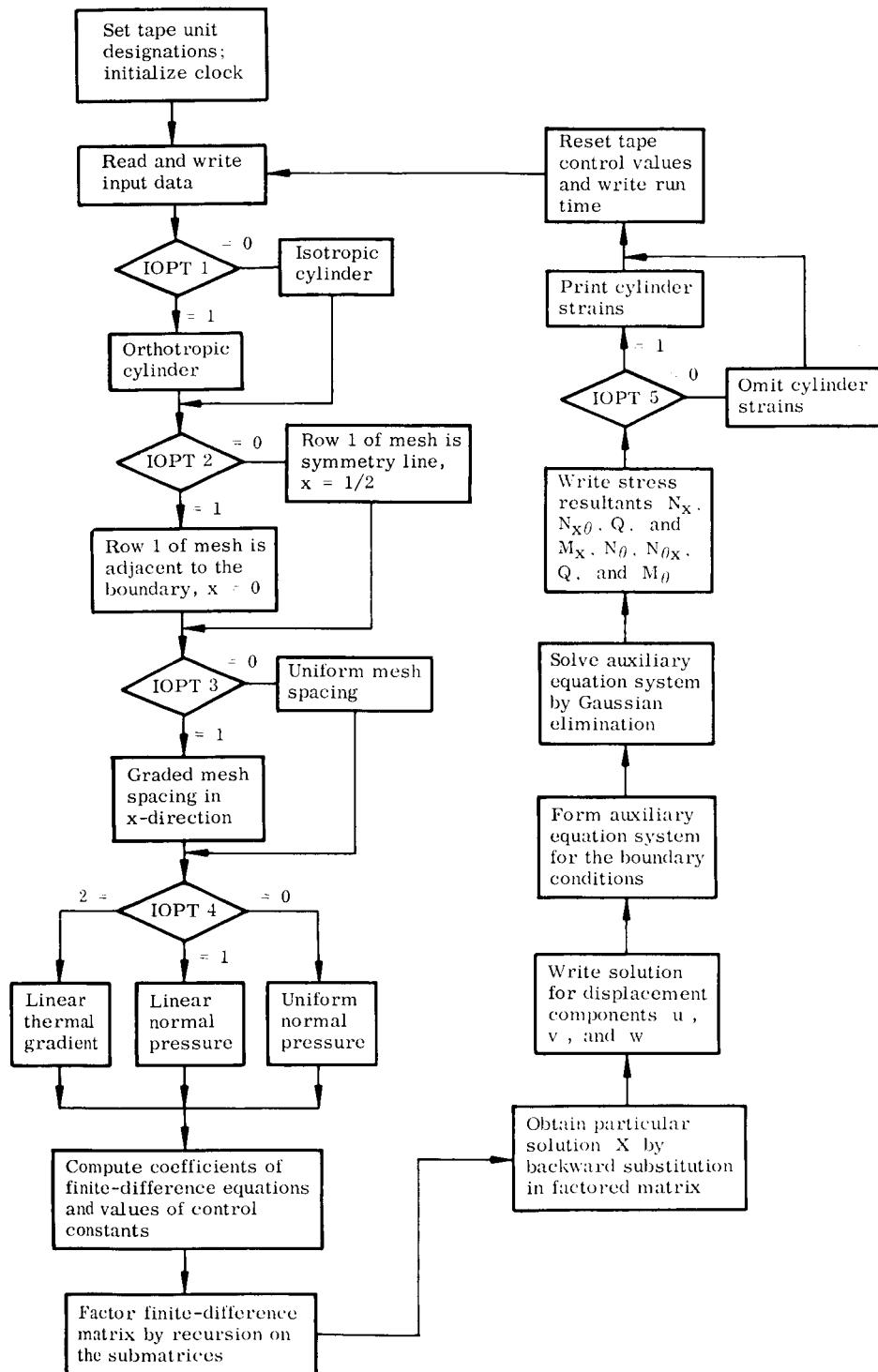


Fig. 11 Flow Chart

Table 1
DESCRIPTION OF SYMBOLS

Symbol		Description
$I_{\phi}PT1$	0	Isotropic cylinder
	1	Anisotropic cylinder
$I_{\phi}PT2$	0	Row 1 of mesh is symmetry line, $x = 0.5$
	1	Row 1 of mesh is adjacent to boundary
$I_{\phi}PT3$	0	Uniform mesh spacing
	1	Graded mesh spacing in x-direction
$I_{\phi}PT4$	0	Uniform normal pressure
	1	Linear normal pressure
	2	Linear temperature gradient through the skin thickness
$I_{\phi}PT5$	0	Omit cylinder strains
	1	Print cylinder strains
$R_{\phi}W$		Number of rows in the finite-difference mesh
$C_{\phi}L$		Number of columns in the finite-difference mesh
XH		Basic distance between rows in the mesh
XK		Basic distance between columns in the mesh
ZNU		Poisson's ratio
RH_{ϕ}		Shape factor ($R\theta_c/L$)
THC		Half angle of segment (see Fig. 3)
R1H		Radius to thickness ratio (R/h)
D1H		Depth of stiffener to thickness ratio (d/h)
D1B		Ratio of depth to spacing of stiffeners (d/b)
T1H		Ratio of stiffener width to skin thickness (t/h)
C1H		Ratio of eccentricity of stiffener to skin thickness (c/h)
P_{ϕ}		Initial value of linear pressure
DP		Linear pressure coefficient [$P(ZETA) = P_{\phi} + DP*ZETA$]
TE		External temperature
TI		Internal temperature

Table 1 (cont'd)

Symbol	Description
$T\phi$	Ambient temperature for zero stress
OC	Coefficient of thermal expansion
$MM(J)$, $J = 1, R\phi W$	Grading mesh constants; mesh spacing used for difference equations on row J is equal to $XH/2. ** MM(J)$
$MM(31)$	Number of rows to be plotted
$MM(32)$	Four row numbers for which plot output is desired
$MM(33)$	(U , V , W , N_x , M_x , N_θ , M_θ)
$MM(34)$	
$MM(35)$	
$CILBL(I, 1)$, $I = 1, 6$	Curve labels appearing on the plot output to identify the rows selected $CILBL(I, 1)$ corresponds to $MM(32)$; etc.
$CILBL(I, 2)$, $I = 1, 6$	
$CILBL(I, 3)$, $I = 1, 6$	
$CILBL(I, 4)$, $I = 1, 6$	

Table 2
INPUT DATA SEQUENCE AND FORMAT

Card	Fortran Symbol	Format
1	RECORD	72H
2	I _φ PT1 , I _φ PT2 , I _φ PT3 , I _φ PT4 , I _φ PT5	5I1
3	R _φ W , C _φ L , XH , XK	6E12.8
4	ZNU , RH _φ , THC , R1H	6E12.8
5 ^(a)	D1B , D1H , T1H , C1H	6E12.8
6 ^(b)	P _φ , DP	6E12.8
7 ^(c)	TE , TI , T _φ , OC	6E12.8
8 ^(d)	MM(J) , J = 1 , R _φ W	35I2
9	MM(J) , J = 31 , 35	5I2
10 ^(e)	CILBL(I , 1) , I = 1,6	6A6
11 ^(e)	CILBL(I , 2) , I = 1,6	6A6
12 ^(e)	CILBL(I , 3) , I = 1,6	6A6
13 ^(e)	CILBL(I , 4) , I = 1,6	6A6

- (a) Omitted unless I_φPT1=1
- (b) Omitted unless I_φPT4=1
- (c) Omitted unless I_φPT4=2
- (d) Omitted unless I_φPT3=1
- (e) Omitted if MM(31)=0

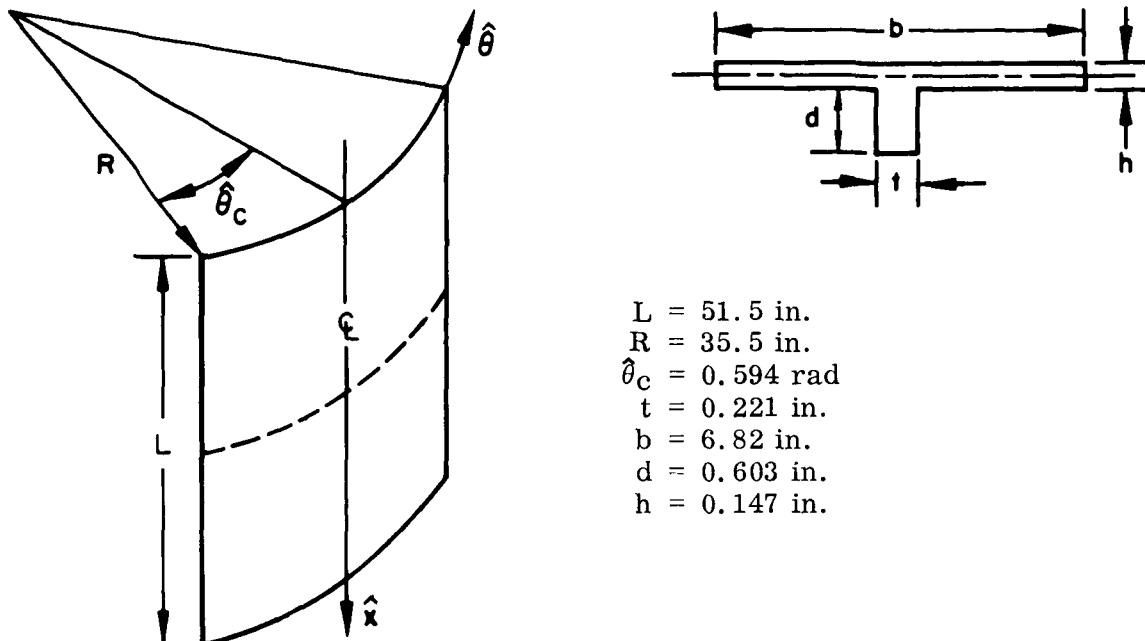


Fig. 12 Cylindrical Panel for the Example

Values of input quantities are given in Table 3 and a listing of the corresponding input data cards is presented in Table 4. For convenience, the coordinate value x corresponding to the row number is as follows:

$x = 0$	- Row 15	$x = 0.1$ - Row 7
$x = 0.00625$	- Row 14	$x = 0.15$ - Row 6
$x = 0.0125$	- Row 13	$x = 0.2$ - Row 5
$x = 0.01375$	- Row 12	$x = 0.25$ - Row 4
$x = 0.25$	- Row 11	$x = 0.3$ - Row 3
$x = 0.0375$	- Row 10	$x = 0.4$ - Row 2
$x = 0.05$	- Row 9	$x = 0.5$ - Row 1
$x = 0.075$	- Row 8	

The computer results are in the form of printed digital values and plots. The sample printed output given in Table 5 presents displacements components (u , v , w) stress resultants (N_x , N_θ , $N_{x\theta}$, $N_{\theta x}$, M_x , M_θ , $M_{x\theta}$, Q_x , Q_θ), and boundary stress resultants (N_{tan} , N_{norm} , Q , M). [Note that these quantities are in non-dimensional form.] Plotted output includes displacement components (u , v , w)

Table 3
INPUT VALUES FOR THE EXAMPLE

Symbol	Value	Symbol	Value
I _φ PT1	1	MM(4)	1
I _φ PT2	0	MM(5)	1
I _φ PT3	1	MM(6)	1
I _φ PT4	0	MM(7)	2
I _φ PT5	0	MM(8)	2
R _φ W	14.0	MM(9)	3
C _φ L	20.0	MM(10)	3
XH	0.1	MM(11)	4
XK	0.05	MM(12)	4
ZNU	0.3	MM(13)	4
RH _φ	0.4	MM(14)	4
THC	0.594	MM(31)	04
R1H	242.0	MM(32)	01
D1H	4.1	MM(33)	06
D1B	0.0884	MM(34)	09
TH	1.5	MM(35)	13
C1H	2.55	CILBL(I, 1)I=1,6	x = 0.5, symmetry line
P _φ , DP, TE, } TI, T _φ , OC }	Not required	CILBL(I, 2)I=1,6	x = 0.15, Row 6
MM(1)	0	CILBL(I, 3)I=1,6	x = 0.05, Row 9
MM(2)	0	CILBL(I, 4)I=1,6	x = 0.0125, Row 13
MM(3)	1		

Table 4
INPUT DATA IN REQUIRED FORMAT

and stress resultants (N_x , N_θ , M_x , M_θ) along four specified rows, and boundary stress resultants (N_{tan} , N_{norm} , Q , M), along the boundary curve. This plotted output is shown in Figs. 13a through f.

Table 5
SAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE
CYLINDER DISPLACEMENT COMPONENTS (U, V, W)

COL	ROW	U	V	W
1,	17	0.	0.	4.639469E-02
1,	16	1.867072E-03	0.	1.068088E-02
1,	15 BOUNDARY	0.	0.	0.
1,	14	-1.284630E-03	0.	1.068088E-02
1,	13	-1.970761E-03	0.	3.757576E-02
1,	12	-2.146589E-03	0.	7.598753E-02
1,	11	-1.886211E-03	0.	1.216492E-01
1,	10	-6.268877E-04	0.	2.197437E-01
1,	9	7.368894E-04	0.	3.220843E-01
1,	8	3.060286E-03	0.	5.268981E-01
1,	7	4.810151E-03	0.	6.942787E-01
1,	6	5.546042E-03	0.	9.627899E-01
1,	5	4.607340E-03	0.	1.094271E 00
1,	4	3.100316E-03	0.	1.154270E 00
1,	3	1.885045E-03	0.	1.187821E 00
1,	2	2.771763E-04	0.	1.238483E 00
1,	1 SMTY LINE	0.	0.	1.264708E 00
2,	17	0.	0.	4.643006E-02
2,	16	1.867419E-03	2.395163E-04	1.068889E-02
2,	15 BOUNDARY	0.	0.	0.
2,	14	-1.284551E-03	-2.394961E-04	1.068889E-02
2,	13	-1.970173E-03	-4.794142E-04	3.760434E-02
2,	12	-2.145067E-03	-7.201626E-04	7.604596E-02
2,	11	-1.883325E-03	-9.621046E-04	1.217435E-01
2,	10	-6.204223E-04	-1.449002E-03	2.199158E-01
2,	9	7.474678E-04	-1.941692E-03	3.223415E-01
2,	8	3.080177E-03	-2.936370E-03	5.273326E-01
2,	7	4.840821E-03	-3.934191E-03	6.948892E-01
2,	6	5.599483E-03	-5.904731E-03	9.637371E-01
2,	5	4.683598E-03	-7.734518E-03	1.095230E 00
2,	4	3.190010E-03	-9.321540E-03	1.154876E 00
2,	3	1.973994E-03	-1.056890E-02	1.187832E 00
2,	2	3.367156E-04	-1.233849E-02	1.237066E 00
2,	1 SMTY LINE	0.	-1.315655E-02	1.262910E 00

Table 5 (cont'd)

CYLINDER STRESS RESULTANTS.					
ROW	COL	NX	NTHETA	NXTHETA	NTHETAX
1,	1	2.6528E-01	1.0063E 00	0.	0.
1,	2	2.6489E-01	1.0060E 00	-0.	0.
1,	3	2.6373E-01	1.0054E 00	-0.	0.
1,	4	2.6190E-01	1.0043E 00	-0.	0.
1,	5	2.5951E-01	1.0028E 00	-0.	0.
1,	6	2.5669E-01	1.0010E 00	-0.	0.
1,	7	2.5358E-01	9.9882E-01	-0.	0.
1,	8	2.5027E-01	9.9642E-01	-0.	0.
1,	9	2.4685E-01	9.9382E-01	-0.	0.
1,	10	2.4340E-01	9.9109E-01	-0.	0.
1,	11	2.4003E-01	9.8826E-01	-0.	0.
1,	12	2.3687E-01	9.8536E-01	-0.	0.
1,	13	2.3402E-01	9.8239E-01	-0.	0.
1,	14	2.3161E-01	9.7929E-01	-0.	0.
1,	15	2.2972E-01	9.7601E-01	-0.	0.
1,	16	2.2842E-01	9.7241E-01	-0.	0.
1,	17	2.2771E-01	9.6839E-01	-0.	0.
1,	18	2.2759E-01	9.6377E-01	-0.	0.
1,	19	2.2805E-01	9.5843E-01	-0.	0.
1,	20	2.2911E-01	9.5224E-01	-0.	0.
1,	21	2.3095E-01	9.4519E-01	0.	0.
2,	1	2.6197E-01	1.0076E 00	0.	0.
2,	2	2.6171E-01	1.0074E 00	-1.8205E-03	-1.8229E-03
2,	3	2.6093E-01	1.0072E 00	-3.5577E-03	-3.5624E-03
2,	4	2.5963E-01	1.0068E 00	-5.1359E-03	-5.1426E-03
2,	5	2.5788E-01	1.0063E 00	-6.4952E-03	-6.5036E-03
2,	6	2.5573E-01	1.0058E 00	-7.5968E-03	-7.6066E-03
2,	7	2.5322E-01	1.0053E 00	-8.4271E-03	-8.4377E-03
2,	8	2.5037E-01	1.0048E 00	-8.9975E-03	-9.0083E-03
2,	9	2.4720E-01	1.0043E 00	-9.3438E-03	-9.3542E-03
2,	10	2.4379E-01	1.0036E 00	-9.5235E-03	-9.5329E-03
2,	11	2.4025E-01	1.0026E 00	-9.6109E-03	-9.6186E-03
2,	12	2.3673E-01	1.0012E 00	-9.6908E-03	-9.6962E-03
2,	13	2.3341E-01	9.9932E-01	-9.8495E-03	-9.8522E-03
2,	14	2.3049E-01	9.9680E-01	-1.0165E-02	-1.0165E-02
2,	15	2.2816E-01	9.9355E-01	-1.0697E-02	-1.0693E-02
2,	16	2.2659E-01	9.8946E-01	-1.1475E-02	-1.1470E-02
2,	17	2.2586E-01	9.8447E-01	-1.2497E-02	-1.2489E-02
2,	18	2.2602E-01	9.7851E-01	-1.3713E-02	-1.3704E-02
2,	19	2.2698E-01	9.7157E-01	-1.5032E-02	-1.5024E-02
2,	20	2.2853E-01	9.6362E-01	-1.6319E-02	-1.6314E-02

Table 5 (cont'd)

CYLINDER STRESS RESULTANTS.						
ROW	COL	MX	MTHETA	MXTHETA	QX	QTHETA
1,	1	5.0592E-05	-1.0763E-03	0.	-0.	0.
1,	2	5.0581E-05	-1.0724E-03	-0.	-0.	7.5491E-05
1,	3	5.0443E-05	-1.0610E-03	-0.	-0.	1.1768E-04
1,	4	4.9633E-05	-1.0426E-03	-0.	-0.	1.0103E-04
1,	5	4.7371E-05	-1.0178E-03	-0.	-0.	-4.6353E-06
1,	6	4.3004E-05	-9.8810E-04	-0.	-0.	-2.3431E-04
1,	7	3.6363E-05	-9.5559E-04	-0.	-0.	-6.2220E-04
1,	8	2.7829E-05	-9.2348E-04	-0.	-0.	-1.1931E-03
1,	9	1.8175E-05	-8.9564E-04	-0.	-0.	-1.9557E-03
1,	10	8.3288E-06	-8.7630E-04	-0.	-0.	-2.8997E-03
1,	11	-8.0897E-07	-8.6968E-04	-0.	-0.	-3.9963E-03
1,	12	-8.4913E-06	-8.7978E-04	-0.	-0.	-5.2001E-03
1,	13	-1.4187E-05	-9.1013E-04	-0.	-0.	-6.4521E-03
1,	14	-1.7612E-05	-9.6375E-04	-0.	-0.	-7.6835E-03
1,	15	-1.8745E-05	-1.0430E-03	-0.	-0.	-8.8188E-03
1,	16	-1.7856E-05	-1.1498E-03	-0.	-0.	-9.7790E-03
1,	17	-1.5573E-05	-1.2853E-03	-0.	-0.	-1.0484E-02
1,	18	-1.3013E-05	-1.4505E-03	-0.	-0.	-1.0857E-02
1,	19	-1.1997E-05	-1.6457E-03	-0.	-0.	-1.0824E-02
1,	20	-1.5405E-05	-1.8712E-03	-0.	-0.	-1.0318E-02
1,	21	-2.7679E-05	-2.1270E-03	0.	-0.	-9.2818E-03
2,	1	3.1744E-05	-1.0950E-03	0.	3.1202E-05	0.
2,	2	3.1315E-05	-1.0920E-03	2.3703E-06	2.7395E-05	-1.2359E-05
2,	3	3.0085E-05	-1.0865E-03	4.6311E-06	1.6108E-05	-5.0195E-05
2,	4	2.8769E-05	-1.0762E-03	6.6772E-06	-2.9826E-06	-4.3526E-06
2,	5	2.8420E-05	-1.0578E-03	8.4078E-06	-3.0281E-05	1.2611E-04
2,	6	2.9992E-05	-1.0282E-03	9.7279E-06	-6.5603E-05	2.3306E-04
2,	7	3.3853E-05	-9.8898E-04	1.0550E-05	-1.0750E-04	1.5135E-04
2,	8	3.9718E-05	-9.4522E-04	1.0800E-05	-1.5314E-04	-2.2747E-04
2,	9	4.6857E-05	-9.0355E-04	1.0420E-05	-1.9870E-04	-9.3213E-04
2,	10	5.4352E-05	-8.7049E-04	9.3820E-06	-2.3985E-04	-1.9362E-03
2,	11	6.1281E-05	-8.5183E-04	7.6981E-06	-2.7236E-04	-3.1832E-03
2,	12	6.6815E-05	-8.5246E-04	5.4324E-06	-2.9248E-04	-4.6021E-03
2,	13	7.0238E-05	-8.7644E-04	2.7123E-06	-2.9738E-04	-6.1142E-03
2,	14	7.0938E-05	-9.2716E-04	-2.6545E-07	-2.8542E-04	-7.6387E-03
2,	15	6.8365E-05	-1.0075E-03	-3.2344E-06	-2.5619E-04	-9.0954E-03
2,	16	6.2018E-05	-1.1202E-03	-5.8658E-06	-2.1043E-04	-1.0408E-02
2,	17	5.1462E-05	-1.2680E-03	-7.7813E-06	-1.4955E-04	-1.1505E-02
2,	18	3.6410E-05	-1.4542E-03	-8.5713E-06	-7.4986E-05	-1.2320E-02
2,	19	1.6901E-05	-1.6830E-03	-7.8134E-06	1.2648E-05	-1.2790E-02
2,	20	-6.4005E-06	-1.9598E-03	-5.0897E-06	1.1411E-04	-1.2850E-02

Table 5 (concl'd)

BOUNDARY STRESS RESULTANTS.

ROW	COL	NTAN	NNORM	Q	M
15,	1	0.	1.3740E-01	-9.7865E-02	-4.5881E-03
15,	2	-9.9253E-03	1.3764E-01	-9.7941E-02	-4.5917E-03
15,	3	-1.9856E-02	1.3839E-01	-9.8159E-02	-4.6019E-03
15,	4	-2.9786E-02	1.3972E-01	-9.8498E-02	-4.6168E-03
15,	5	-3.9693E-02	1.4176E-01	-9.8921E-02	-4.6334E-03
15,	6	-4.9522E-02	1.4464E-01	-9.9379E-02	-4.6478E-03
15,	7	-5.9183E-02	1.4854E-01	-9.9808E-02	-4.6548E-03
15,	8	-6.8540E-02	1.5361E-01	-1.0013E-01	-4.6485E-03
15,	9	-7.7403E-02	1.5998E-01	-1.0026E-01	-4.6221E-03
15,	10	-8.5528E-02	1.6772E-01	-1.0010E-01	-4.5683E-03
15,	11	-9.2609E-02	1.7679E-01	-9.9514E-02	-4.4787E-03
15,	12	-9.8280E-02	1.8704E-01	-9.8377E-02	-4.3448E-03
15,	13	-1.0212E-01	1.9807E-01	-9.6528E-02	-4.1573E-03
15,	14	-1.0369E-01	2.0921E-01	-9.3773E-02	-3.9062E-03
15,	15	-1.0253E-01	2.1932E-01	-8.9879E-02	-3.5816E-03
15,	16	-9.8260E-02	2.2662E-01	-8.4554E-02	-3.1739E-03
15,	17	-9.0609E-02	2.2831E-01	-7.7435E-02	-2.6758E-03
15,	18	-7.9514E-02	2.2004E-01	-6.8014E-02	-2.0857E-03
15,	19	-6.4861E-02	1.9454E-01	-5.5224E-02	-1.4113E-03
15,	20	-4.7527E-02	1.4206E-01	-3.7239E-02	-6.8500E-04
15,	21	5.2778E-06	0.	-3.8332E-04	-0.
15,	21	-0.	0.	-1.1373E-03	-0.
14,	21	4.7472E-02	9.5795E-02	-9.2014E-03	-1.5851E-04
13,	21	3.6854E-02	9.3074E-02	-1.4575E-02	-2.6829E-04
12,	21	5.7328E-02	1.5213E-01	-2.0018E-02	-4.8001E-04
11,	21	5.8159E-02	1.7713E-01	-2.3955E-02	-6.6753E-04
10,	21	7.8445E-02	2.6504E-01	-3.1613E-02	-1.0926E-03
9,	21	8.8080E-02	3.3439E-01	-3.4840E-02	-1.4496E-03
8,	21	1.0422E-01	4.7853E-01	-3.9606E-02	-2.0841E-03
7,	21	1.0879E-01	5.9205E-01	-3.6757E-02	-2.4067E-03
6,	21	1.0117E-01	7.7824E-01	-2.7202E-02	-2.6660E-03
5,	21	7.8073E-02	8.6801E-01	-1.8730E-02	-2.5554E-03
4,	21	5.6605E-02	9.0768E-01	-1.3892E-02	-2.3926E-03
3,	21	4.0230E-02	9.2840E-01	-1.1751E-02	-2.2986E-03
2,	21	1.7408E-02	9.5526E-01	-1.1864E-02	-2.2927E-03
1,	21	-0.	9.4519E-01	-9.2818E-03	-2.1270E-03

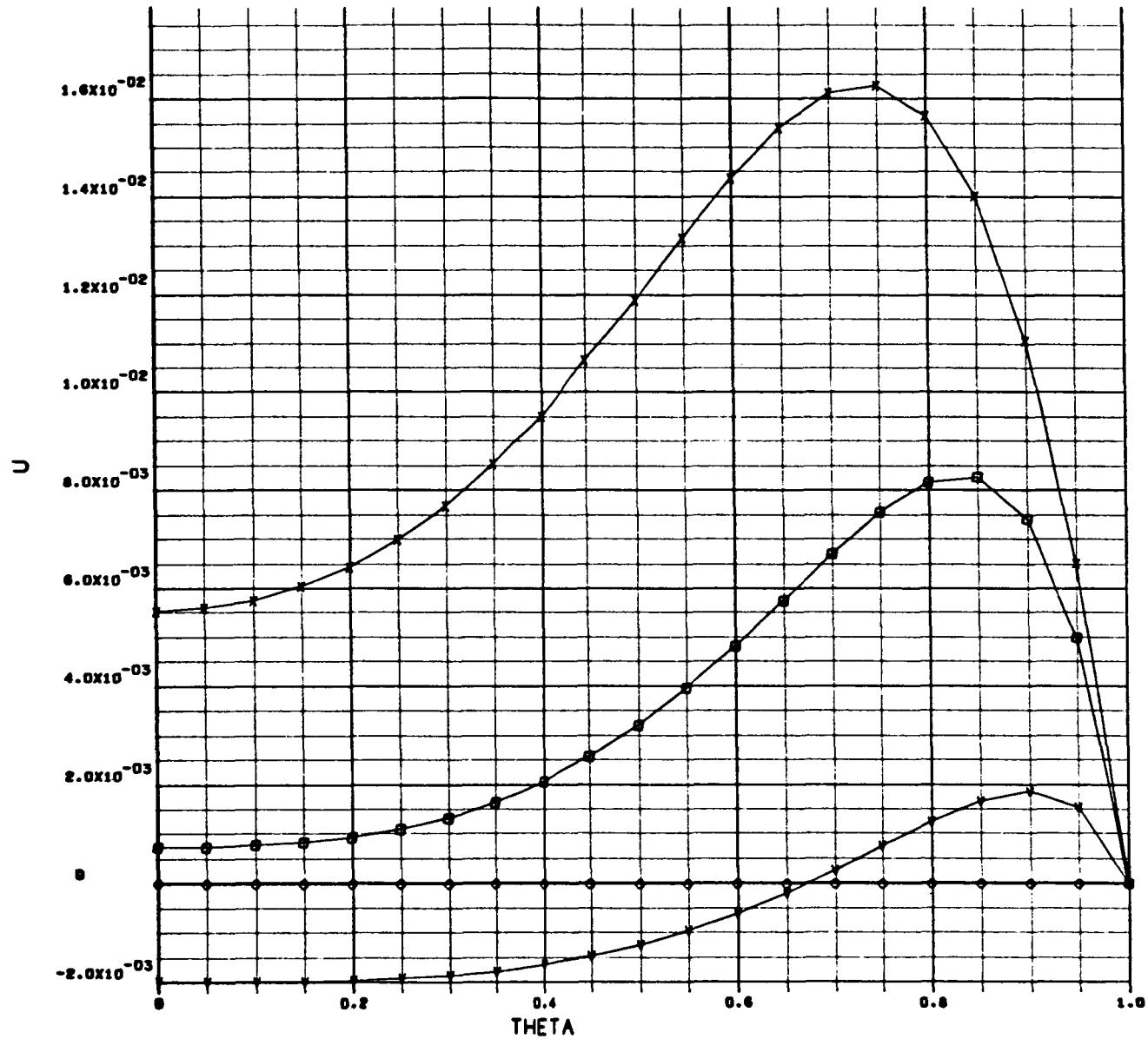
EXAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE

• X=.5 , SYMMETRY LINE

X X=.15 , ROW 6

◎ X=.05 , ROW 9

▼ X=.0125 , ROW 13



a - Cylinder Displacement Components

Fig. 13 Output Plot for Sample Problem Orthotropic Cylinder Normal Pressure

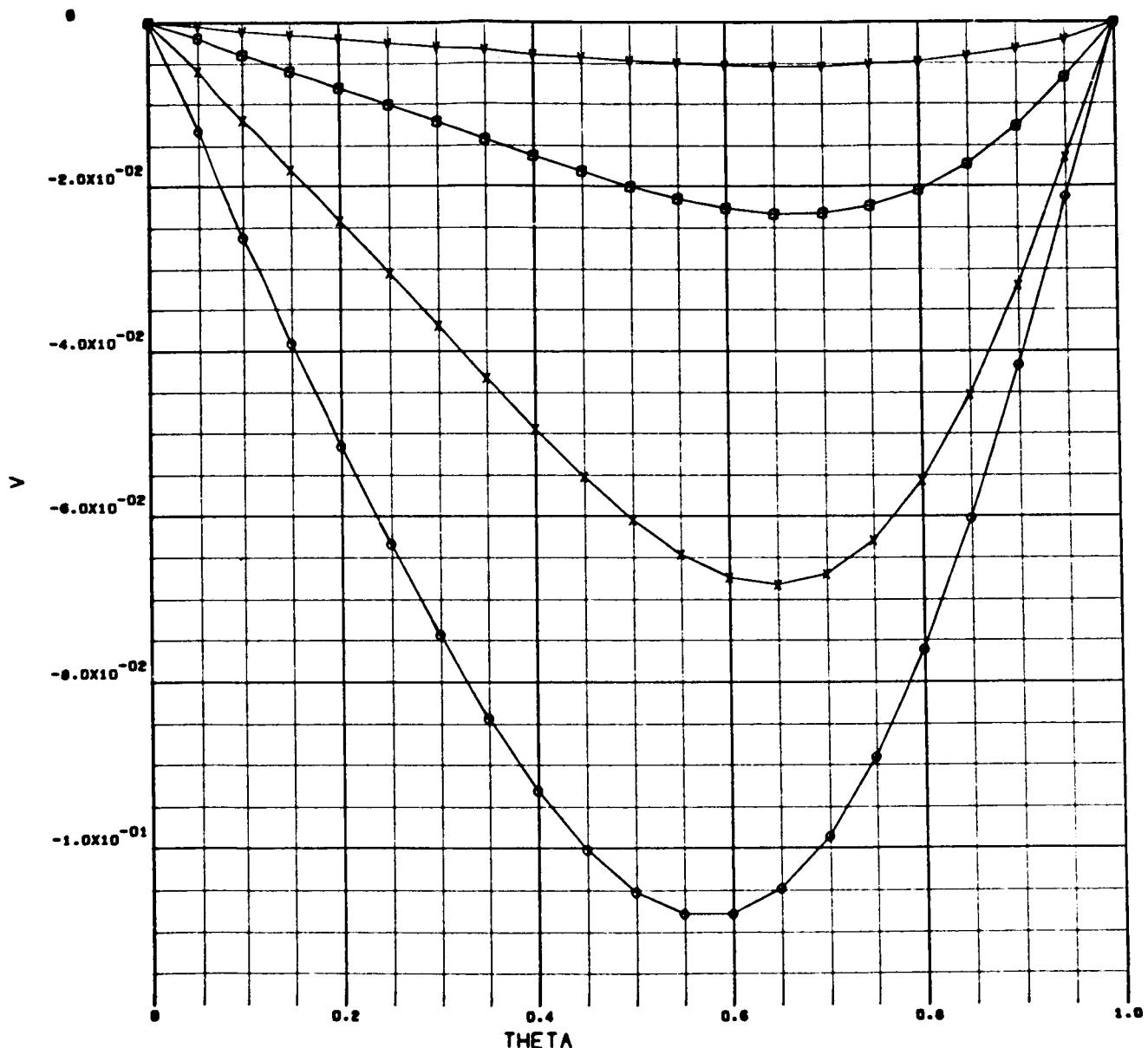
EXAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE

● X=.5 ,SYMMETRY LINE

X X=.15 , ROW 6

● X=.05 , ROW 9

Y X=.0125 , ROW 13



b - Cylinder Displacement Components

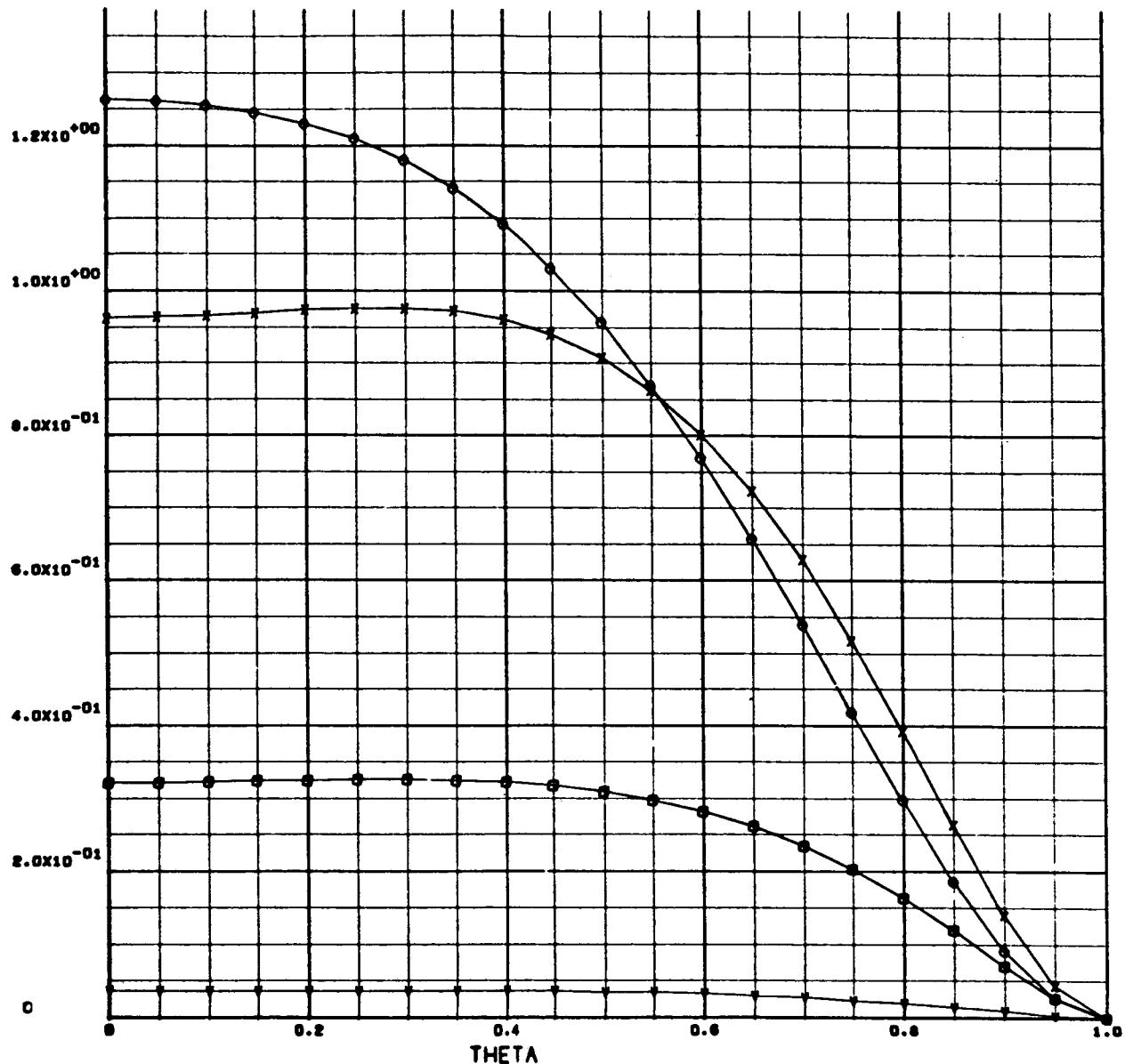
EXAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE

● X=.5 ,SYMMETRY LINE

X X=.15 , ROW 6

■ X=.05 , ROW 9

▼ X=.0125 , ROW 13



c - Cylinder Displacement Components

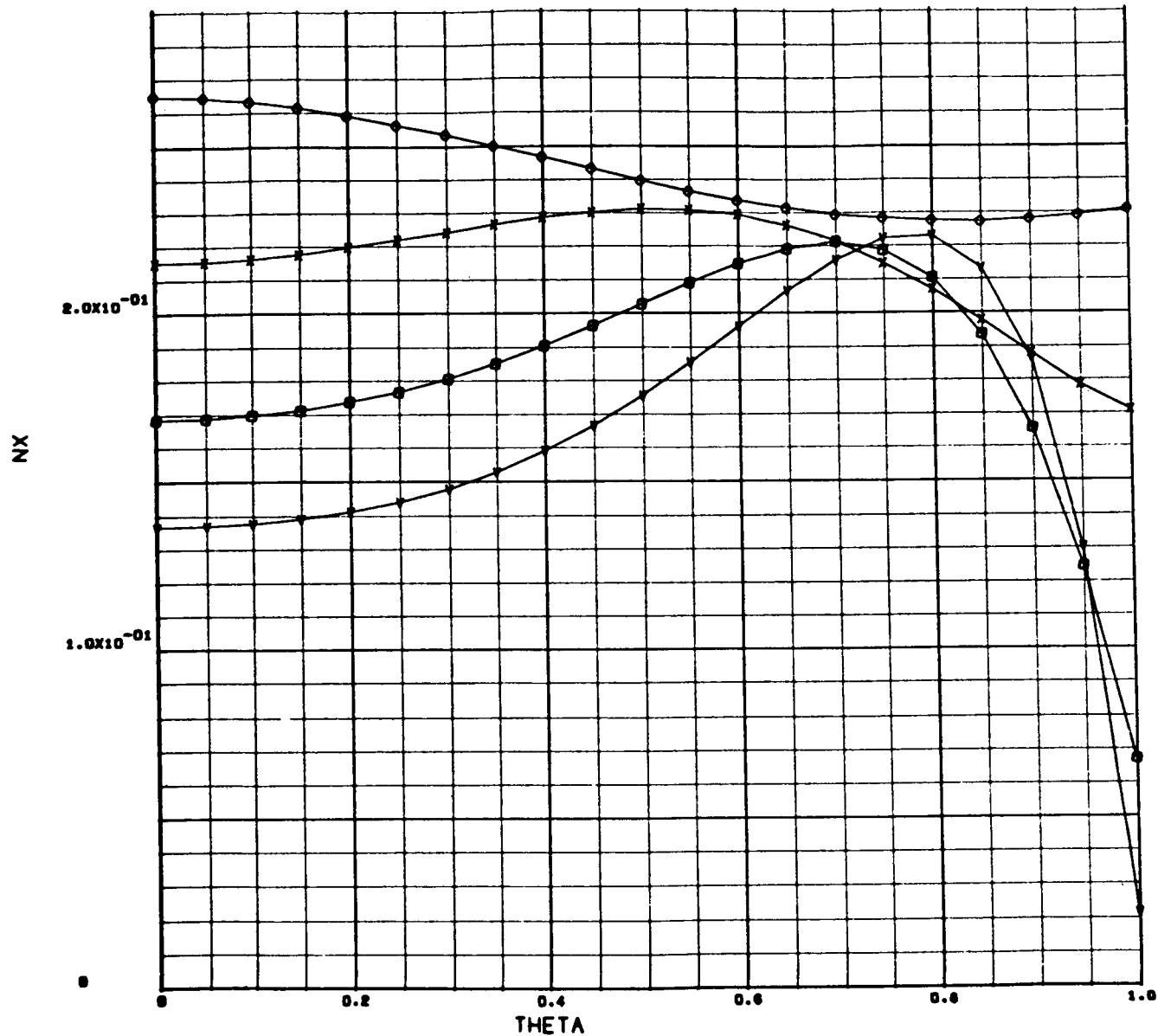
EXAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE

● X=.5 ,SYMMETRY LINE

X X=.15 , ROW 6

■ X=.05 , ROW 9

▼ X=.0125 , ROW 13



d - Cylinder Stress Resultants

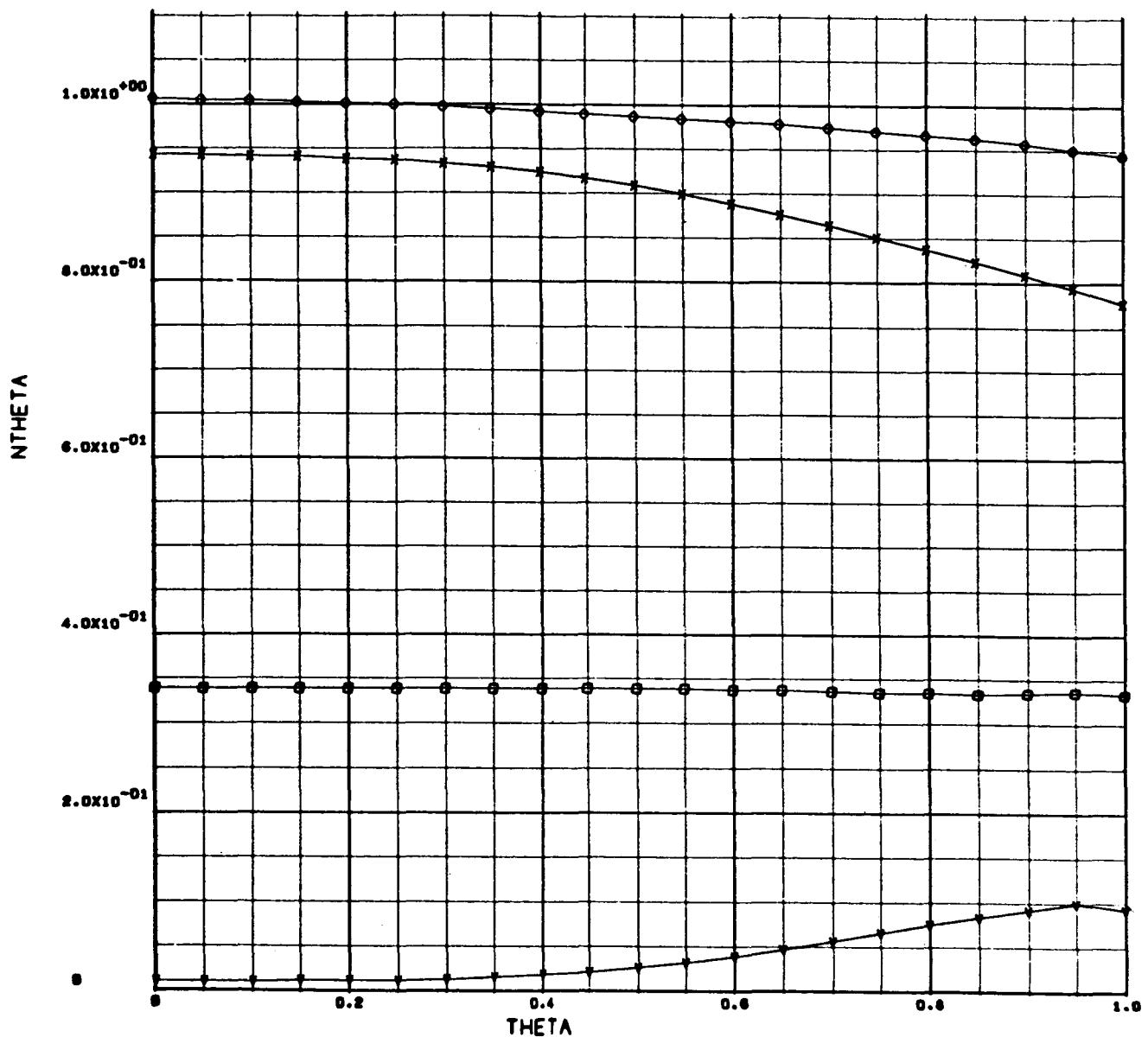
EXAMPLE PROBLEM ORTHOTROPIC CYLINDER NORMAL PRESSURE

● X=.5 ,SYMMETRY LINE

■ X=.05 , ROW 9

× X=.15 , ROW 6

▼ X=.0125 , ROW 13



e - Cylinder Stress Resultants

EXAMPLE PROBLEM: ORTHOTROPIC CYLINDER NORMAL PRESSURE

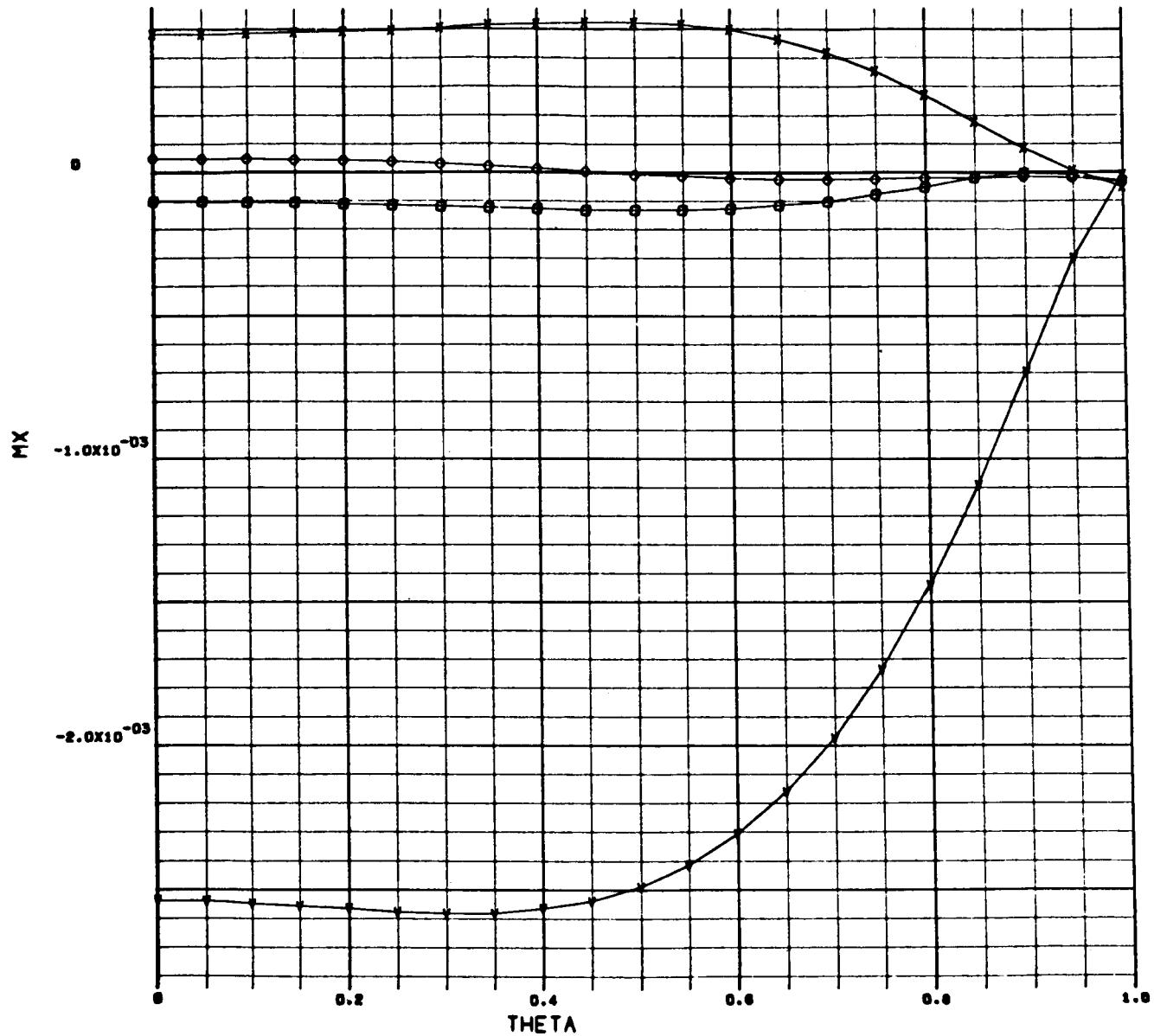
BAH300
006 000

● X=.5 ,SYMMETRY LINE

■ X=.05 , ROW 9

X X=.15 , ROW 6

Y X=.0125 , ROW 13



f - Cylinder Stress Resultants

4.3 LISTING OF THE PROGRAM

The complete source program is given in Table 6.

Table 6

STUDY OF JUNCTURE STRESS FIELDS: CYLINDRICAL SEGMENT

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*   CHAIN (1,2)                                A1P00010
*   FORTRAN                                     A1P00020
*   LIST                                         A1P00030
*   CAMP CYLINDRICAL SEGMENT - STUDY OF JUNCTURE STRESS FIELDS- A1P00040
C CONTRACT NAS 8-114800 TO NASA G.C. MARSHALL SPACE FLIGHT CENTER A1P00050
C HUNTSVILLE, ALABAMA                           A1P00060
C BY SOLID MECHANICS, AEROSPACE SCIENCES LABORATORY, 52-20. A1P00070
C LOCKHEED MISSILES AND SPACE COMPANY, PALO ALTO CALIF A1P00080
A1P00090
A1P00100
A1P00110
A1P00120
A1P00130
A1P00140
A1P00150
A1P00160
A1P00170
A1P00180
A1P00190
A1P00200
A1P00210
A1P00220
A1P00230
A1P00240
A1P00250
A1P00260
A1P00270
A1P00280
A1P00290
A1P00300
A1P00310
A1P00320
A1P00330

C THIS PROGRAM IS FOR USE ONLY ON FORTRAN II, VERSION II
C GRADED MESH IN X DIRECTION. MAXIMUM OF 24 ROWS, 80 COLUMNS
C DIMENSION COMENT (12)
COMMON NDIM, NROW, NCOL, MM, MFLAG, B, C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P00200
1 C9, C10, C11, C12
COMMON TMP, YK, YL
COMMON ZNU, RHO, THC, R1H, D1R, D1H, T1H, C1H
COMMON XH, XK, QDXH, P0, DP
COMMON TE, T1, T0, TC, TD, TDLD
COMMON ROT
COMMON EX, ENU, FTX, SC, DX, DTX, DN1
COMMON TIME, TIM2, KTIME
DIMENSION MM(40)
DIMENSION TMP(5184), YK(5184), YL(4,4,324)
DIMENSION DISPT(9), STRSS(9), ORDABL(6,11), ABLE(6), CILSL(6,4)
DIMENSION ABSC(80), AORD(25), BOND(9)

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DIMENSION CBLRL (6, 4) A1P00340
DIMENSION AX[BL (6) A1P00350
DIMENSION RFCORD (12) A1P00360
DIMENSION RNT (80) A1P00370
DIMENSION TTT (89), T7(89), XX(89), XX1(89), 7(89), 71(89), 72(89) A1P00380
1 FORMAT (1H1, 12A6) A1P00390
2 FORMAT (6F12.8) A1P00400
3 FORMAT (1H1) A1P00410
4 FORMAT (35I2) A1P00420
5 FORMAT (1H1, 3X, 3HCOL , 15, 8I14/4H ROW) A1P00430
6 FORMAT ( 14, 1P9E14•6) A1P00440
7 FORMAT (10I1 ) A1P00450
8 FORMAT (12A6) A1P00460
14 FORMAT (26H0 FINITE DIFFERENCE MFSH• I2, 7H ROWS, 12, 27H COLUMNNSA1P00470
1• MESH SPACING• H= F15•7, 4H, K= F15•7 ) A1P00480
15 FORMAT (23H INPUT CONSTANTS• NU=1PF14•6, 7H, RHO=1PF14•6, 10H, TAIP00490
THETAC=1PF14•6, 7H, R7H IPFT4•6 ) A1P00500
16 FORMAT (39H0 A DIAGONAL SURMATRIX IS SINGULAR, J= A1P00510
17 FORMAT (47H0 CYLINDER DISPLACEMENT COMPONENTS (U, V, W) / A1P00520
1 INHO COL ROW , I7X, THU, I5X, THV, I5X, THW ) A1P00530
18 FORMAT (53H ROW COL NX NTHETA NXTHETA NTHETAX A1P00540
19 FORMAT (125HOROW COL EPSX EPST GAMMA R*ALP00550
IXI R*X12 OMEGAX OMEGAT PAIP00560
2HI )
20 FORMAT (31H0 CYLINDER STRESS RESULTANTS. ) A1P00570
920 FORMAT (29H GRADED MFSH IN X DIRECTION. / 2X, 4H ROW, 15I3 / A1P00580
1 6X, 15I8 )
921 FORMAT (9H SPACING • 15( 7H XH/2**11) / 9X, 15( 7H XH/2**11) ) A1P00590
922 FORMAT (27H ROW 1 IS A SYMMETRY LINE ) A1P00600
923 FORMAT (32H ROW 1 IS ADJACENT TO BOUNDARY ) A1P00610
930 FORMAT (1H 13, 1H, 13, 10X, 1P3E16•6 ) A1P00630
931 FORMAT (1H 13, 1H, 13, 10H BOUNDARY •1P3E16•6 ) A1P00640
932 FORMAT (1H 13, 1H, 13, 10H SMTRY LINE •1P3E16•6 ) A1P00650
934 FORMAT ( 7H D/B=1PF14•6, 7H, T/H=1PE14•6, 7H, T/H=1PF14•6, A1P00670
1 7H, C/H=1PF14•6 ) A1P00680
936 FORMAT (30H0 BOUNDARY STRESS RESULTANTS. / 8HROW COL , A1P00690
1 6X, 4HNTAN, 9X, 5HNORM, 10X, 1HQ, 12X, 1HM ) A1P00700
937 FORMAT ( 13, 2H, 13, 2X, 1P5E12•4 ) A1P00710
938 FORMAT ( 13, 2H, 13, 66X, 1PE16•7 ) A1P00720

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939 FORMAT (59H0	CYLINDER STRAINS, CHANGES OF CURVATURE AND ROTATION	A1P00730
)		A1P00740
IN•	FORMAT (28H EDGE ROTATIONS, OMEGAX. / (1P9F14•6))	A1P00750
940 FORMAT (20H0 ISOTROPIC CYLINDER /)		A1P00760
943 FORMAT (23H0 ANISOTROPIC CYLINDER /)		A1P00770
944 FORMAT (1H0 // 8H CASE NR 12,13H COMPLETED IN F8•3, 9H MINUTES.)		A1P00780
945 FORMAT (38H0 BOUNDARY AT ZETA=CONSTANT IS HINGED)		A1P00790
946 FORMAT (37H0 BOUNDARY AT THETA=CONSTANT IS HINGED)		A1P00800
947 FORMAT (31H0 LINFAIR NORMAL PRESSURE. PO= E15•7•5H, DP= F15•7)		A1P00810
950 FORMAT (33H0 LINEAR TEMPERATURE GRADIENT. TF=F15•7,5H, TI=E15•7,		A1P00820
951 FORMAT (33H0 LINEAR TEMPERATURE GRADIENT. TF=F15•7,5H, TI=E15•7,		A1P00830
1 5H, TO=E15•7•5H, OC=F15•7)		A1P00840
953 FORMAT (19H0 DFAD LOAD. LD=F15•7)		A1P00850
962 FORMAT (9H ROW COL MX MTHFTTA	MXTHTTA QX)	A1P00860
))		A1P00870
OTHFTTA		A1P00880
966 FORMAT (6A6)		A1P00890
C SET TAPE ASSIGNMENTS.		A1P00900
KTAPE=15		A1P00910
MTAPE=7		A1P00920
B XMTP=00000002221		A1P00930
B XZTP=00000002223		A1P00940
B XNTP=00000001224		A1P00950
B NTAPE=4		A1P00960
TZTAPE=3		A1P00970
REWIND KTAPF		A1P00980
REWIND MTAPF		A1P00990
REWIND NTAPF		A1P01000
B BLNK=606060606060		A1P01010
B XRLR=606067606060		A1P01020
CALL WTAP (XZTP, TEMP, 15000, 0)		A1P01030
READ AND WRITE INPUT DATA		A1P01040
C CALL CLOCK (TIME)		A1P01050
82 READ INPUT TAPE 5, 8, RECORD		A1P01060
WRITE OUTPUT TAPE 6, 1, RECORD		A1P01070
READ INPUT TAPE 5, 7, IOPT1,IOPT2, IOPT3, IOPT4, IOPT5,IOPT6,IOPT7,A1P01080		A1P01090
1 , IOPT8		A1P01100
IOPT8 = IOPT4		A1P01110
NSTOP=IOPT6		

```

IOPT4 = 0.
IOPT5 = 0.
IOPT6 = 0.
C   IOPT1=0  ISOTROPIC CYLINDER
C   IOPT1=1  ANISOTROPIC CYLINDER
C   IOPT2=0  SYMMETRY LINE AT THETA=0
C   IOPT2=1  NO SYMMETRY LINE AT THETA=0.
C   IOPT3=0  CONSTANT MESH SPACING
C   IOPT3=1  GRADED MESH SPACING IN ZETA DIRECTION
C   IOPT 4 = 0 UNIFORM NORMAL PRESSURE
C   IOPT 4 = 1 LINEAR NORMAL PRESSURE
C   IOPT 4 = 2 LINEAR TEMPERATURE GRADIENT
C   IOPT 5 = 0 OMIT PRINT OUT OF STRAINS,CURVATURE CHANGE AND ROTATION
C   IOPT 5 = 1 PRINT STRAINS, ETC.
READ INPUT TAPE 5, 2, ROW, COL, XH, XK
READ INPUT TAPE 5, 2, ZNU, RHO, THC, RH
NRW=ROW
NCOL=COL
NC1=NCOL+1
NR1=NRW+1
NR3=3*(NRW+6)
WRITE OUTPUT TAPE 6,14, NRW, NCOL, XH, XK
WRITE OUTPUT TAPE 6, 15, ZNU, RHO, THC, RH
IF(IOPT1) 40, 40, 42
40  WRITE OUTPUT TAPE 6, 943
     DIR=0.
     DH=0.
     TH=0.
     CTH=0.
     GO TO 44
42  WRITE OUTPUT TAPE 6, 944
     READ INPUT TAPE 5, 2, DIR, DH, TH, CTH
     WRITE OUTPUT TAPE 6, 934, DIR, DH, TH, CTH
44  CONTINUE
     TF(TOPT8-1) TZO,TINI,TIN
101 RFAD INPUT TAPE 5, 2, PO, DP
     WRITE OUTPUT TAPE 6, 950, PO, DP
     GO TO 120.
105 IF((TOPT8-2) 106,106,108,

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106 READ INPUT TAPE 5, 2, TF, TI, TO, OC          A1P01510
      WRITE OUTPUT TAPE 6, 951, TE, TI, TO, OC        A1P01520
      GO TO 120
108 READ INPUT TAPE 5, 2, NDLD                  A1P01530
      WRITE OUTPUT TAPE 6, 953, NDLD
120 GDXH=XH                                     A1P01540
      IF (IOPT4) 32, 32, 31                         A1P01550
      READ INPUT TAPE 5, 2, (ROT(I), I=1, NCOL)     A1P01560
      WRITE OUTPUT TAPE 6, 940, (ROT(I), I=1, NCOL)
31 CONTINUE                                     A1P01570
      IF (IOPT5) 35, 35, 34                         A1P01580
      WRITE OUTPUT TAPE 6, 946
32 CONTINUE                                     A1P01590
      IF (IOPT6) 37, 37, 36                         A1P01600
      WRITE OUTPUT TAPE 6, 947                         A1P01610
33 CONTINUE                                     A1P01620
      IF (IOPT3) 23, 23, 22                         A1P01630
      WRITE OUTPUT TAPE 6, 920, (1, I=1, NROW)       A1P01640
      WRITE OUTPUT TAPE 6, 921, (MM(I), I=1, NROW)
34 WRITE OUTPUT TAPE 6, 946
35 IF (MM(1)=MM(NROW)) CALL GRADE (0)
36 WRITE OUTPUT TAPE 6, 947
37 CONTINUE                                     A1P01650
      IF (IOPT3) 23, 23, 22                         A1P01660
      WRITE OUTPUT TAPE 6, 920, (MM(I), I=1, NROW)
      WRITE OUTPUT TAPE 6, 921, (MM(I), I=1, NROW)
38 MM(NR1)=MM(NROW)
      CALL GRADE (0)
      GO TO 25
39 DO 24 J=1, 30
40 MM(J)=0
41 CONTINUE                                     A1P01670
      IF (IOPT2) 26, 26, 27                         A1P01680
      NDIM=3*NROW-1
42 CONTINUE                                     A1P01690
      WRITE OUTPUT TAPE 6, 922
      NCRV1=1
      GO TO 28
43 CONTINUE                                     A1P01700
      IF (IOPT2) 26, 26, 27                         A1P01710
      NDIM=3*NROW
      NCRV1=2
      NDIM=3*NROW
44 WRITE OUTPUT TAPE 6, 923                         A1P01720
      NSM=NDIM-NROW
45 NSQ=72*NDIM
      READ INPUT TAPE 5, 4, (MM(I), I=31, 35)
      IF (MM(31)) 79, 79, 51

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R      51 READ INPUT TAPF 5, 966, ((CILRL(1, J), I=1, 6), J=1, 4)          A1P01900
      XXXN=25515160K060
      NFND=0          A1P01910
      NCRV=MM(31)     A1P01920
      NPTS=NCOL/26+1  A1P01930
      NFLAG=0          A1P01940
      CALL PLTLBL (DISP, STRSS, ORDLBL, ABLBL, BOND)   A1P01950
      TH=0          A1P01960
      ABSCT(I)=TH    A1P01970
      DELH=1./COL    A1P01980
      DO 52 I=1, NC1  A1P01990
      AORD(I)=TH    A1P02000
      AORD(I)=TH    A1P02010
      TH=0          A1P02020
      DO 53 I=1, NRI  A1P02030
      AORD(I)=TH    A1P02040
      I0=NR1-I      A1P02050
      I2=XMAXOF(I0-1,1)  A1P02060
      IF (MM(I2)-MM(I0+1)) 58, 58, 56
      A1P02070
      I0=I0+1      A1P02080
      58 CONTINUE    A1P02090
      DELX=XH/(2.*MM(I0))  A1P02100
      53 TH=TH+DELX  A1P02110
      AORD(NRI+1)=TH  A1P02120
      DO 57 I=1, 6  A1P02130
      57 AXLBL(I)=RLNK  A1P02140
      AXLBL(3)=XLRL  A1P02150
      79 CONTINUE    A1P02160
      CALL COEFZ (0)  A1P02170
      MFLAG=0          A1P02180
      IF (KTIME) 81, 81, 400
      81 CONTINUE    A1P02190
      C      SPACE TAPES OVER FIRST ACCIDENT PRONE SECTION  A1P02210
      CALL WTAPE (XNTP, TEMP, 15000, 0)
      CALL WTAPE (XMTP, TFMD, 15000, 0)
      C      COMPUTE L, M, N AND Z MATRICES OF FORWARD SWEEP  A1P02220
      400 KTIME=KTIME+1  A1P02230
      N1=NSM          A1P02240
      N2=NSM          A1P02250
      N3=NSM          A1P02260
      A1P02270
      A1P02280

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I=1		A1P02290
GO TO 405		A1P02300
401 CALL WTAPE (XZTP, Z1, N1, 1)		A1P02310
CALL RTAPE (XMTP, TMP, NSQ, 0)		A1P02320
CALL MTX (YK, I, 2)		A1P02330
IF (I-2) 405, 403, 40?		A1P02340
402 CALL BACK (XNTP)		A1P02350
CALL ZFRO (YL, NSQ)		A1P02360
CALL RTAPF (XMTD, TMP, NSQ, 1)		A1P02370
CALL MTXS (TMP, YL, I, 1)		A1P02380
CALL RTAPE (XMTP, TMP, NSQ, 0)		A1P02390
CALL ADDM (YK, YL, YK, -NSQ)		A1P02400
CALL RTAPE (XNTP, YL, NSQ, 0)		A1P02410
403 NT=NDIM		A1P02420
CALL RTAPE (XMTP, TMP, NSQ, 1)		A1P02430
CALL MATM (YK, TMP, TMP, NDIM, N2, NDIM)		A1P02440
IF (I-2) 405, 405, -404		A1P02450
404 CALL RTAPE (XNTP, YL, NSQ, 1)		A1P02460
CALL MTXS (YL, TMP, I, 1)		A1P02470
405 CALL MTX (YL, I, 3)		A1P02480
IF (I-2) 408, 406, 406		A1P02490
406 CALL ADDM (YL, TMP, YL, -NSQ)		A1P02500
IF (I-NCOL) 407, 408, 408		A1P02510
407 CALL RTAPE (XNTP, TMP, NSQ, 0)		A1P02520
408 CALL INVERT (YL, N1, ISING)		A1P02530
IF (I-1) 414, 414, 410		A1P02540
410 CALL MATM (YK, Z1, T2, NDIM, N2, 1)		A1P02550
IF (I-NCOL) 411, 421, 421		A1P02560
411 CALL RTAPE (XNTP, TMP, NSQ, I)		A1P02570
CALL MATM (YK, TMP, TMP, NDIM, N2, NDIM)		A1P02580
414 CALL MTX (YK, I, 4)		A1P02590
IF (I-2) 417, 416, 416		A1P02600
416 CALL ADDM (YK, TMP, YK, -NSQ)		A1P02610
417 IF (I-NCOL+1) 418, 419, 419		A1P02620
418 CALL ZFRO (TMP, NSQ)		A1P02630
CALL MTXS (YL, TMP, I, 5)		A1P02640
CALL WTAPE (XNTP, TMP, NSQ, 0)		A1P02650
419 CALL MATM (YL, YK, YK, NI, NI, NDIM)		A1P02660
CALL WTAPE (XMTP, YK, NSQ, 0)		A1P02670

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      IF (I-NCOL+1) 420, 421, 421
      420 CALL WTAPE (XNTP, TMP, NSQ, I)
      CALL BACK (XNTP)
      421 CALL CONS (T1,I)
      IF (T-1) 426, 426, 422
      422 CALL ADDM (T1, T2, T1, -NDIM)
      IF (I-2) 426, 424, 423
      423 CALL MTXS (Z2, T2, I,-1)
      424 DO 425 J=1,NDIM
      425 Z2(J)=Z1(J)

      IF (I-NCOL) 426, 428, 428
      426 CALL WTAPE (XMTP, YK, NSQ, 1)
      CALL BACK (XMTP)
      IF (I-2) 430, 429, 428
      428 CALL ADDM (T1, T2, T1,-NDIM)
      429 CALL BACK (XMTP)
      430 CALL MATM (YL, T1, Z1, N1, N1, 1)
      CALL WTAPF (XZTP, Z1, N1, 0)
      IF (I-NCOL) 431, 450, 450
      431 IF (I-2) 434, 433, 432
      432 N3=NDIM
      433 N2=NDIM
      434 I=I+1
      GO TO 401
      C DECOMPOSITION COMPLETED . BEGIN BACKWARD SWFFP .
      450 CALL WTAPE (XZTP, Z1, N1,1)
      CALL BACK (XZTP)
      CALL BACK (XNTP)
      N4=(NCOL+3)*(NROW+6)*3
      CALL ZFRO (TMP,N4)
      451 IF (I-NCOL) 452, 454, 454
      452 DO 453 J=T,NDIM
      XX1(J)=XX(J)
      453 XX(J)=Z1(J)

      CALL RTAPE (XMTP, YK, NSQ, 1)
      CALL BACK (XMTP)
      IF (I-1) 454, 454, 4541
      4541 CALL BACK (XMTP)
      454 CALL RTAPE (XZTP, Z1, N1, 0)

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CALL RTAPE (XZTP, Z1, N1, 1)	A1P03070
IF (I-1) 4543, 4543, 4542	A1P03080
4542 CALL RACK (XZTP)	A1P03090
4543 CALL RACK (XZTP)	A1P03100
IF (I-NCOL+1) 458, 457, 462	A1P03110
456 CALL RTAPE (XNTP, YL, NSQ, 0)	A1P03120
457 CALL MATM (YK, XX, T1, N1, NDIM, 1)	A1P03130
CALL ADDM (Z1, T1, Z1, -NDIM)	A1P03140
IF (I-NCOL+1) 458, 462, 462	A1P03150
458 CALL RTAPE (XNTP, YL, NSQ, 1)	A1P03160
IF (I-1) 4591, 4591, 459	A1P03170
459 CALL RACK (XNTP)	A1P03180
4591 CALL RACK (XNTP)	A1P03190
CALL MATM (YL, XX1, T1, N1, NDIM, 1)	A1P03200
CALL ADDM (Z1, T1, Z1, -NDIM)	A1P03210
460 IF (I-2) 463, 461, 462	A1P03220
461 N1=NSM	A1P03230
462 CALL RTAPE (XNTP, YK, NSQ, 0)	A1P03240
463 CALL STORE (Z1, 1)	A1P03250
IF (I-2) 500, 464, 464	A1P03260
464 I=I-1	A1P03270
GO TO 452	A1P03280
500 MFLAG=1	A1P03290
CALL ROUND (I1)	A1P03300
IOUT=51	A1P03310
NC2=NCOL+3	A1P03320
IF (IOPT2) 501, 501, 502	A1P03330
501 NR2=NROW+3	A1P03340
GO TO 503	A1P03350
502 NR2=NROW+6	A1P03360
503 DO 520 I=1, NC2	A1P03370
I3=1*(NROW+6)*3	A1P03380
I2=I3-1	A1P03390
I1=I2-1	A1P03400
IF (IOUT+NR2-50) 505, 505, 504	A1P03410
504 WRITE OUTPUT TAPE 6, 1, RECORD	A1P03420
WRITE OUTPUT TAPE 6, 17	A1P03430
IOUT=6	A1P03440
505 DO 515 JJ=1, NR2	A1P03450

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J=NROW-.JJ+4                                A1P03460
      IF (J-NR1) 506, 511, 510                A1P03470
  506  IF (IOPT2) 507, 507, 508              A1P03480
  507  IF (J-1) 512, 512, 510                A1P03490
  508  IF (JJ) 510, 511, 510                A1P03500
  510  WRITE OUTPUT TAPE 6, 930, I, J, TMP(I1), TMP(I2), TMP(I3)
  GO TO 514                                  A1P03510
  511  WRITE OUTPUT TAPE 6, 931, I, J, TMP(I1), TMP(I2), TMP(I3)
  GO TO 514                                  A1P03520
  512  WRITE OUTPUT TAPE 6, 932, I, J, TMP(I1), TMP(I2), TMP(I3)
  GO TO 514                                  A1P03530
  513  WRITE OUTPUT TAPE 6, 933, I, J, TMP(I1), TMP(I2), TMP(I3)
  GO TO 514                                  A1P03540
  514  I1=I1-3                               A1P03550
  I2=I2-3                                   A1P03560
  I3=I3-3                                   A1P03570
  515  IOUT=IOUT+1                           A1P03580
  520  CONTINUE                               A1P03590
  521  IOUT=50                               A1P03600
  L=32                                         A1P03610
  K=0                                         A1P03620
  NNR=NR1+1                                 A1P03630
  DO 5401  JJ=1, NNR                         A1P03640
  J=JJ-1                                     A1P03650
  IF (J) 526, 526, 529                         A1P03660
  526  IF (IOPT2) 5401, 5401, 529             A1P03670
  529  CONTINUEIF
  IF (IOPT3) 537, 537, 536
  536  CALL GRADE (J)
  537  IF (MM(31)) 5378, 5378, 5370
  5370 IF (MM(L)-J) 5378, 5372, 5378
  5372 IF (L-35) 5374, 5374, 5378
  5374 L=L+1
  K=K+1
  K1=K
  GO TO 5380
  5378 K1=0
  5380 DO 540  I=1, NC1
  IF (IOUT-41) 539, 538, 538
  538  WRITE OUTPUT TAPE 6, 1, RECORD
  TOUT=0
  IF (MFLAG) 532, 530, 531

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531	WRITE OUTPUT TAPE 6, 20	A1P03850
	WRITE OUTPUT TAPE 6, 18	A1P03860
	GO TO 539	A1P03870
530	WRITE OUTPUT TAPE 6, 20	A1P03880
	WRITE OUTPUT TAPE 6, 962	A1P03890
	GO TO 539	A1P03900
532	WRITE OUTPUT TAPE 6, 939	A1P03910
	WRITE OUTPUT TAPE 6, 19	A1P03920
539	IOUT=IOUT+1	A1P03930
	CALL STRESS (I,J,K1)	A1P03940
540	CONTINUE	A1P03950
5401	CONTINUE	A1P03960
	IF (MFLAG) 999, 544, 541	A1P03970
541	MFLAG=0	A1P03980
	GO TO 521	A1P03990
544	WRITE OUTPUT TAPE 6, 1, RECORD	A1P04000
	TO=0	A1P04010
	I1=1	A1P04020
	I2=4	A1P04030
	WRITE OUTPUT TAPE 6, 936	A1P04040
I=1		A1P04050
	DO 560 JJ=1, NR1	A1P04060
	J=NRI+I-JJ	A1P04070
550	WRITE OUTPUT TAPE 6, 937, J, I, (YK(I3), I3=I1, I2)	A1P04080
	I1=I1+4	A1P04090
	I2=I1+3	A1P04100
	IF (I-NCOL) 551, 552, 560	A1P04110
551	I=I+1	A1P04120
	GO TO 550	A1P04130
552	I=NC1	A1P04140
	WRITE OUTPUT TAPE 6, 937, J, NC1, (YK(I3), I3=I1, I2)	A1P04150
	KI=8*(NC1+NR1)-3	A1P04160
	WRITE OUTPUT TAPE 6, 938, J, NC1, YK(K1)	A1P04170
	I1=I1+4	A1P04180
	I2=I1+3	A1P04190
	GO TO 550	A1P04200
560	CONTINUE	A1P04210
	IF (TOPT2) 566, 566, 562	A1P04220
562	WRITE OUTPUT TAPE 6, 937, IO, NC1, (YK(I3), I3=I1, I2)	A1P04230

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K1=K1+4          A1P04240
      WRITE OUTPUT TAPE 6, 938, IN, NCI, YK(K1)    A1P04250
      I1=I1+4          A1P04260
      I2=I1+3          A1P04270
      DO 565 I1=1, NCI          A1P04280
      I=NC1+1-II        A1P04290
      WRITE OUTPUT TAPE 6, 937, IO, I, (YK(I3), I3=11, 12)  A1P04300
      I1=I1+4          A1P04310
      I2=I1+3          A1P04320
      565 CONTINUE      A1P04330
      566 IF (TOP7) 990,999, 545          A1P04340
      545 MFLAG=-1        A1P04350
      GO TO 521        A1P04360
      999 CONTINUE      A1P04370
      80 CALL CLOCK (TIM2)          A1P04380
      TIME=TIM2-TIM1          A1P04390
      TIME=TIM2
      PRINT 945, KTIME, TIME          A1P04400
      WRITE OUTPUT TAPE 6, 945, KTIME, TIME          A1P04410
      IF (MM(31)) 1000, 1000, 601          A1P04420
      601 DO 650 K=1,3          A1P04430
      WRITE TAPF K TAPF, RFCORD          A1P04440
      WRITE TAPE KTAPE, NCI, NCRV, NPTS, NEND, NFLAG          A1P04450
      WRITE TAPE KTAPE, ARLRL, (ORDLRL(M,K),M=1,6), DISP          A1P04460
      WRITE TAPF K TAPF, CILRL          A1P04470
      DO 610 I=1, NCI          A1P04480
      WRITE TAPE KTAPE, ARSC(I)          A1P04490
      610 CONTINUE          A1P04500
      L=32
      L3=31+MM(31)
      DO 620 J=0, NR1          A1P04520
      IF (MM(L)-J) 620, 611, 620          A1P04530
      611 I1=3*(J+2)+K          A1P04540
      DO 615 I=1, NC1          A1P04550
      615 I1=I1+NR3          A1P04560
      L=L+1
      IF (L-L3) 620, 620, 650          A1P04570
      620 CONTINUE          A1P04580

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650 CONTINUE          A1P04630
DO 720 K=1, 4          A1P04640
K1=K+3                A1P04650
WRITE TAPE KTape, RECORD          A1P04660
----- WRITE TAPE KTape, NCI, NCRV, NPTS, NEND, NFLAG          A1P04670
----- WRITE TAPE KTape, ABLBL, (ORDLBL(M, K1), M=1, 6), STRSS          A1P04680
----- WRITE TAPE KTape, CRLRL          A1P04690
DO 667 I=1, NCI          A1P04700
WRITE TAPE KTape, ARSC(I);          A1P04710
662 CONTINUF          A1P04720
DO 660 M=1, NCRV          A1P04730
DO 660 I=1, NCI          A1P04740
660 WRITE TAPE KTape, YL(K,M,I)          A1P04750
720 CONTINUF          A1P04760
DO 740 K1=1, 4          A1P04770
K=K1+7                A1P04780
NCRV=NCRV1           A1P04790
NCR=0                 A1P04800
NCX=NC1               A1P04810
725 WRITE TAPE KTape, RECORD          A1P04820
WRITE TAPE KTape, NCX, NCRV, NPTS, NEND, NFLAG          A1P04830
IF (NCR) 723, 723, 724          A1P04840
723 CONTINUE          A1P04850
WRITE TAPE KTape, ABLBL, (ORDLBL(M,K), M=1, 6), BOND          A1P04860
GO TO 726           A1P04870
724 CONTINUE          A1P04880
WRITE TAPE KTape, AXLBL, (ORDLBL(M, K), M=1, 6), BOND          A1P04890
726 CONTINUE          A1P04900
CALL PLBL (CBLBL, NCR)          A1P04910
WRITE TAPE KTape, CRLRL          A1P04920
IF (NCR) 727, 727, 729          A1P04930
727 DO 728 I=1, NCI          A1P04940
728 WRITE TAPE KTape, ARSC(I)          A1P04950
I1=K1                 A1P04960
GO TO 731           A1P04970
729 DO 730 I=1, NR1           A1P04980
730 WRITE TAPE KTape, AORD(I)           A1P04990
731 CONTINUF          A1P05000
KAD=4                 A1P05010

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DO 735 M=1, NCRV A1P05020
DO 732 L=1, NCX A1P05030
    WRITE TAPE KTAPE, YK(I1)
732 I1=I1+KAD A1P05040
    T1=4*(NC1+NC1+NRT1)+K1 A1P05050
735 KAD=-4 A1P05060
    IF (NCR) 736, 736, 740 A1P05070
736 NCR=1 A1P05080
    I1=4*NC1+K1 A1P05090
    KAD=4 A1P05100
    NCX=NRT1 A1P05110
    NCRV=1 A1P05120
    IF (K1=4) 715, 714, 714 A1P05130
714 NFEND=NSTOP A1P05140
715 CONTINUE A1P05150
    GO TO 725 A1P05160
740 CONTINUE A1P05170
    WRITE TAPE KTAPE, XXN A1P05180
    BACKSPACE KTAPE A1P05190
    CALL RFSFT A1P05200
    IF (NSTOP) 82, 82, 1001 A1P05210
1001 CONTINUE A1P05220
    CALL CHAIN (2,2) A1P05230
1000 CALL RESET A1P05240
    GO TO 82 A1P05250
    END A1P05260
    * FORTRAN A1P05270
    * SURROUNINF ADDM (X, Y, Z, M) A1P05280
    DIMENSION X(3600), Y(3600), Z(3600) A1P05290
    IF (M) 20, 1, 1 A1P05300
    1 DO 10 I=1,M A1P05310
    10 Z(I)=X(I)+Y(I) A1P05320
    11 GO TO 50 A1P05330
    20 M=-M A1P05340
    DO 25 I=1,M A1P05350
    25 Z(I)=X(I)-Y(I) A1P05360
    50 RETURN A1P05370
    END A1P05380
    * FORTRAN A1P05390
    A1P05400

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SUBROUTINE ROUND ( IJ )
COMMON NDIM, NROW, NCOL, MM, I1, KP1, KP2, KM1, KM2, MFLAG, B, C A1P05410
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8 A1P05420
COMMON A1, A2, A3, A4, A5, A6 A1P05430
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P05440
   1 C9, C10, C11, C12 A1P05450
COMMON TMP, YK, YL A1P05460
DIMENSION TMP(5184), YK(5184), YL(5184) A1P05470
DIMENSION MM(35) A1P05480
   1 I1=1 A1P05490
NR1=NROW+1 A1P05500
   1 I1=0 A1P05510
NC1=NCOL+1 A1P05520
   1 NC1=NCOL+1 A1P05530
   1 N5=I2*I2 A1P05540
CALL ZERO (YK, N5)
IF (IOPT3) 401, 401, 400 A1P05550
400 CALL GRADE(I1) A1P05560
401 CONTINUE A1P05570
IF (IOPT2) 499, 499, 450 A1P05580
   1 I1=I1+1 A1P05590
450 J=0 A1P05600
DO 4001 K=37, 40 A1P05610
4001 MM(K)=0 A1P05620
   1 DO 460 I2=1, NCOL A1P05630
   1 I1=I1+1
CALL EQ1 (YK, I1,I2, J, 1) A1P05640
   1 IF (I2-1) 453, 453, 451 A1P05650
451 I1=I1+1
CALL EQ2 (YK, I1,I2, J, 1) A1P05660
   1 I1=I1+1 A1P05670
YK(I1+2970)=C A1P05680
   1 CALL EQ3 (YK, I1,I2, J, 1) A1P05690
460 CONTINUE A1P05700
456 I1=I1+1 A1P05710
   1 CALL EQ2 (YK, I1,I2, J, 1) A1P05720
   1 I1=I1+1 A1P05730
   1 CALL EQ1 (YK, I1,I2, J, 1) A1P05740
   1 I1=I1+1 A1P05750
   1 CALL EQ3 (YK, I1,I2, J, 1) A1P05760
YK(I1+2970)=C A1P05770
   1 CALL EQ3 (YK, I1,I2, J, 1) A1P05780
   1 I1=I1+1 A1P05790

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IF (IOPT3) 499, 499, 498
498 CALL GRADE (1)
499 J=1
DO 515 I2=1, NC1
  I=NC1+I-I2
    501 IF (IOPT2) 502, 502, 503
    502 IF (J-1) 504, 504, 503
  503 I1=I1+1
    CALL FQ1 (YK, I1, I, J, 1)
  504 IF (I-1) 506, 506, 505
  505 I1=I1+1
    CALL FQ2 (YK, I1, I, J, 1)
  506 IF (J-NR1) 508, 507, 507
  507 IF (I-NC1) 508, 509, 509
  508 I1=I1+1
    YK(I1+2970)=C
    CALL EQ3 (YK, I1, I, J, 1)
  509 IF (J-NR1) 510, 515, 515
  510 J=J+1
    IF (IOPT3) 512, 512, 514
  514 CALL GRADF(J)
  512 IF (J-NR1) 501, 511, 511
  511 I1=I1+1
    CALL FQ2 (YK, I1, I, J, 1)
  I1=I1+1
    YK(I1+2970)=C
    CALL EQ3 (YK, I1, I, J, 1)
  515 CONTINUE
C   BOUNDARY EQUATIONS FORMFD IN YK
    CALL QROSTM (YK, T1, 5)
    CALL STORE (YL, n)
    RETURN
  516 CONTINUE
END
*      FORTRAN
*      LIST
      SUBROUTINE CF(I, J, MF, K, AA, X)
DIMENSION X(72, 72)
      A1P05800
      A1P05810
      A1P05820
      A1P05830
      A1P05840
      A1P05850
      A1P05860
      A1P05870
      A1P05880
      A1P05890
      A1P05900
      A1P05910
      A1P05920
      A1P05930
      A1P05940
      A1P05950
      A1P05960
      A1P05970
      A1P05980
      A1P05990
      A1P06000
      A1P06010
      A1P06020
      A1P06030
      A1P06040
      A1P06050
      A1P06060
      A1P06070
      A1P06080
      A1P06090
      A1P06100
      A1P06120
      A1P06130
      A1P06140
      A1P06150
      A1P06160
      A1P06170
      A1P06180

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COMMON NDIM, NRROW, NCOL, MM, I1, KP1, KP2, KM1, KM2, MFLAG, B,C A1P06190
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8 A1P06200
COMMON A1, A2, A3, A4, A5, A6 A1P06210
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P06220
C C9, C10, C11, C12 A1P06230
COMMON TMP, YK, YL A1P06240
COMMON ZNU, RHO, THC, R1H, DIR, D1H, T1H, C1H A1P06250
COMMON TE, TI, TO, OC, TC, TD, DDLD A1P06260
COMMON ROT A1P06270
DIMENSION TMP(5184), YK(5184), YL(5184) A1P06290
DIMENSION MM(35) A1P06300
DIMENSION ROT(80) A1P06310
C COMPUTE COEFFICIENT OF KTH UNKNOWN FOR MESH PT T,J. A1P06320
I2=1 A1P06330
J2=J A1P06340
M=MF A1P06350
NC1=NCOL+1 A1P06360
NR1=NRROW+1 A1P06370
TF(M) 110, 100, 120 A1P06380
110 IF (M+1) 111, 112, 113 A1P06390
111 J2=J2-1 A1P06400
M=0 A1P06410
GO TO 100 A1P06420
112 FCTR=.5 A1P06430
J4=J2-1 A1P06440
GO TO 101 A1P06450
120 IF (M-1) 122, 122, 121 A1P06460
121 J2=J2+1 A1P06470
M=0 A1P06480
GO TO 100 A1P06490
122 FCTR=.5 A1P06500
J4=J2+1 A1P06510
GO TO 101 A1P06520
100 FCTR=1. A1P06530
101 IF (I2-1) 2, 4, 6 A1P06540
C COL I2 IS LEFT OF X-AXIS. REFLECT OVER SYMMETRY LINE. CHANGE SIGN A1P06550
C TF K=? A1P06560
C 2 I2=-I2+2 A1P06570

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      IF (K=2) 5, 3, 6          A1P06580
      GO TO 6          A1P06580
C     COL 12 IS X-AXIS. OMIT COEFFICIENT 1E K=2          A1P06600
      4 IF (K=2) 6, 99, 6          A1P06610
      6 IF (J2-NR1) 14, 99, 7          A1P06620
      7 IF (J2-NR1-1) 8, 8, 11          A1P06630
      8 IF (K=2) 11, 11, 9          A1P06640
      9 IF (I2-NC1) 10, 99, 11          A1P06650
C     ROW J2 IS ONE BEYOND BOUNDARY. REFLECT W.          A1P06660
      10 J2=J2-2          A1P06670
      201 IF (IOPT5) 210, 210, 205          A1P06680
      210 IF (MFLAG) 50, 50, 212          A1P06690
      212 IF (IOPT4) 50, 50, 214          A1P06700
      214 CONTINUEF          A1P06710
      YK(I1+2970)=YK(I1+2070)+2.*AA*FCCTR*RNT(I2)*XH/2.*MM(NROW)          A1P06720
      GO TO 50          A1P06730
      11 J2=NR1          A1P06740
      12 IF (I2-NC1) 80, 99, 13          A1P06750
      13 I2=NC1          A1P06760
      GO TO 80          A1P06770
      14 IF (IOPT2) 15, 15, 19          A1P06780
      15 IF (J2-1) 16, 18, 25          A1P06790
      16 J2=-J2+2          A1P06800
      17 IF (K-1) 17, 17, 25          A1P06810
      18 IF (K=1) 99, 99, 25          A1P06820
      19 IF (J2) 20, 99, 25          A1P06830
      20 IF (J2-1) 24, 21, 21          A1P06840
      21 IF (K-2) 24, 24, 22          A1P06850
      22 IF (I2-NC1) 23, 99, 24          A1P06860
      23 J2=1          A1P06870
      GO TO 201          A1P06880
      24 J2=0          A1P06890
      GO TO 12          A1P06900
      25 IF (I2-NC1) 50, 99, 26          A1P06910
      26 IF (I2-NC1-1) 27, 27, 13          A1P06920
      27 IF (K-2) 13, 13, 28          A1P06930

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C   COL I2 IS ONE BEYOND BOUNDARY. REFLECT W. CHANGE SIGN IF HINGED    A1P06970
28  I2=I2-2
      IF (IOPT6) 50, 50, 205
205  FCTR=-FCTR
      50  JT=INDX(I2, J2, K)
          IF (MFLAG) 51, 51, 52
      51  X(I1, J1)=X(I1, J1)+AA*FCTR
      52  YK(I1+2970)=YK(I1+2970)-AA*FCTR*TMP(J1)
          GO TO 125
125
      80  JT=INDX(I2, J2, K)
          J3=(J1+5-I1)*270+11
          YK(J3)=YK(J3)+AA*FCTR
      130
      99  RETURN
130
      135  J2=J4
      M=0
      GO TO 50
135
      END
*   FORTRAN
      SURROUTINE COFFZ (M)
      COMMON NDIM, NROW, NCOL, MM, MFLAG, R, C
      COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
      COMMON A1, A2, A3, A4, A5, A6
      COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P07240
      1 C9, C10, C11, C12
      COMMON TMP, YK, YL
      COMMON ZNU, RH0, TH0, R1H, D1R, D1H, TH1, C1H
      COMMON XXH, XXK, GDXH, PR, DB
      COMMON TE, TI, TN, NC, TC, TD, NDLD
      COMMON PCT
      COMMON FX, FNU, FTX, SC, DX, DTX, DNII
      DIMENSION R(80)
      DIMENSION MM(40)
      DIMENSION TMP(5184), YK(5184), YT(5184)
      IF (M) 50, 50, 10

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10 XH=XXH/(2.***M) A1P07360
   GO TO 51 A1P07370
50 XH=XXH A1P07380
51 CONTINUE A1P07390
TF(TOPT4) 6, 6, 5 A1P07400
5 C=0 A1P07410
GO TO 7 A1P07420
6 C=1 A1P07430
7 CONTINUE A1P07440
TC=0 A1P07450
TD=0 A1P07460
ZK=1./((12.*R1H**2) A1P07470
IF (LOPT8-2) 20, 12, 15 A1P07480
.2 ZT1=.5*(TE+TI) -TO A1P07490
ZT2=R1H*(TE-TI) A1P07500
ZT3= TI-TO A1P07510
TC=QC*(ZT1/((1.-ZNU)+T1H*D1B*ZT3) A1P07520
TD=QC*(ZT2/((12.*R1H**2*((1.-ZNU))-C1H*T1H*D1B*D1B/R1H))
C= QC*((ZT1/((1.-ZNU)+T1H*D1B*ZT3)) A1P07530
   GO TO 20 A1P07540
15 R=1. A1P07550
   C=1. A1P07560
20 CONTINUE A1P07570
C COMPUTE CONSTANTS A1P07580
TP1= (1.-ZNU*ZNU) A1P07590
TP2= RHO/THC A1P07600
TP3= TP2**2 A1P07610
EX= (1.+T1H*D1R*TP1)/TP1 A1P07620
ENU= ZNU/TP1 A1P07630
ETX= 1./(2.*(1.+ZNU)) A1P07640
C1R=C1H/R1H A1P07650
TIR=T1H/PTH A1P07660
SC=C1R*T1H*D1R A1P07670
DX = ZK/TP1*(1.+T1H*D1R*(D1H**2+12.*C1H**2)*TP1) A1P07680
DTX= ZK*((1.+ZNU)*(T1H*D1H*D1B)) A1P07700
DNU= ZNU*ZK/TP1 A1P07710
YA1= EX*TP3 A1P07720
YA2= ETX/THC**2 A1P07730
YA3= TP2/THC*(FNU+FTX) A1P07740

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YA4 = ENU*TP2		A1P07750
YA5 = SC*TP2**3		A1P07760
YB2 = TP3*(ETX+2.*DTX)		A1P07770
YR3 = (FX+DX-2.*SC)/THC**2		A1P07780
YR4 = -TP3/THC*(DNU+2.*DTX)		A1P07790
YB5 = -(DX-SC)/THC**3		A1P07800
YB6 = (FX-SC)/THC		A1P07810
YC1 = +ENU*TP2		A1P07820
YC2 = +YB4		A1P07830
YC3 = +YB5		A1P07840
YC4 = +YB6		A1P07850
YC5 = +DX*TP3**2		A1P07860
YC6 = +2.*(DNU+DTX)*TP3/THC**2		A1P07870
YC7 = +DX/THC**4		A1P07880
YC8 = +FX		A1P07890
YC9 = +SC*TP2**3		A1P07900
YC10 = +2.*SC/THC**2		A1P07910
A1 = YA1/XH**2		A1P07920
A3 = YA2/XK**2		A1P07930
A2 = -2.*(A1+A3)		A1P07940
A4 = YA3/(4.*XH*XK)		A1P07950
A5 = YA4/(2.*XH)-YA5/XH***3		A1P07960
A6 = YA5/(2.*XH***3)		A1P07970
B1 = A4		A1P07980
B2 = YR2/XH**2		A1P07990
B4 = YR2/XK**2		A1P08000
B3 = -2.*(R2+R4)		A1P08010
B6 = YR4/(2.*XH**2*XK)		A1P08020
B7 = YB5/(2.*XK**3)		A1P08030
B5 = -2.*(B6+B7)+YR6/(2.*XK)		A1P08040
C12 = YC9/(2.*XH**3)		A1P08050
C1 = YC17/(2.*XH)-Z.*CT2		A1P08060
C2 = -YC3/XK**2		A1P08070
C5 = YC7/(2.*XH**2*XK)		A1P08080
C3 = -2.*C5-C2+YC7/(2.*XK)		A1P08090
C4 = -C3-4.*C2		A1P08100
C6 = YC5/XH**4		A1P08110
C9 = YC5/(XH*XK)**2		A1P08120
C10 = YC7/(XK**4)		A1P08130

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C7= -(4.*C6+2.*C9) A1P08140
C8= -2.*((C9+2.*C10)+YC10/XK**2) A1P08150
C11= 6.*((C6+C10)+4.*C9-2.*YC10/XK**2+YC8 A1P08160
CALL NORM (A1, A, 6) A1P08170
CALL NORM (B1, B, 7) A1P08180
CALL NORM (C1, C, 12) A1P08190
RETURN A1P08200
END A1P08210
* FORTRAN A1P08220
SUBROUTINE CONS (X, I) A1P08230
COMMON NDIM, NROW, NCOL, MM, MFLAG, R, C A1P08240
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8 A1P08250
COMMON A1, A2, A3, A4, A5, A6 A1P08260
COMMON B1, B2, B3, B4, B5, B6, C1, C2, C3, C4, C5, C6, C7, C8, A1P08270
1 C9, C10, C11, C12 A1P08280
COMMON TMP, YK, YL A1P08290
COMMON ZNU, RH0, THC, RIH, DTB, DTH, TIH, CIH A1P08300
COMMON XH, XK, GDXH, PO, DP A1P08310
COMMON TE, TI, TO, OC, TC, TD, DDLD A1P08320
COMMON ROT A1P08330
DIMENSION TMP (5184), YK (5184), YL (5184) A1P08340
DIMENSION ROT (80) A1P08350
DIMENSION XT777 A1P08360
DIMENSION MM(40) A1P08370
3 DO 5 J=1, NDIM A1P08380
5 X(J)=0. A1P08390
TH= THC/FLOATF(NCOL)*FLOATF(I-1)
Z=1.
DO 40 J=1, NROW A1P08400
IF (IOPT3) 7, 7, 6 A1P08420
6 CALL GRADE (J) A1P08440
7 IT=INDEX(I, J, 3) A1P08450
IF (IOPT8-1) 10, 15, 30 A1P08460
10 X(I1)=C A1P08470
15 IF (MM(30)) 25, 25, 20 A1P08480
20 Z=Z-GDXH A1P08490
25 Z=Z-GDXH A1P08500
X(I1)=C*(PO+DP*Z) A1P08510
A1P08520

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A1P08530
A1P08540
A1P08550
A1P08560
A1P08570
A1P08580
A1P08590
A1P08600
A1P08610
A1P08620
A1P08630
A1P08640
A1P08650
A1P08660
A1P08670
A1P08680
A1P08690
A1P08700
A1P08710
A1P08720
A1P08730
A1P08740
A1P08750
A1P08760
A1P08770
A1P08780
A1P08790
A1P08800
A1P08810
A1P08820
A1P08830
A1P08840
A1P08850
A1P08860
A1P08870
A1P08880
A1P08890
A1P08900
A1P08910

GO TO 40
30 IF( (INOPT8-2) .LT. 10 ) 10, 10, 35
35 IF( (I-1) .LT. 39 ) 39, 39, 36
35 X( I1-1 ) = -R*DLD*SINF( TH )
39 X( I1 ) = -C*DLD*COSF( TH )
40 CONTINUE
IF( INOPT4 ) 80, 80, 12
12 X( I1 ) = X( I1 ) + 2.0*( C6*XH*RDT( I ) / 2.0*MM( NROW ))
99 RETURN
END
* FORTRAN
* LIST
FUNCTION DFF( II, JJ, KK, MF )
COMMON NDIW, NRW, NCOL, MM, MFLAG, R, C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P08690
1 C9, C10, C11, C12
COMMON TMP, YK, YL
DIMENSION TMP( 5184 ), YK( 5184 ), YL( 5184 )
DIMENSION MM( 40 )
N2=3*( NRW+6 )
I=II
J=JJ
K=KK
M=MF
IF( (M) .LE. 50, 50, 70
60 IF( (M+1) .LE. 61, 62, 62
61 J=J-1
M=0
GO TO 50
62 J2=J-1
GO TO 50
70 IF( (M-1) .LT. 72, 72, 71
71 J=J+1
M=0
GO TO 50
72 J2=J+1
50 SIGN=1.

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1 IF (I-1) 1, 5, 5
1 IF (I-2) 1, 5, 5
2 SIGN=-1.
5 IF (IOPT2) 6, 6, 12
6 IF (J-1) 7, 12, 12
7 J=2-J
12 IF (K-1) 8, 8, 12
8 SIGN=-1.
12 I2= (I-1)*N2+3*J+K+6
12 IF (M) 16, 13, 16
13 IF (SIGN) 14, 15, 15
14 DFF=-TMP(I2)
GO TO 90
15 DFF=TMP(I2)
GO TO 90
16 I3=(I-1)*N2+3*J2+K+6
DEF=SIGN*(TMP(I2)+TMP(I3))/2.
90 RETURN
END
*      FORTRAN
      SUBROUTINE EQ1 (X, III, II, JJ, MX)
COMMON NDTM, NROW, NCOL, MM, K0, KP1, KP2, KM1, KM2, MFLAG, B,C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
C      COMPUTE COEFFICIENTS FOR EQUATION ONE FOR MESH PT I,J
DIMENSION MM(25)
I1=II
I=II
J=JJ
M=MX
GO TO 130, 20, 30, 40, 50), M
20 CONTINUE
I1=0
CALL CF (I-1,J,I1,A3,X)
I1=KM1
CALL CF (I-1,J-1,I1,2,-A4,X)
I1=KDT
CALL CF (I-1,J+1,I1,2, A4,X)

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GO TO 50
I1=0
CALL CF ( I, J, II, 1, A2, X)
II=KM1
CALL CF ( I,J-1,II,1,A1, X)
CALL CF ( I, J-1, II,2, A5, X)
II=RPT
CALL CF ( I,J+1,II,1,A1, X)
CALL CF ( I, J+1, II,3,-A5, X)
II=KM2
CALL CF ( I, J-2, II, 2, A5, X)
II=KP2
CALL CF ( I, J+2, II, 3,-A6, X)
IF (MFLAG) 50, 50, 40
40 CONTINUE
I1=0
CALL CF ( I+1,J,II,1,A3, X)
II=KM1
CALL CF ( I+1,J-1,II,2, A4, X)
I1=KP1
CALL CF ( I+1,J+1,II,2,-A4, X)
45 IF (I-1) 20, 20, 46
46 IF (MFLAG) 50, 50, 20
50 RETURN
END
*          FORTRAN
C          SUBROUTINE EQ2 (X, III, II, JJ, MX)
C          COMPUTE EQUATION TWO FOR MESH PT I, J
COMMON NDIM, NROW, NCOL, MM, K0, KP1, KP2, KM1, KM2, MFLAG, B,C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
COMMON B1, R2, R3, R4, B5, R6, R7, C1, C2, C3, C4, C5, C6, C7,C8,A1Pn9630
1 CO, C10, C11, C12
1 DIMENSION MM(35)
I1=III
I=II
J=JJ
M=MX

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GO TO (30, 20, 30, 40, 50), M          A1P09700
20 CONTINUE                                A1P09710
    I1=0                                     A1P09720
    CALL CF(I-1,J, I1,2, B4, X)             A1P09730
    CALL CF(I-1,J, I1,3, -B5, X)            A1P09740
    I1=K1                                    A1P09750
    CALL CF(I-1,J-1,I1,3,-B6, X)           A1P09760
    CALL CF(I-1,J-1,I1,1,-B1, X)           A1P09770
    I1=KP1                                   A1P09780
    CALL CF(I-1,J+1,I1,1, B1, X)           A1P09790
    CALL CF(I-1,J+1,I1,3,-B6, X)           A1P09800
    GO TO 50                                 A1P09810
30 CONTINUE                                A1P09820
    I1=K1                                    A1P09830
    CALL CF(I, J-1, I1,2, B2, X)           A1P09840
    I1=KP1                                   A1P09850
    CALL CF(I, J+1, I1,2, B2, X)           A1P09860
    I1=0                                     A1P09870
    CALL CF(I, J, I1, 2, R3, X)             A1P09880
    33 IF (MFLAG) 34, 34, 35               A1P09890
    34 IF (I-2) 50, 35, 36                  A1P09900
    35 CONTINUE                               A1P09910
    CALL CF(I-2,J, I1, 3,-B7, X)           A1P09920
    IF (MFLAG) 50, 50, 37                  A1P09930
    36 IF (I-NCOL) 50, 37, 37              A1P09940
    37 CONTINUE                               A1P09950
    CALL CF(I+2,J, I1, 3, R7, X)           A1P09960
    IF (MFLAG) 50, 50, 40                  A1P09970
40 CONTINUE                                A1P09980
    I1=0                                     A1P09990
    CALL CF(I+1,J, I1,2, B4, X)           A1P10000
    CALL CF(I+1,J, I1,3, -R5, X)           A1P10010
    I1=K1                                    A1P10020
    CALL CF(I+1,J-1,I1,1, R1, X)           A1P10030
    CALL CF(I+1,J-1,I1,3, R6, X)           A1P10040
    I1=KP1                                   A1P10050
    CALL CF(I+1,J+1,I1,1,-R1, X)           A1P10060
    CALL CF(I+1,J+1,I1,3, R6, X)           A1P10070
    45 IF (I-1) 20, 20, 46                  A1P10080

```

46 IF (MFLAG) 50, 50, 20
50 RETURN

END

```
*   FORTRAN
      SUBROUTINE EQ3 (X, I11, I1, JJ, MX)
      COMPUTE EQUATION THRF FOR MESH PT I,J.
      COMMON NDTM, NROW, NCOL, MM, K0, KP1, KP2, KM1, KM2, MFLAG, B,C
      COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
      COMMON A1, A2, A3, A4, A5, A6
      COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7,C8,
      1 C9, C10, C11, C12
      DIMENSION MM(35)
      I=I1
      J=JJ
      M=MX
      GO TO (30, 20, 30, 40, 50), M
      20 CONTINUE
      I1=0
      CALL CF (I-1,J,I1,2, C4, X)
      CALL CF (I-1,J, I1, 3, C8, X)
      I1=KM1
      CALL CF (I-1, J-1, I1, 3, C9, X)
      CALL CF (I-1, J-1, I1, 2,-C5, X)
      I1=KB1
      CALL CF (I-1, J+1, I1, 2,-C5, X)
      CALL CF (I-1, J+1, I1, 3, C9, X)
      GO TO 50
      30 CONTINUE
      I1=KM1
      CALL CF (I, J-1,I1, 3, C7, X)
      CALL CF (I, J-1,I1,1, C1, X)
      I1=KP1
      CALL CF (I, J+1,I1,1,-C1, X)
      CALL CF (I, J+1,I1, 3, C7, X)
      I1=KM2
      CALL CF (I, J-2, I1, 3, C6, X)
      I1=KD2
```

A1P10090
A1P10100
A1P10110
A1P10120
A1P10130
A1P10140
A1P10150
A1P10160
A1P10170
A1P10180
A1P10190
A1P10200
A1P10210
A1P10220
A1P10230
A1P10240
A1P10250
A1P10260
A1P10270
A1P10280
A1P10290
A1P10300
A1P10310
A1P10320
A1P10330
A1P10340
A1P10350
A1P10360
A1P10370
A1P10380
A1P10390
A1P10400
A1P10410
A1P10420
A1P10430
A1P10440
A1P10450
A1P10460
A1P10470

```

CALL CF (I, J+2, II, 3, C6, X)          A1P10480
TF (J-NROW) 33, 37, 32                 A1P10490
32 CONTINUE
CALL CF (I, J-2, KM2, 1, 2.*C12, X)     A1P10500
CALL CF (I, J-I, KM1, I,-4.*C12, X)     A1P10510
CALL CF (I, J, 0, 1, 6.*C12, X)         A1P10520
CALL CF (I, J+1, 0, 1, -4.*C12, X)      A1P10530
GO TO 380                                A1P10540
33 CONTINUE
IF (LOPT2) 37, 37, 35                  A1P10550
35 IF (J-1) 36, 36, 37                 A1P10560
36 CONTINUE
CALL CF (I, J-1, 0, 1, 4.*C12, X)       A1P10570
CALL CF (I, J, 0, 1,-6.*C12, X)         A1P10580
CALL CF (I, J+1, KP1, 1, 4.*C12, X)     A1P10590
CALL CF (I, J+2, KP2, 1,-2.*C12, X)     A1P10600
GO TO 380                                A1P10610
37 CONTINUE
CALL CF (I, J+2, II, 1,-C12, X)         A1P10620
38 CONTINUE
IF (I=KM2)                                A1P10630
CALL CF (I, J-2, II, 1, C12, X)          A1P10640
380 CONTINUE
380 CONTINUE
I=0
CALL CF (I, J, II, 3, C11, X)           A1P10650
CALL CF (I,J, II, 2, 3.*C2, X)         A1P10660
TF (MFLAG) 381, 381, 38                A1P10670
381 IF (I-2) 50, 38, 39                 A1P10680
38 CONTINUE                                A1P10690
380 CONTINUE
CALL CF (I-2,J, II, 2, C2, X)          A1P10700
CALL CF (I-2,J, II, 3, C10, X)         A1P10710
IF (MFLAG) 50, 50, 390                 A1P10720
TF (MFLAG) 381, 381, 38                A1P10730
39 IF (I-NCOL) 50, 390, 390            A1P10740
390 CONTINUE
CALL CF (I+2,J, II, 3, C10, X)         A1P10750
390 CONTINUE
IF (MFLAG) 50, 50, 40                 A1P10760
40 CONTINUE
I=0
CALL CF (I+1,J, II, 3, C3, X)          A1P10770
CALL CF (I+1,J, II, 3, C8, X)          A1P10780
TF (MFLAG) 50, 50, 40                 A1P10790
40 CONTINUE
A1P10800
A1P10810
A1P10820
A1P10830
A1P10840
A1P10850
A1P10860

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```

I1=KM1                               ALP10870
CALL CF(I+1,J-1,I1,3,C9,X)          ALP10880
CALL CF(I+1,J-1,I1,2,C5,X)          ALP10890
I1=KP1                               ALP10900
CALL CF(I+1,J+1,I1,2,C5,X)          ALP10910
CALL CF(I+1,J+1,I1,3,C9,X)          ALP10920
45 IF(I-1) 20, 20, 46                ALP10930
46 IF(MFLAT5) 50, 50, 20            ALP10940
50 RETURN                           ALP10950
END                                 ALP10960
*                                     ALP10970
FORTRAN                           ALP10980
SOURROUTINE GRANF (J)
DIMENSION RS(6)                     ALP10990
DIMENSION A(25), AA(25,6)           ALP11000
COMMON NDIM, NROW, NCOL, MM, KO, KP1, KP2, KM1, MFLAG, BC
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A
COMMON TMP, YK, YL
COMMON TNH, RHO, THC, R1H, D1R, D1H, T1H, C1H
COMMON XH, XK, GDXH, DN, DP
DIMENSION TMP(5184), YL(5184)
DIMENSION MM(25)
DO 2 I=37, 40
2 MM(1)=0
1 IF (J) 51, 51, 1
1 CONTINUE
JM=MM(J)
GDXH=XH/2.***JM
JM1=MM(J-1)
JM2= MM(J-2)
JP1= MM(J+1)
JP2= MM(J+2)
IF (J-2) 10, 10, 4
4 IF (J-NROW+1) 5, 30, 30
5 IF (JP2-JM2) 10, 20, 30
C MESH SPACING INCREASING
10 IF (JP1-JM) 11, 12, 12
C DOUBLE MESH SPACING AFTER THIS ROW
11 KP1=-1

```

```

KP2=-2          A1P11260
    17 IF (JP2-JP1) 40, 40, 99   A1P11270
    18 IF (JP2-JP1) 14, 18, 18   A1P11280
    19 KP2=-1          A1P11290
    20 TE(J-2) 40, 15, 15          A1P11300
    21 IF (JM-JM1) 16, 99, 99
C      MESH SPACING CHANGES IN BOTH DIRECTIONS          A1P11310
    22 KP2=-2          A1P11320
    23 GO TO 40          A1P11330
    24 IF (J-2) 40, 37, 13          A1P11340
    25 IF (JM-JM1) 19, 20, 20
C      DOUBLE THE MESH SPACING          A1P11350
    26 KP2=-2          A1P11360
    27 GO TO 40          A1P11370
    28 GO TO 99          A1P11380
C      MESH SPACING IS DECREASING          A1P11390
    29 IF (JM-JM1) 32, 37, 31
C      CUT MESH SPACING IN HALF          A1P11400
    30 KP1=1          A1P11410
    31 GO TO 40          A1P11420
    32 IF (JM1-JM2) 37, 37, 33
    33 IF (J-NROW+1) 35, 34, 34          A1P11430
C      MESH SPACING AT LAST ROW WAS CUT IN HALF          A1P11440
    34 KP2=1          A1P11450
    35 GO TO 99          A1P11460
    36 IF (JP1-JM) 34, 34, 36
C      SPACING CHANGES IN BOTH DIRECTIONS          A1P11470
    37 KP2=2          A1P11480
    38 GO TO 99          A1P11490
    39 IF (J-NROW+1) 38, 20, 20          A1P11500
C      MESH CUT IN HALF AFTER NEXT ROW          A1P11510
    40 KP2=2          A1P11520
    41 GO TO 99          A1P11530
C      SELECT COEFFICIENTS FOR CURRENT SPACING          A1P11540
    42 DO 45 K=JM+KADD          A1P11550
    43 KP2=2          A1P11560
    44 GO TO 99          A1P11570
    45 IF (JP1-JM) 20, 20, 39
C      MESH CUT IN HALF AFTER NEXT ROW          A1P11580
    46 KP2=2          A1P11590
    47 GO TO 99          A1P11600
C      SELECT COEFFICIENTS FOR CURRENT SPACING          A1P11610
    48 DO 45 L=1, 25          A1P11620
    49 IF (JM-KADD) 30, 30, 30          A1P11630
    50 KP2=2          A1P11640

```

```

45 A(L)=AA(L, K)          A1P11650
C=AA(26, K)                A1P11660
GO TO 99                   A1P11670
C FORM THE COEFFICIENTS FOR ALL SPACINGS
51 MIN=MM(1)                A1P11680
MAX=MIN                   A1P11690
DO 60 K=2, NROW             A1P11700
   IF (MAX-MM(K)) 54, 55, 55
54 MAX=MM(K)                A1P11710
   IF (MIN-MM(K)) 60, 60, 60
55 MIN=MM(K)                A1P11720
60 CONTINUE                 A1P11730
KAND=I-MIN                 A1P11740
DO 70 K1=MIN, MAX           A1P11750
   K=K1+KAND                 A1P11760
CALL COEFZ(K1)              A1P11770
DO 65 L=1,25                 A1P11780
   AA(L, K)=A(L)
65 AA(L, K)=R                A1P11790
   AA(26,K)=C
70 CONTINUE
*      FUNCTION INDEX(IT, JJ, KK)
*      COMMON NDIM, NROW, NCOL, MM, MFLAG, B, C
*      COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
*      DIMENSION MM(40)
*      I=II
*      J=JJ
*      KK
*      IF (MFLAG) 100, 100, 200
100 CONTINUE
   IF (IOPT2) 10, 10, 20
10  IF (I-1) 1, 1, 5
   1 IF (K-1) 2, 2, 3
2 INDEX=(J-1)*2
   GO TO 50

```

```

3 INDEX= J*2-1          A1P12040
   GO TO 50              A1P12050
5 INDEX= 3*K-4           A1P12060
   GO TO 50              A1P12070
20 IF (I-1) 21, 21, 79   A1P12080
21 IF (K-1) 22, 22, 74   A1P12090
22 INDEX=J*2-1          A1P12100
   GO TO 50              A1P12110
24 INDEX=J*2             A1P12120
   GO TO 50              A1P12130
29 INDEX=3*K-3           A1P12140
   GO TO 50              A1P12150
200 IF (I-NCOL) 201, 201, 220
201 IF (J-NCOL) 202, 202, 210
202 IF (J) 230, 230, 205
205 INDEX= (I-1)*(NROW+6)*3+3*K+6
   GO TO 50              A1P12200
210 IF (LOPT2) 211, 211, 216
211 INDEX=3*(NROW+NCOL-I) +2+K
212 IF (I-1) 214, 214, 50
213 IF (K-3) 50, 215, 215
215 INDEX=INDEX-1
   GO TO 50              A1P12220
216 INDEX=3*(NROW+NCOL+NCOL-I)+K+5
217 IF (I-1) 212
   GO TO 212              A1P12230
220 INDEX=3*K-4
   IF (LOPT2) 50, 50, 221
221 INDEX=INDEX+3*NCOL+3
   GO TO 50              A1P12240
230 INDEX=3*I+K-4
   IF (I-1) 231, 231, 50
231 IF (K-1) 232, 232, 50
232 INDEX=INDEX+1
   GO RETURN              A1P12250
   END                   A1P12260
* FORTRAN
* SUBROUTINE INVERT(A,IMAX,ISING)
* SUBROUTINE TO INVERT A MATRIX
* DIMENSION A(72, 72), IN(72), TEMP(72)

```

```

ISING=0          A1P12430
N=IMAX          A1P12440
IMAXO=N-1       A1P12450
I1=1            A1P12460
1   I3=I1        A1P12470
IN(I1)=0         A1P12480
SUM=ARSF(A(I1,I1)) A1P12490
DO3 I=I1,N      A1P12500
IF(SUM-ARSF(A(I,I1)))2,3,3
2   I3=I          A1P12510
INT(I1)=I        A1P12520
SUM=ARSF(A(I,I1))
3   CONTINUE
IF(I3-I1)4,6,4
4   D05J=1,N      A1P12560
SUM=A(I1,J)      A1P12570
A(I1,J)=A(I3,J)
5   A(I3,J)=SUM  A1P12590
I3=I1+1          A1P12600
6   I3=I1+1      A1P12610
IF(A(I1,I1))97,99,97
97  D07I=I3,N    A1P12620
A(I1,I1)=A(I1,I1)/A(I1,I1)
7   A(I1,I1)=A(I1,I1)/A(I1,I1)
J2=I1-1          A1P12650
IF(J2)8,11,8
8   D09J=I3,N    A1P12660
A(I1,I1)=A(I1,I1)-DPSUM(A,I1,J,1,J2)
9   A(I1,I1)=A(I1,I1)-DPSUM(A,I1,J,1,J2)
11  J2=I1          A1P12680
I1=I1+1          A1P12690
DO12 I=I1,N      A1P12700
12  A(I1,I1)=A(I1,I1)-DPSUM(A,I1,J,1,J2)
IF(I1-N)1,14,1    A1P12720
14  DO60 JP=1,N    A1P12730
J=N+1-JP          A1P12740
A(I1,I1)=1.0/A(I1,I1)
IF(J-1)603,700,603 A1P12750
603  D060 IP=2,J    A1P12760
I=J+1-IP          A1P12770
IP=I+1            A1P12780
SUM=-DPSUM(A,I,J,IP0,J) A1P12790
A1P12800
A1P12810

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```

600 A(I,J)=SUM/A(I,I)
700 DO151 J=1,IMAX
    JPO=J+1
    DO151 I=JPO,N
        IMO=I-1
        SUM=-DPSUM(A,I,J,J+1,IMO)
        SUM=SUM-A(I,J)
    151 A(I,J)=SUM
        DO901 I=1,N
        DO900 J=1,N
            IF (I-J) 897,897,898
    897 TEMP(J)=DPSUM(A,I,J,J+1,N)
            TEMP(J)=TEMP(J)+A(I,J)
            GO TO 900
    898 TEMP(J)=DPSUM(A,I,J,I,N)
    900 CONTINUE
        DO901 J=1,N
    901 A(I,J)=TFMP(J)
        DO500 I=2,N
            M=N+1-I
            IF(IN(M))502,500,502
    502 ISS=IN(M)
        DO503 L=1,N
            SUM=A(L,ISS)
            A(L,ISS)=A(L,M)
    503 A(L,M)=SUM
            500 CONTINUE
            GO TO 805
    99 ITNG=1
    805 RETURN
    END
*   FORTRAN
    SUBROUTINE MTX(X, I, M)
    COMMON NDIM, NROW, NCOL, MM, I1, KP1, KP2, KM1, KM2, MFLAGS, B,C
    COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
    DIMENSION MM(35)
    DIMENSION X(72, 72)
    T1=0
    DO 1 II=1, NDIM

```

```

DO 1 JJJ=1, NDIM
1 X(TT, JJ)=0.
DO 100 J=1, NROW
  IF (IOPT3) 50, 50, 7
  7 CALL GRADE(TJ)
  50 IF (J-1) 57, 57, 58
  57 IF (IOPT2) 59, 59, 58
  58 TT=TT+1
    CALL EQ1 (X, II, I, J, M)
  59 IF (I-1) 61, 61, 60
  60 II=II+1
    CALL EQ2 (X, II, I, J, M)
  61 II=II+1
    CALL EQ3 (X, II, I, J, M)
100 CONTINUE
      RETURN
END
* FORTRAN
* LIST
      SUBROUTINE MTXS (X, V, II, M)
COMMON NDIM, NROW, NCOL, MM, MFLAG, R, C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P13440
  1 C9, C10, C11, C12
DIMENSION MM(40)
C MULTIPLICATION BY A SINGLE DIAGONAL
DIMENSION X(72, 72), Y(72, 72)
I=II
3 IF (M-1) 20, 100, 100
C CLEAR OUT Y
20 DO 23 J=1, NDIM
  23 Y(J,1)=0.
  L1=1
    GO TO 2
100 L1=NDIM
  IF (M-1) 2, 2, 50
2  FORM A*X AND STORE IN Y
  2 DO 5 J=1, NROW
  5

```

```

IF (IOPT3) 7, 7, 6          A1P13600
6 CALL GRADE (J)           A1P13610
7 I1=INDX (I, J, 2)         A1P13620
I2=I1+1                     A1P13630
J1=INDX (I-2, J, 3)         A1P13640
J2=J1-1                     A1P13650
IF (I-3) 31, 31, 32         A1P13660
31 CC=0*                   A1P13670
GO TO 33                   A1P13680
32 CC=C2                   A1P13690
33 DO 4 K=1, L1             A1P13700
     Y(I1, K)=Y(I1, K)-X(J1, K)*B7
4   Y(I2, K)=Y(I2, K)+X(J2, K)*CC+X(J1, K)*C10
5 CONTINUE
GO TO 99
C FORM X*E AND STORE IN Y
50 CONTINUE
DO 60 J=1, NROW
IF (IOPT3) 52, 52, 51
51 CALL GRADE (J)
52 IF (I-1) 49, 49, 48
49 CC=2.*C10
BB=0.
GO TO 57
48 CC=C10
RB=R7
57 I2= INDX (I, J, 3)
I1=I2-1
J1=INDX (I+2, J, 2)
J2=J1+1
IF (I-1) 53, 53, 55
53 DO 54 K=I, L1
54 Y(K, J1)=Y(K, J1)-X(K, I2)*C2
55 DO 58 K=1, L1
58 Y(K, J2)=Y(K, J2)+X(K, I1)*BB+X(K, I2)*CC
60 CONTINUE
99 RETURN
END
* FORTRAN

```

SUBROUTINE NORM (X, CONS, K)

```

DIMENSION X(12), Y(12)
DO 1 I=1,K
 1 Y(I)=ARSF(X(I))
DO 10 I=2,K
 10 IF (Y(I)-Y(1)) 2,10,10
 2 TEMP=Y(1)
 2 Y(1)=Y(I)
 2 Y(I)=TEMP
10 CONTINUE
DO 20 I=1,K
 20 X(I)=X(I)/Y
CONS=CONS/Y
RETURN
END
*
FORTRAN
SUBROUTINE QROSTM (A, M, K)
DIMENSION A (270, 12)
N=271

```

```

ID=I+K
I1=1
1 I3=I1+1

```

```

I2=ID
M1=I1+K
IF (M1-M) 4, 4, 3
 3 M1=M

```

```

4 DO 5 I=12, M1
 5 I2=I2-1
 5 A(I,I2)=A(I,I2)/A(I,I)
 14 J2=I1-1
 13=I1+1
 5 IF (J2) 6, 11, 6
 6 K1=K
 12=ID
 5 M1= I1+K-I
 5 IF (M1-M) 8, 8, 7
 7 M1=M
 8 DO 9 J=13, M1
 9 K1= K1-1

```

```

K2 = XMAXOF(I1-K1, 1)
I2=I2+1
A(I1,I2) = A(I1,I2) - QDPSUM (A, I1, J, K2, J2)
I2=ID+K+1
J1=XMAXOF(I1-K, 1)
A(I1,I2) = A(I1,I2) - QDPSUM (A, I1, N, J1, J2)
IF (I1-M) I1, 18, 11
I1 J2=I1
I1=I1+1
K1=K+1
I2=ID+1
M1=I1+K-1
IF (M1-M) 13, 13, 12
I2 M1=M
DO 15 I=I1,M1
I2= I2-1
KI=KI-I
K2=XMAXOF(I1-K1, 1)
15 A(I,I2) = A(I,I2) - QDPSUM (A, I, I1, K2, J2)
IF (I1-M) I, I4, I
18 K1= ID+K+1
DO 17 I=1,M
J2= M-I
I3=J2+1
A(I3, K1) = A(I3,K1) / A(I3, ID)
IF (J2) 16, 16, 16
16 J1=XMAXOF (I3-K, 1)
DO 17 J=J1, J2
I4=I3+K+1-J
17 A(J,K1)=A(J,K1)-A(I3,K1)*A(J,I4)
19 RETURN
END
FORTRAN
SURROUNTNF <T0RF (X, II)
DIMENSION MM(40)
COMMON NDIM, NROW, NCOL, MM, MFLAG, B, C
COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
COMMON A1, A2, A3, A4, A5, A6
COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7, C8, A1P14760
* A1P14720
A1P14730
A1P14740
A1P14750

```

```

1 C9, C10, C11, C12 A1P14770
COMMON TMP, YK, YL
DIMENSION TMP (5184), YK (5184), YL (5184)
DIMENSION X(59) A1P14780
A1P14790
A1P14800
A1P14810
A1P14820
A1P14830
A1P14840
A1P14850
A1P14860
A1P14870
A1P14880
A1P14890
A1P14900
A1P14910
A1P14920
A1P14930
A1P14940
A1P14950
A1P14960
A1P14970
A1P14980
A1P14990
A1P15000
A1P15010
A1P15020
A1P15030
A1P15040
A1P15050
A1P15060
A1P15070
A1P15080
A1P15090
A1P15100
A1P15110
A1P15120
A1P15130
A1P15140
A1P15150

N2= (NROW+6)*3
NC1=NCOL+1
SN1=1.
SN2=1.
IF (LOPT5) 21, 21, 20 A1P14840
20 SN2=-1.
21 IF (LOPT6) 50, 50, 22 A1P14850
22 SN1=-1.
50 CONTINUE A1P14860
TF (MFLAG) 100, 100, 200
100 N1=(I-1)*N2+10 A1P14870
I2=0
DO 110 J=1, NROW
110 IF (J-1) 101, 101, 103 A1P14880
101 IF (LOPT2) 104, 104, 103 A1P14890
103 I2=I2+1
TMP(N1)=X(I2)
104 IF (I-1) 107, 107, 106 A1P14900
106 I2=I2+1
TMP (N1+1)=X(I2)
107 I2=I2+1
TMP (N1+2)=X(I2)
110 N1=N1+3
GO TO 999
200 IF (LOPT2) 220, 220, 201
201 I2=I2+270
DO 210 I=1, NCOL
N1= (I-1)*N2+1
I2=I2+1
TMP (N1+3)=YK (I2)
204 I2=I2+1
TMP (N1+4)=YK (I2)
208 I2=I2+1
TMP (N1+2)=YK (I2)

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    TMP(N1+5) = TMP(N1+11)*SN1          A1P15160
    N1=NC1*N2+4                          A1P15170
    TMP(N1)=YK(I2+1)                      A1P15180
    TMP(N1+1)=YK(I2+2)                    A1P15190
    TMP(N1+2)=YK(I2+3)                    A1P15200
    TMP(N1+6)=YK(I2+4)                    A1P15210
    I2=I2+4                                A1P15220
    GO TO 222                             A1P15230
220 I2=11*270                           A1P15240
222 NO=NCT1*N2+10                         A1P15250
    N1=NO+N2                            A1P15260
    NN=NO-N2-N2                         A1P15270
    TMP(N0+1)=YK(I2+1)                    A1P15280
    TMP(N1+2)=YK(I2+2)                    A1P15290
    I2=I2+2                                A1P15300
    TMP(N0+2)=TMP(NN+2)*SN2             A1P15310
230 DO 240 J=2, NROW
    NO=NO+3
    N1=N1+3
    NN=NN+3
    TMP(N0)=YK(I2+1)
    TMP(N0+1)=YKT(I2+2)
    TMP(N0+2)=TMP(NN+2)*SN2
    TMP(N1+2)=YK(I2+3)
240 I2=I2+2
    NO=NO+6
    TMP(N0)=YK(I2+1)
    TMP(N0+1)=YK(I2+2)
    TMP(N0+2)=YK(I2+3)
    I2=I2+3
    DO 250 T=1, NCOL
    I1=NCOL+1-I
    N1=I1*N2-5
    I2=I2+1
    TMP(N1)=YK(I2)
    TMP(N1+2)=TMP(N1-4)*SN1
    TF(TT-T) 252, 257, 251
251 I2=I2+1

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```

      TMP(N1+1)=YK(12)
      I2=12+
      TMP(N1+5)=YK(12)
      250 CONTINUE
      999 RETURN
      END
      *          FORTRAN
      SUBROUTINE STRFSS (TT,JJ,RR)
      COMMON NDIM, NROW, NCOL, MM, K0, KP1, KM1, KP2, MFLAG, B,C
      COMMON IOPT1, IOPT2, IOPT3, IOPT4, IOPT5, IOPT6, IOPT7, IOPT8
      COMMON A1, A2, A3, A4, A5, A6
      COMMON B1, B2, B3, B4, B5, B6, B7, C1, C2, C3, C4, C5, C6, C7,C8,A1P15660
      1 C9, C10, C11, C12
      COMMON TMP, YK, YL
      COMMON ZNU, RHO, THC, R1H, D1B, D1H, T1H, C1H
      COMMON XXH,XK,GDXH, PO, DP
      COMMON TE, TI, TO, OC, TC, TD, DDLD
      COMMON ROT
      COMMON EX, ENU, FTX, SC, DX, DTX, DN1
      DIMENSION ROT (80)
      DIMENSION MM(35)
      DIMENSION TMP(5184), YK(4,1296), YL(4,4,324)
      101 FORMAT (13,1H,13,1P9F12.4)
      J=JJ
      K=KK
      NC1=NCOL+1
      NR1=NROW+1
      IF (IOPT3) 1, 1, 5
      1 XH=XXH
      GO TO 10
      5 J1=XMAXOF (J, 1)
      10 I=II
      J1=MINT(J1)
      XH= XXH/2.**J1
      10 I=II
      TP3=RHO/THC
      UX=(DEF(I,J-1,1, KM1)-DEF(I,J+1,1,KP1))/({2.*XH})
      UT=(DEF(I+1,J,1, 0)-DEF(I-1,J, 1, 0))/({2.*XK})
      VX=(DEF(T,J-1,2,KM1)-DEF(T,J+1,2,KP1))/({2.*XH})
      VT=(DEF(I+1,J,2, 0)-DEF(I-1,J, 2, 0))/({2.*XK})
      A1P15550
      A1P15560
      A1P15570
      A1P15580
      A1P15590
      A1P15600
      A1P15610
      A1P15620
      A1P15630
      A1P15640
      A1P15650
      A1P15660
      A1P15670
      A1P15680
      A1P15690
      A1P15700
      A1P15710
      A1P15720
      A1P15730
      A1P15740
      A1P15750
      A1P15760
      A1P15770
      A1P15780
      A1P15790
      A1P15800
      A1P15810
      A1P15820
      A1P15830
      A1P15840
      A1P15850
      A1P15860
      A1P15870
      A1P15880
      A1P15890
      A1P15900
      A1P15910
      A1P15920
      A1P15930

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WXX= (DEF(I,J-1,3,KM1)-2.*DEF(I,J,3,0)+DEF(I,J+1,3,KP1))/XH**2 A1P15940
WXT= (DEF(I+1,J-1,3,KM1)-DEF(I+1,J+1,3,KP1))-DEF(I-1,J-1,3,KM1)+ A1P15950
1 DEF (I-1, J+1, 3, KP1) / (4.*XH*XK) A1P15960
WT= (DEF (I+1,J,3,0)-2.*DEF (I,J,3,0)+DEF (I-1,J,3,0))/XK**2 A1P15970
EPSX=TP3*UX A1P15980
EPST=VT/THC+DEF(I,J,3,0) A1P15990
TP4=FP$X A1P16000
TP5=EPST A1P16010
WX= (DEF(I,J-1,3,KM1)-DEF(I,J+1,3,KP1))/(2.*XH) A1P16020
WT=(DEF(I+1,J,3,0)-DEF(I-1,J,3,0))/(2.*XK) A1P16030
GAMEUT7THC+TP3*VX A1P16040
RX1=-TP3**2*WXX A1P16050
RX2=(VT-WTT/THC)/THC A1P16060
RX12=TP3*(VX-WXT/THC) A1P16070
IF (MFLAG) 20, 40, 40 A1P16080
20 CONTINUE A1P16090
OMEGAX=-TP3*WX A1P16100
OMEGAT=DEF(I,J,2,0)-WT/THC A1P16110
PHI=(TP3*VX-UT/THC)/2* A1P16120
WRITE OUTPUT TAPE 6, T01, J, I, FP$X, EPST, GAM, RX1, RX2, RX12, A1P16130
1 OMEGAX, OMEGAT, PHI A1P16140
GO TO 99 A1P16150
40 CONTINUE A1P16160
ZK=1./((2.*R1H**2)) A1P16170
UXX= (DEF(I,J-1,1,KM1)-2.*DEF(I,J,1,0)+DEF(I,J+1,1,KP1))/XH**2 A1P16180
VXX= (DEF(I,J-1,2,KM1)-2.*DEF(I,J,2,0)+DEF(I,J+1,2,KP1)) / ( A1P16190
1 XH*XH) A1P16200
VXT= (DEF(I+1,J-1,2,KM1)-DEF(I+1,J+1,2,KP1))-DEF(I-1,J-1,2,KM1) A1P16210
1 ) + DEF(I-1,J+1,2,KP1)/(4.*XH*XK) A1P16220
VTT=(DEF(I+1,J,2,0)-2.*DEF(I,J,2,0)+DEF(I-1,J,2,0))/XK**2 A1P16230
WXXX= (DEF(I,J-2,3,KM2)-2.*DEF(I,J-1,3,KM1)+2.*DEF(I,J+1,3,KP1)) A1P16240
1 -DEF(I,J+2,3,KP2))/(2.*XH**3) A1P16250
WXT= (DEF(I+1,J-1,3,KM1)-2.*DEF(I,J-1,3,KM1)+2.*DEF(I,J+1,3,KP1))/(2.* A1P16270
1 ) -DEF(I+1,J+1,3,KP1)+DEF(I-1,J-1,3,KM1)-DEF(I-1,J+1,2,KP1) / (2.* A1P16270
2 XH*XK*XK) A1P16280
WTT= (DEF(I+2,J,3,0)-2.*DEF(I+1,J,3,0)+2.*DEF(I-1,J,3,0)-DEF(I-2,A1P16290
1 J,3,0))/(2.*XK**3) A1P16300
WXT= (DEF(I+1,J-1,3,KM1)-2.*DEF(I+1,J,3,0)+2.*DEF(I-1,J,3,0)+ A1P16310
1 DEF(I+1,J+1,3,KP1)-DEF(I-1,J-1,3,KM1)-DEF(I-1,J+1,3,KP1))/(2.* A1P16320

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2 XH*XH*XX) ALP16330
ANX=EPSX*EX+EPST*ENU-SC*RX1 ALP16340
ANX=ANX-T C ALP16350
ANT=EPSX*ENU+FPST*EX-SC*RX2 ALP16360
ANT=ANT-T C ALP16370
ANT=FTX*GAM ALP16380
ANXT=ANTX+DTX*RX12 ALP16390
AMX=DX*RX1+DNU*RX2-SC*EPST ALP16400
AMX=AMX-TD ALP16410
AMT=DNU*RX1+DX*RX2-SC*EPST ALP16420
AMT=AMT-TD ALP16430
AMXT=DTX*RX12 ALP16440
AQX=-DX*TP3**3*WXXX-SC*TP3**2*UXXX+(DNU+DTX)*TP3/THC*(VXT-WXTT/THC) ALP16450
AQT=DTX*TP3**2*VXXX+(DX-SC)*VTTTHC**2-(DTX+DNU)*TP3**2*WXXXT/THC ALP16460
1 -DX*WTTT/THC**3-SC*WT/THC ALP16470
IF (K) 49, 49, 47 ALP16480
47 YL(1, K, I)=ANX ALP16490
YL(2, K, I)=ANT ALP16500
YL(3, K, I)=AMX ALP16510
YL(4, K, I)=AMT ALP16520
49 CONTINUE ALP16530
IF (MFLAG) 45, 45, 46 ALP16540
45 WRTF'OUTPUT-TAPF 6,101,J,I, AMX , ANT, ANXT, AGX , AQT ALP16550
GO TO 48 ALP16560
46 WRTF'OUTPUT TAPF 6,101,J,I, ANX, ANT, ANXT, ANT X ALP16570
48 CONTINUE ALP16580
IF (J) 60, 60, 51 ALP16590
51 IF (NR1-J) 70, 70, 52 ALP16600
52 IF (NC1-T) 80, 80, 80 ALP16610
C LOWFR BOUNDARY ALP16620
60 J3=NC1+NR1+2+NC1-1 ALP16630
61 YK(T,J3)=ANXT+2.*ZK*AMXT ALP16640
YK(2, J3)=ANX ALP16650
YK(3, J3)=(AGX+(1.-ZN1))/THC *TP3*(VXT-WXTT/THC ) ) ALP16660
YK(4, J3)=AMX ALP16670
IF (I-NC1) 99, 69, 69 ALP16680
C SPFCIAL POINT ALP16690
69 KT=2*T(NCI+NR1)+1 ALP16700
YK(1, K1)=-2.*AMXT ALP16710

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GO TO 80
C   UPPER BOUNDARY
    70 YK(1, I)=ANXT+2.*ZK*AMXT
    YK(2, I)=ANX
    YK(3, I)=-TAOX+(1.-ZNU)/THC *TP3*(VXT-WXTT/THC )
    YK(4, I)=AMX
    IF (I-NCL) 97, 79, 79
C   SPECIAL CORNER POINT
    79 K1=2*(NC1+NR1)
    YK(1, K1)=2.*AMXT
    YK(1, J3)=NC1+NR1+1-J
    YK(1, J3)=ANT
    YK(2, J3)=ANT
    YK(3, J3)=AC1+(1.-ZNU)*TP3*TP3*(VXX-WXXT/THC )
    YK(4, J3)=AMT
    97 CONTINUE
    99 RETURN
    END
*   FORTRAN
      SUBROUTINE ZFRC (X,N)
      DIMENSION X(8500)
      DO 1 T=1,N
      1 X(I)=0.
      RETURN
      END
*   FAP
      COUNT 150
      ENTRY CLOCK
      ENTRY WCKA
      SUBROUTINE CLOCK
      ENTRY WCKP
      ****
*   THIS SUBROUTINE PLACES THE CLOCK IN A GIVEN LOCATION SPECIFIED
*   BY THE CALL STATEMENT THAT IS.
*   CALL CLOCK LOCATION TO E STORED IN FORTRAN OR
*   CALL CLOCK (FOR ABSOLUTE ASSEMBLIES USE TSX CLOCK
*   PZE (LOCATION TO RE STORED) IN FAP
*   OR
      A1P16720
      A1P16730
      A1P16740
      A1P16750
      A1P16760
      A1P16770
      A1P16780
      A1P16790
      A1P16800
      A1P16810
      A1P16820
      A1P16830
      A1P16840
      A1P16850
      A1P16860
      A1P16870
      A1P16880
      A1P16890
      A1P16900
      A1P16910
      A1P16920
      A1P16930
      A1P16940
      A1P16950
      A1P16960
      A1P16970
      A1P16980
      A1P16990
      A1P17000
      A1P17010
      A1P17030
      A1P17020
      *A1P17040
      *A1P17050
      *A1P17060
      *A1P17070
      *A1P17080
      *A1P17090
      *A1P17100

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*	SVN	(LOCATION TO BE STORED)	GIVES THE CLOCK IN FLOATING POINT	*A1P17110
*	PZE	(LOCATION TO BE STORED)	GIVES THE CLOCK IN ECD	*A1P17120
*	IN FORTRAN THE CLOCK WILL ALWAYS BE GIVEN IN FLOATING POINT			*A1P17130
*	THE CLOCK IS ALSO PRINTED ON-LINE			*A1P17140
*		*****		*A1P17150
	CLOCK	SPACE 3		A1P17160
	BSS	0		A1P17170
	RSC	0		A1P17180
	BSS	0		A1P17190
	SXA	CLOCK+55,1		A1P17200
	SXA	CLOCK+56,2		A1P17210
	SXA	CLOCK+57,4		A1P17220
	RRA			A1P17230
	SPRA	10		A1P17240
	RCHA	CLOCK+105		A1P17250
	TCOA	*		A1P17260
	STZ	CLOCK+61		A1P17270
	STZ	CLOCK+62		A1P17280
	CLA	CLOCK+227		A1P17290
	STO	CLOCK+218		A1P17300
	AXT	0,3		A1P17310
	AXT	2,4		A1P17320
	LDQ	CLOCK+104,1		A1P17330
	PXD	0,0		A1P17340
	CAQ	CLOCK+153,0,1		A1P17350
	CVR	CLOCK+217,0,6		A1P17360
	ORS	CLOCK+63,4		A1P17370
	TIX	*-4,4,1		A1P17380
	CLA	CLOCK+226,2		A1P17390
	STO	CLOCK+218		A1P17400
	TIX	*+1,2,1		A1P17410
	TXI	*+1,1,2		A1P17420
	TXL	CLOCK+12,1,17		A1P17430
	PXD	0,0		A1P17440
	LDQ	CLOCK+61		A1P17450
	LGL	30		A1P17460
	LDQ	CLOCK+62		A1P17470
	LGL	18		A1P17480
	ROL	6		A1P17490

IOC P	CLOCK + 83, , 2	WRITE	
TO CP	CLOCK + 87, , 2	READ	AIP17900
IOC P	CLOCK + 85, , 2	WRITE	AIP17910
IOC D	CLOCK + 89, , 16	READ	AIP17920
PZF	*+10		AIP17930
PZE	*+9, , 1000		AIP17970
PZE	*+8, , 2000		AIP17940
PZE	*+7, , 3000		AIP17950
PZE	*+6, , 4000		AIP17960
PZE	*+5, , 5000		AIP17990
PZF	*+4, , 6000		AIP18000
PZF	*+3, , 7000		AIP18010
PZF	*+2, , 8000		AIP18020
PZE	*+1, , 9000		AIP18030
PZE	*+10		AIP18040
PZE	*+9, , 100		AIP18050
PZE	*+8, , 200		AIP18060
PZE	*+7, , 300		AIP18070
PZE	*+6, , 400		AIP18080
PZE	*+5, , 500		AIP18090
PZE	*+4, , 600		AIP18100
PZF	*+3, , 700		AIP18110
PZF	*+2, , 800		AIP18120
PZE	*+10		AIP18130
PZE	*+9, , 10		AIP18140
PZE	*+8, , 20		AIP18150
PZE	*+7, , 30		AIP18160
PZE	*+6, , 40		AIP18170
PZE	*+5, , 50		AIP18180
PZE	*+4, , 60		AIP18190
PZF	*+3, , 70		AIP18200
PZF	*+2, , 80		AIP18210
PZF	*+1, , 90		AIP18220
PZE			AIP18230
PZF	*+1		AIP18240
PZF	*+2		AIP18250
PZE	*+3		AIP18260
PZE	*+4		AIP18270

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PZE      ,5
PZE      ,6
PZE      ,7
PZE      ,8
PZF      ,9
BCI      8,00000000001000010001100010000010001100001111
RCI      8,0010000110010010010011000110001111
RCT      8,010000100010010010011000110001111
RCI      8,011000110010010010011000110001111
RCT      8,010000110010010010011000110001111
RCT      8,100001000110010010011000110001111
RCT      8,10100010010010010011000110001111
RCT      8,11000110010010010011000110001111
RCT      8,111000110010010010011000110001111
PZE      *
PZE      *-1
PON      *-2,*,4096
PON      **-3
PZF      *-4,*,7*4096
PZF      *-5,*,6*4096
PZF      *-6,*,5*4096
PZE      *-7,*,4*4096
PZE      *-8,*,3*4096
PZE      *-9,*,2*4096
PZE      *-10,*,1*4096
END      *
FAP      COUNT   50
* DPSUM FUNCTION DPSUM(A,I,J,K,M). COMPUTE SUM OF A(I,L)*A(L,J), L=K,M.
* SET DPSUM=0 IF K GRTR THAN M
* SET THE VFLD OF RDIM AND ROWDIM TO THE ROW DIMENSION OF A
* ENTRY DP SUM
DPSUM    CLA*   5,4      LOAD M
          CAS*   4,4      COMPARE WITH K
          TRA    *+4      M IS BIGGER
          TRA    *+3      M=K
          PXA    0,0      M LESS THAN K      SET DPSUM=0
          TRA    6,4      AXT,1
          SXA    AXT+1,2
          SXA    AXT+1,2

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			A1P18670
STD	TEST		
CLA	1,4		A1P18680
ADD	=1		A1P18690
STA	AIL		A1P18700
CLA*	2,4		A1P18710
ARS	18		A1P18720
STA	1		A1P18730
CLA*	4,4		A1P18740
PDX	,2		A1P18750
ARS	18		A1P18760
SUB	=1		A1P18770
XCA			A1P18780
MPY	RDIM		A1P18790
XCA			A1P18800
ADD	1		A1P18810
PAX	,1		A1P18820
CLA*	3,4		A1P18830
ARS	18		A1P18840
SUB	=1		A1P18850
XCA			A1P18860
MPY	RDIM		A1P18870
XCA			A1P18880
SSM			A1P18890
ADD	AIL		A1P18900
STA	ALJ		A1P18910
STZ	SUM		A1P18920
STZ	SUM+1		A1P18930
LDO	*,1		A1P18940
AL			A1P18950
AL	FMP	**,2	A1P18960
DFAO	SUM		A1P18970
DST	SUM		A1P18980
TXI	**1,1,ROWDIM		A1P18990
TXI	**1,2,1		A1P19000
TXL	AIL,2,**		A1P19010
AXT	**,1		A1P19020
AXT	**,2		A1P19030
TRA	6,4		A1P19040
R	COMMON 1		A1P19050
	SUM FQU R+202		

I PZE
 RDIM FOU 72
 RDIM DFC 72
 END

*	FAP	COUNT 90		A1P19060
*		MATRIX MULTIPLICATION IN DOUBLE PRECISION		A1P19070
*	MATM	TX, Y, Z, NR, NC, NCY)		A1P19080
*		COMPUTE Z=X*Y. NR AND NC ARE NUMBER OF ROWS AND COLUMNS IN X.	NCYA1P19140	
*		IS NUMBER OF COLUMNS IN Y. Z=Y OK. BUT NOT 7=X.	A1P19150	
	ENTRY MATM			A1P19160
MATM	SXA	AXT+1,1		A1P19170
	SXA	AXT+1,2		A1P19180
	SXA	AXT+2,4		A1P19190
	CLA	1,4	X-MATRIX	A1P19200
	STA	AX		A1P19210
	CLA	2,4		A1P19220
	STA	AY		A1P19230
	CLA	3,4		A1P19240
	STA	AZ		A1P19250
	CLA*	4,4	NR OF ROWS IN X	A1P19260
	ARS	18		A1P19270
	STA	NR		A1P19280
	CLA*	5,4	NR OF COLUMNS IN X	A1P19290
	ARS	18		A1P19300
	STA	NC		A1P19310
	CLA*	6,4		A1P19320
	ARS	18		A1P19330
	STA	NCY		A1P19340
	CLA	=1		A1P19350
	STO	J		A1P19360
MTO	LDQ	RWDIM		A1P19370
	MPY	J		A1P19380
	XCA			A1P19390
	SSM			A1P19400
	ADD	AZ		A1P19410
	ADD	RWDIM		A1P19420
	ADD	=1		A1P19430
	STA	AZIJ		A1P19440

LDO	RWDIM	J		A1P19450
MPY				A1P19460
XCA				A1P19470
SSM				A1P19480
ADD	AY			A1P19490
ADD	RWDIM			A1P19500
ADD	=1			A1P19510
STA	YCOL			A1P19520
LXA	NR,1			A1P19530
SXD	MT4,1			A1P19540
AXT	I,1			A1P19550
LXA	NC,2			A1P19560
CLA	**,2			A1P19570
YCOL	TEM,2			A1P19580
STO				A1P19590
TIX	YCOL,2,1			A1P19600
SXA	I,1			A1P19610
MT1				A1P19620
LDO	RWDIM			A1P19630
MPY	NC			
XCA				
PAX	*4			A1P19640
CLA	RWDIM			A1P19650
AND	AX			A1P19660
AND	=1			A1P19670
SUB	I			A1P19680
STA	MT2			A1P19690
STA	AXTK			A1P19700
LXA	NC,2			A1P19710
STZ	TFM			A1P19720
STZ	TFM+1			A1P19730
PXA	0,0			A1P19740
NZT	**,4			A1P19750
MT2	TXT	MT3+1,2,-1		A1P19760
AXTK	LNO	**,*4		A1P19770
AYKJ	FMP	TEM,2		A1P19780
DFAO		TEM		A1P19790
DST	TEM			A1P19800
MT3	TNX	*+2,2,1		A1P19810
AZIJ	TIX	MT2,4,RWDIM		A1P19820
	STO	**,*1		A1P19830

M T 4	T X 1	*+1,1,1,1	M T 1,1,1,*	**=NR	A1P19840
	CLA	J			A1P19850
	AND	=1			A1P19860
					A1P19870
	CAS	NCY			A1P19880
	TRA	AXT			A1P19890
	TRA	*+1			A1P19900
	STO	J			A1P19910
	TRA	MTO			A1P19920
	AXT	***,1			A1P19930
	AXT	***,2			A1P19940
	AXT	***,4			A1P19950
	TRA	7,4			A1P19960
RWDIM	DEC	72			A1P19970
ROWN	FQU	72			A1P19980
TEM	ROOL	130			A1P19990
AX	PZE	0			A1P20000
AY	PZE	0			A1P20010
AZ	PZF	0			A1P20020
NR	PZE	0			A1P20030
NC	PZE	0			A1P20040
NCY	PZE	0			A1P20050
I	PZF	0			A1P20060
J	PZF	0			A1P20070
	FND				A1P20080
	FAP				A1P20090
	COUNT	40			A1P20100
	*PLRL	SURROUTINE PLRL (CRLRL, NCR)			A1P20110
	ENTRY	PLRL			A1P20120
	PLRL	SXA	AXT,1		A1P20130
		SXA	AXT+1,2		A1P20140
	NZT*	2,4			A1P20150
					A1P20160
	RIGHT	TRA	UPPFR		A1P20170
		CLA	1,4		A1P20180
					A1P20190
					A1P20200
RT	CLA	RT+6,1			A1P20210
	STO	***,2			A1P20220

	TNX	AXT,1,1	A1P20230
UPPFR	TXT	R1,2,1	A1P20240
	CLA	1,4	A1P20250
	STA	UP+1	A1P20260
	AXT	J2,I	A1P20270
	AXT	0,2	A1P20280
UP	CLA	UPP+12,1	A1P20290
	STO	**?,?	A1P20300
	TXN	AXT,1,1	A1P20310
	TXI	UP,2,1	A1P20320
AXT	AXT	**?,?	A1P20330
	AXT	**?,?	A1P20340
	TRA	3,4	A1P20350
PT	PTT	6, PTTHT BOUNDARY	A1P20360
UPD	RCI	6, CURVE 1= UPPER BOUNDARY	A1P20370
	RCI	6, CURVE 2= LOWER BOUNDARY	A1P20380
	FND		A1P20390
*	FAP	COUNT 80	A1P20400
		*PLTLBL SUBROUTINE PLTLBL (DTSP, STRSS, ORLBL, ARLBL, BOND)	A1P20420
	PLTLBL	ENTRY PLTLBL	A1P20430
	PLTLBL	STA AXT,1	A1P20440
		SXA AXT+1,2	A1P20450
D1	PLTLBL	CLA 1,4	A1P20460
		STA n1+1	A1P20470
		AXT 9,1	A1P20480
		AXT 0,2	A1P20490
D1	PLTLBL	CLA DISP+9,1	A1P20500
		STO **?,2	A1P20510
		TXN *+2,1,1	A1P20520
		TXI D1,2,1	A1P20530
		CLA 2,4	A1P20540
		STA s1+1	A1P20550
		AXT 0,2	A1P20560
S1	PLTLBL	CLA CTRSS+0,1	A1P20570
		STO **?,2	A1P20580
		TXN *+2,1,1	A1P20590
		TXI S1,2,1	A1P20610

	CLA	3,4		
	STA	01+1		A1P20620
	AXT	66,1		A1P20630
	AXT	0,2		A1P20640
	AXT	0,2		A1P20650
01	CLA	ORDU+66,1		A1P20660
	STO	**,2		A1P20670
	TXI	*+2,1,1		A1P20680
	TXI	01,2,1		A1P20690
	CLA	4,4		A1P20700
	STA	A1+1		A1P20710
	AXT	6,1		A1P20720
	AXT	0,2		A1P20730
	CLA	A8LRL+6,1		A1P20740
A1	STO	**,2		A1P20750
	TXN	*+2,1,1		A1P20760
	TXI	A1,2,1		A1P20770
	CLA	5,4		A1P20780
	STA	R1+1		A1P20790
	AXT	9,1		A1P20800
	AXT	8,2		A1P20810
	CLA	R0N+0,1		A1P20820
R1	STO	**,2		A1P20830
	TXI	B1,2,1		A1P20840
	AXT	**,1		A1P20850
	TXN	*+2,1,1		A1P20860
	TXI	B1,2,1		A1P20870
	AXT	**,2		A1P20880
	TRA	6,4		A1P20890
	ORDU	BCI	6,	U
	ORDV	BCI	6,	V
	ORDW	BCI	6,	W
	ORDNX	BCI	6,	NX
	ORDNT	BCI	6,	NTHFTA
	ORDMX	BCI	6,	MX
	ORDMT	BCI	6,	MTHETA
	ORDN	BCI	6,	NTAN
	ORDNB	BCI	6,	NNORM
	ORDQ	BCI	6,	Q
	ORDM	BCI	6,	M
	ABLBL	BCI	6,	THETA

DISP	BCI	9,	FIG.	CYLINDER DISPLACEMENT COMPONENTS	A1P21010
STRESS	BCI	9,	FIG.	CYLINDER STRESS RESULTANTS	A1P21020
BON	BCI	9,	FIG.	BOUNDARY STRESS RESULTANTS	A1P21030
END					A1P21040
*	FAD				A1P21050
	COUNT	70			A1P21060
*	FUNCTION	ONDCLM (A, T, J, K2, J2)			A1P21070
	ENTRY	ONPSUM			A1P21080
QDPSUM	SXA	AXT,1			A1P21090
	SXA	AXT+1,2			A1P21100
	SXA	AXT+2,4			A1P21110
	CLA	1,4			A1P21120
	STA	DP1			A1P21130
	STA	DP1+1			A1P21140
	LXA	PWMD,1			A1P21150
	SXD	DP2,1			A1P21160
CLA*	7,4				A1P21170
ARS	18				A1P21180
STA	1				A1P21190
CLA*	4,4				A1P21200
ARS	18				A1P21210
STA	K2				A1P21220
ADD	K				A1P21230
SUB	T				A1P21240
XCA					A1P21250
MDY	DPMD				A1P21260
XCA					A1P21270
ADD	T				A1P21280
SUR	=T				A1P21290
DAY	9,1				A1P21300
CLA*	2,4				A1P21310
ARS	18				A1P21320
STA	T				A1P21330
CLA	ROWD				A1P21340
ADD	=T				A1P21350
SUB	I				A1P21360
TZF	DP				A1P21370
CLA	ROWN				A1P21380
SUR	=1				A1P21390

PAC	*2		A1P21400
SXN	NP3,?		A1P21410
CLA	I		A1P21420
ADD	K		A1P21430
SUR	K?		A1P21440
XCA			A1P21450
MPY	ROWD		A1P21460
XCA			A1P21470
SUR	=1		A1P21480
ADD	K?		A1P21490
PAX	*2		A1P21500
TRA	NP0		A1P21510
DOP	AXT	1,2	A1P21520
SXN	NP3,?		A1P21530
CLA	K		A1P21540
ADD	K		A1P21550
ADD	=1		A1P21560
XCA			A1P21570
MPY	ROWD		A1P21580
XCA			A1P21590
ADD	K2		A1P21600
SUR	=1		A1P21610
DAX	*2		A1P21620
DOP	CLA*	5,4	A1P21630
ARC	18		A1P21640
SUR	K?		A1P21650
ADD	=1		A1P21660
PAX	,4		A1P21670
STZ	TFMP		A1P21680
STZ	TEMP+1		A1P21690
DOP	LDO	**,1	A1P21700
FMP	**,2		A1P21710
DFAD	TFMP		A1P21720
DST	TEMP		A1P21730
DOP	XTI	*+1,1,**	A1P21740
DOP	TXI	*+1,2,**	A1P21750
TXI	NP1,4,1		A1P21760
AXT	AXT	**,1	A1P21770
AXT	AXT	**,2	A1P21780

	AXT	***,4		
ROWD	TRA	6,4	A1P21790	
K	DEC	270	A1P21800	
K	DEC	5	A1P21810	
I	PZE		A1P21820	
K>	PZF		A1P21830	
K>	TEMP	BOOL	A1P21840	
	END		A1P21850	
*	FAD		A1P21860	
	COUNT	170	A1P21870	
*	SURROUNTMES	RTAPEF, WTAPF, RACK, AND RFSET	A1P21880	
	ENTRY	WTAPE	A1P21890	
	ENTRY	RTAPE	A1P21900	
	ENTRY	REW	A1P21910	
	ENTRY	BACK	A1P21920	
	ENTRY	RESET	A1P21930	
WTAPE	CLA	WRS	A1P21940	
	STD	WRTP	A1P21950	
	STD	FLAG1	A1P21960	
	TRA	REGTN	A1P21970	
RTAPE	CLA	RDS	A1P21980	
	STD	WRTP	A1P21990	
	STD	FLAG1	A1P22000	
BEGIN	SXA	AXT,1	A1P22010	
	SXA	AXT+1,2	A1P22020	
	STZ	FLAG	A1P22030	
	CLA*	1,4	A1P22040	
	STA	BSR	A1P22050	
	ANA	MASK	A1P22060	
	TZF	*+,2	A1P22070	
	STL	FLAG2	A1P22080	
	TRA	*+,2	A1P22090	
	STZ	FLAG2	A1P22100	
NZT*	NZT*	4,4	A1P22110	
	TRA	WR	A1P22120	
CK	TZF	CKR	A1P22130	
CKA	TCA	*	A1P22140	
	TRCA	ERR	A1P22150	
	TRA	AXT	A1P22160	
			A1P22170	

CKR	TOR	*						
	TRCB	FRR						A1P22180
	TRA	AXT						A1P22190
	ZFT	FLAG	TAPPF ERROR.	IS THIS THE FIRST TIME				A1P22200
ERD	TRA	PLNK	NM					A1P22210
	STL	FLAG						A1P22220
RSP	RSP	**						A1P22230
	CLA*	3,4						A1P22240
WDP	STD	IOC	WORD COUNT					A1P22250
	ARS	18						A1P22260
	SSM							A1P22270
	ADD	2,4						A1P22280
	ADD	=1	BOTTOM OF ARRAY IN AC					A1P22290
	STA	10C						A1P22300
	CLA*	1,4						A1P22310
	STA	WRTP	SFT TAPE UNIT ADDRESS					A1P22320
	ANA	TPNR	PICK OUT LAST DIGIT					A1P22330
	DAX	,1	TAPE M.	M TO XR 1				A1P22340
	CLA	TPCNT+1,1						A1P22350
	STA	LDI						A1P22360
	STA	COMP						
	STA	STORE						
	PDX	,2						A1P22370
	LDO	**,2						A1P22380
	ZET	FLAG	IF RFRUN, DONT BUMP COUNTERS					A1P22390
	TRA	RFRUN						
	SSP		NOT A BACKSPACE OR SIGN PLUS					
	ADD	ONF						
	PDX	,2						A1P22400
	STD	TPCNT+1,1						A1P22410
	STA	TPCNT+1,1						A1P22420
	ZET	FLAG	TS TT READ OR WRITE					A1P22430
	TRA	WR1	READ OPERATION					A1P22440
	XCA	WRITE OPERATION.	MUST MAKE UP NEXT RECORD ID NR.					A1P22450
	ADD	ONF						
	ADD	=1						A1P22460
	STO	**,2	STORF NEXT RECORD ID WORD IN TABLE					A1P22470
	STO	IDENT	FIRST WORD OF RECORD					A1P22480
WPI	NZT	FLAG2	CHANNEL A OR B					A1P22490
								A1P22500
								A1P22510
								A1P22520
								A1P22530
								A1P22540
								A1P22550
								A1P22560

TDA	TDA	*		A1P22580
TRCA	*+1			A1P22590
XFC	WDT P			A1P22600
RCHA	ICID			A1P22610
LCHA	IOC			A1P22620
ZET	FLAG1	IF RFADING.	GO CHECK RFCORD ID	A1P22630
TRA	COMP			A1P22640
ZET	FLAG	IS THIS A RERUN		A1P22650
TRA	CKA	YFS.	GO AND CHECK FOR REDUNDANCY	A1P22660
TRA	AXT	NO.	GO BACK AND COMPUTE SUM MORE	A1P22670
TPR	TCOR	*		A1P22680
	TRCR	**+1		A1P22690
XEC	WRTD			A1P22700
RCHA	TCIN			A1P22710
LCHR	IOC			A1P22720
ZFT	FLAG1			A1P22730
TRA	COMP			A1P22740
ZET	FLAG			A1P22750
TRA	TKR			A1P22760
TRA	AXT			A1P22770
COMP	CLA	**+,2		A1P22780
SIR	TDENT	DOFS FIRST WORD OF RECORD AGREE WITH TH		A1P22790
TNZ	SHIFT	NO.	TRY ANOTHER RECORD	A1P22800
STP	TPCNT+1,1			A1P22810
NZT	FLAG	YFS.	THIS A RFDTN	A1P22820
TRA	AXT	IF NOT,	GO BACK	A1P22830
CLA	FLAG2			A1P22840
TRA	CK			A1P22850
RERUN	ZET			A1P22860
TRA	FLAG1			A1P22870
XCA	RESFT ID WORD			A1P22880
TRA	STORE+1			A1P22890
ALNY	ZFT	FLAG1	IS THIS A READ	A1P22900
XFC	TRA	OUT	IF SO GIVE UP	A1P22910
XFC	WDT P	SCP		A1P22920
XFC	WRT P			A1P22930
XFC	WDT D			A1P22940
				A1P22950

CLA	MAX	=1		A1P22960
STIR	QUIT	QUIT AFTER MAX TIMES		A1P22970
TMI	MAX			A1P22980
STO	MAX			A1P22990
CLA	IDENT			A1P23000
TRA	STORF-1			A1P23010
CLA	MAX			A1P23020
SHIFT	STIR	=1		A1P23030
CLA	MAX			A1P23040
TRA	OUT			A1P23050
STO	MAX			
CLA	TPCNT+1,1			A1P23060
TPL	WR1	LAST ACTIVITY WAS RFAD OR WRITE SO GO AHEAD		A1P23070
XFC	BSR	LAST ACTIVITY WAS BACKSPACE. MUST GO BACK		A1P23080
XFC	BSR			A1P23090
TRA	WR1	LOOK FOR MISSING RECORD AGAIN		A1P23100
AXT	AXT	**,1		A1P23110
AXT	AXT	***,2		A1P23120
TRA	5,4			A1P23130
OUT	CALL	DUMP		A1P23140
RFW	CLA*	1,4		A1P23150
STA	STA	*+,1		A1P23160
RFW	RFW	**		A1P23170
SXA	SXA	AX,1		A1P23180
ANA	ANA	TPNR		A1P23190
PAX	PAX	,1		A1P23200
PXA	PXA	0,0		A1P23210
STD	STD	TPCNT+1,1		A1P23220
AX	AXT	**,1		A1P23230
TRA	TRA	2,4		A1P23240
ACK	CLA*	1,4		A1P23250
STA	STA	BSR		A1P23260
XFC	XFC	BSR		A1P23270
SXA	SXA	BX,1		A1P23280
ANA	ANA	TPNR		A1P23290
DAx	DAx	,1		A1P23300
CLA	DAx	TPCNT+1,1		A1P23310
SSP	SSP			A1P23320
STIR	STIR	**ITE		A1P23330
SSM	SSM			A1P23340

BX	STO	TPCNT+1,1	A1P23350
	AXT	'1	A1P23360
	TRA	2,4	A1P23370
	NOP		A1P23380
PF _{EF} T	PXA	0,0	A1P23390
	STD	TPCNT-3	A1P23400
	STD	TPCNT-2	A1P23410
	STD	TPCNT-1	A1P23420
	STD	TPCNT	A1P23430
ICTD	LOST	IDENT,1	A1P23450
INC	IORT	0,0	A1P23460
WRTP	WRS	**	A1P23470
WRS	RDS	**	A1P23480
RDS	PZE	TRL4	A1P23490
	DZF	TRL3	A1P23500
		TRL2	A1P23510
TCNT	TRL1		A1P23520
TPNR	OCT	'7	A1P23530
MASK	OCT	00000001000	A1P23540
MAX	DFC	40	A1P23550
ONE	OCT	1000000	A1P23560
FLAG			A1P23570
FLAG1			A1P23580
FLAG2			A1P23590
IDENT	PZE		A1P23600
TRL1	RFS	80	A1P23610
TRL2	RFS	80	A1P23620
TRL3	RFS	80	A1P23630
TRL4	RFS	80	A1P23640
			A1P23650
			A1P23660
*	END		A1P23670
*	CHAIN (2,2)		A1P23680
*	FORTRAN		A1P23690
C	INDEX (1270)		A1P23700
C	PROGRAM TO CONTROL PLOTTING FROM SCRATCH TAPE		A1P23710
C	CAN PLOT SEQUENTIAL GRAPHS EACH HAVING SEVERAL CURVES		A1P23720
C	NXP = NUMBER OF VALUES OF INDEPENDENT VARIABLE FOR A GIVEN GRAPH		A1P23730

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C      NCRVS = NUMBER OF CURVFS TO BE PLOTTED ON A GIVEN GRAPH          A1P23740
C      NPTS = NUMBER OF INTERVALS BETWEEN DEPENDENT VARIABLE VALUES       A1P23750
C      AT WHICH IDENTIFYING SYMBOLS ARE TO BE PLACED                      A1P23760
C      IF NFND = 0, OTHER GRAPHS ARE TO FOLLOW                           A1P23770
C      IF NFND = 1, CURRENT GRAPH IS THE LAST                            A1P23780
C      IF NFLAG = 0, CONTROL WILL BE RETURNED TO CHAIN (1,3)            A1P23790
C      IF NFLAG = 1, RUN WILL BE TERMINATED BY CALL EXIT                  A1P23800
C      DIMENSION XPT(200),YPT(200,4),ABLBL(6),ORDLBL(6),GPHLBL(9),        A1P23810
C      1 CILRL(6,4),RECORD(12),YT(200)                                     A1P23820
C      EQUIVALENCE (KR,ER), (XXN, NERR)                                    A1P23830
C      KP=15                                                               A1P23840
C      REWIND KP                                                       A1P23850
C      XXN=255151606060                                                 A1P23860
B      20 READ TAPE KP, RECORD                                         A1P23870
      READ TAPE KP, NXP, NCRVS, NPTS, NEND, NFLAG                         A1P23880
      READ TAPE KP, ARLRL, ORDLBL, GPHLBL                               A1P23890
      READ TAPE KP, CTTRL                                              A1P23900
      DO 32 KT = 1,NXP                                               A1P23910
      READ TAPE KP, XP(KT)                                             A1P23920
      DO 33 L = 1,NCRVS                                              A1P23930
      DO 33 KT = 1,NXP                                               A1P23940
      READ TAPE KP, YT(KT)                                             A1P23950
      33 YPT(KT,L) = YT(KT)
      READ TAPE KP, FR
      BACKSPACE KP
      IF (KR-NFRR) 61, 60, 61
      60 NFND=1
      61 CONTINUF
      CALL PLOT(NXP,NCRVS,NPTS,NEND,NFLAG,XPT,YPT,RECORD,ABLBL,ORDLBL,    A1P24020
      1 GPHLBL,CILRL)
      IF (NFND) 20,20,21
      21 TIFT(NFLAG)40•40•41
      40 CALL CHAIN (1,2)
      41 CALL EXIT
      END
      * FORTRAN
      * SURROUNTING PLOTS(NXP,NCRVS,NPTS,NEND,NFLAG,X,Y,RECORD,ABLBL,    A1P24100
      1 ORDLBL,GPHLBL,CTTRL)                                           A1P24110
C      SC 4020 ROUTINE FOR PLOTTING SEVERAL CURVES ON A SINGLE GRAPH     A1P24120

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C INDEX(1280)
DIMENSION XT(200),YT(200,4),ARBLBL(6),ORDLPL(6),GPFLCT(9),RECORD(I2)ATP24140
DIMENSION KX1(4),KY1(4),KX2(4),CILBL(6,4),CLBL(6),
1 ORD(6),GPHL(6),YP(200),MRK(4)
WRITE OUTPUT TAPF 6,10
10 FORMAT(15H0 PLOT CALLED)
F TABL1V
CALL CAMRAV(9)
XL = X(1)
XR = X(1)
DO 20 I = 2,NYD
XL = MIN1F(XL,X(I))
20 XR = MAX1F(XR,X(I))
DC = 20.0
CALL DXDYV(1,XL,XR,DX,N,II,NX,DC,IERR)
KX1(1) = 185
KX1(2) = 185
KX1(3) = 585
KX1(4) = 585
KY1(1) = 985
KY1(2) = 955
KY1(3) = 985
KY1(4) = 955
KX2(1) = 200
KX2(2) = 200
KX2(3) = 600
KX2(4) = 600
MRK(1) = 38
MRK(2) = 55
MRK(3) = 63
MRK(4) = 53
YR = 0.0
YT = 0.0
DO 40 J = 1,NCPVS
40 YR = MIN1F(YR,Y(I,J))
40 YT = MAX1F(YT,Y(I,J))
YR = YR*1.10
YT = YT*1.10

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L = 1          A1P24520
CALL DXDYV(2,YB,YT,DY,M,JJ,NY,DC,TERRI)    A1P24530
NX = + 3      A1P24540
NY=-2         A1P24550
TT = -TT      A1P24560
JJ = -JJ      A1P24570
CALL SFTMIV(100,10,70,100)                   A1P24580
CALL GRIDIV(L,XL,XR,YR,YT,DY,N,M,TT,JJ,NX,NY)   A1P24590
DO 60 K = 1,6          A1P24600
60 ORDRL(K) = ORDRLR(K)                      A1P24610
CALL CHSTZV(2,2)      A1P24620
CALL RITSTV(12,18,TARL1V)                    A1P24630
CALL RITE2V(75,330,1023,180,1,36,1,ORDL,NLAST)  A1P24640
CALL RITE2V(330,57,1023,90,1,36,1,ABLBL,NLAST)  A1P24650
CALL PRINTV(72,RFCORD,100,1015)               A1P24660
DO 61 K = 1,9          A1P24670
61 GPHL(K) = GPHLBL(K)                      A1P24680
CALL CHSIZV(3,4)      A1P24690
CALL RITSTV(18,20,TARL1V)                    A1P24700
CALL RITE2V(40,20,1023,90,2,54,1,GPHL,NLAST)  A1P24710
DO 70 J = 1,NCRV$          A1P24720
YP(1) = Y(1,J)          A1P24730
NX1 = NXV(X(1))      A1P24740
NY1 = NYV(YP(1))      A1P24750
DO 50 I = 1,NXP          A1P24760
YP(I) = Y(I,J)
NX2 = NXV(X(1))
NY2 = NYV(YP(I))
TF(NX2*NY2)45,50,45
45 CALL LINEV(NX1,NY1,NX2,NY2)
NX1 = NY2
NY1 = NY2
50 CONTINUE
MRKPT = MRK(J)
KX = KX1(J)
KY = KY1(J)
NS = J
CALL PRINTV(KX,KY,NS,ANY)

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KXC = KX2(J)
  70 92 K = 1,6
  93 CLRL(K) = CILRL(K,J)
  97 CALL PRINTV(26,CLRL,KXC,KY)
  70 CONTINUE
  IF(NEND)90,90,91
  91 CALL FOFTV
      WRITE OUTPUT TAPE 6,80
  80 FORMAT(22HO PLOTTING COMPLETED )
  99 RETURN
END

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A1P24910

A1P24920

A1P24930

A1P24940

A1P24950

A1P24960

A1P24970

A1P24980

A1P24990

A1P25000

A1P25010